EVEKTOR - AEROTECHNIK

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AIRPLANE TECHNICAL DESCRIPTION OPERATING, MAINTENANCE AND REPAIR MANUAL

EV-97 EUROSTAR

MODEL 2000
version R

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Model: ***EV-97 ☆EUROSTAR☆***
model 2000 version R

Serial Number:

Registration:

Owner: .....

.....

.....

The manufacturer invites suggestions and reminders concerning this manual,
and appreciates proposals for corrections.
We invite you to share your experiences with us during operation of your

EV-97 EUROSTAR
MODEL 2000

1. GENERAL

1.1 Record of revisions

Any revisions or amendments to this manual shall be issued in the form of bulletins with attached new pages. It is in the interests of every user to enter such revision into the table of revisions and to replace the existing page by the new one. The revised or corrected text shall be indicated by a vertical line on the page fore-edge and the page shall bear a revision number and date of its issue.

Rev. No.	Affected Section	Affected Pages	Date of Issue	Bulletin No.	Date of Approval	Date Inserted	Signature
MB1	1	2, 3	06/2001	EV97-004a	12.6.2001		Airplane Manufacturer
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	4	9, 10					
2	1	2, 3	10/2001				Airplane Manufacturer
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1.3 Contents

2. TECHNICAL DESCRIPTION

2.1 Basic and general information

The ***EV-97 „EUROSTAR“ model 2000 version R*** is a single engine, all metal, low-wing monoplane of semimonocoque construction with two side-by side seats. The aircraft is equipped with fixed, tricycle landing gear.
The standard powerplant consists of the four-cylinder, 4 stroke, ROTAX 912 (80 hp) engine and the two blade, wooden, V 230 C fixed prop.

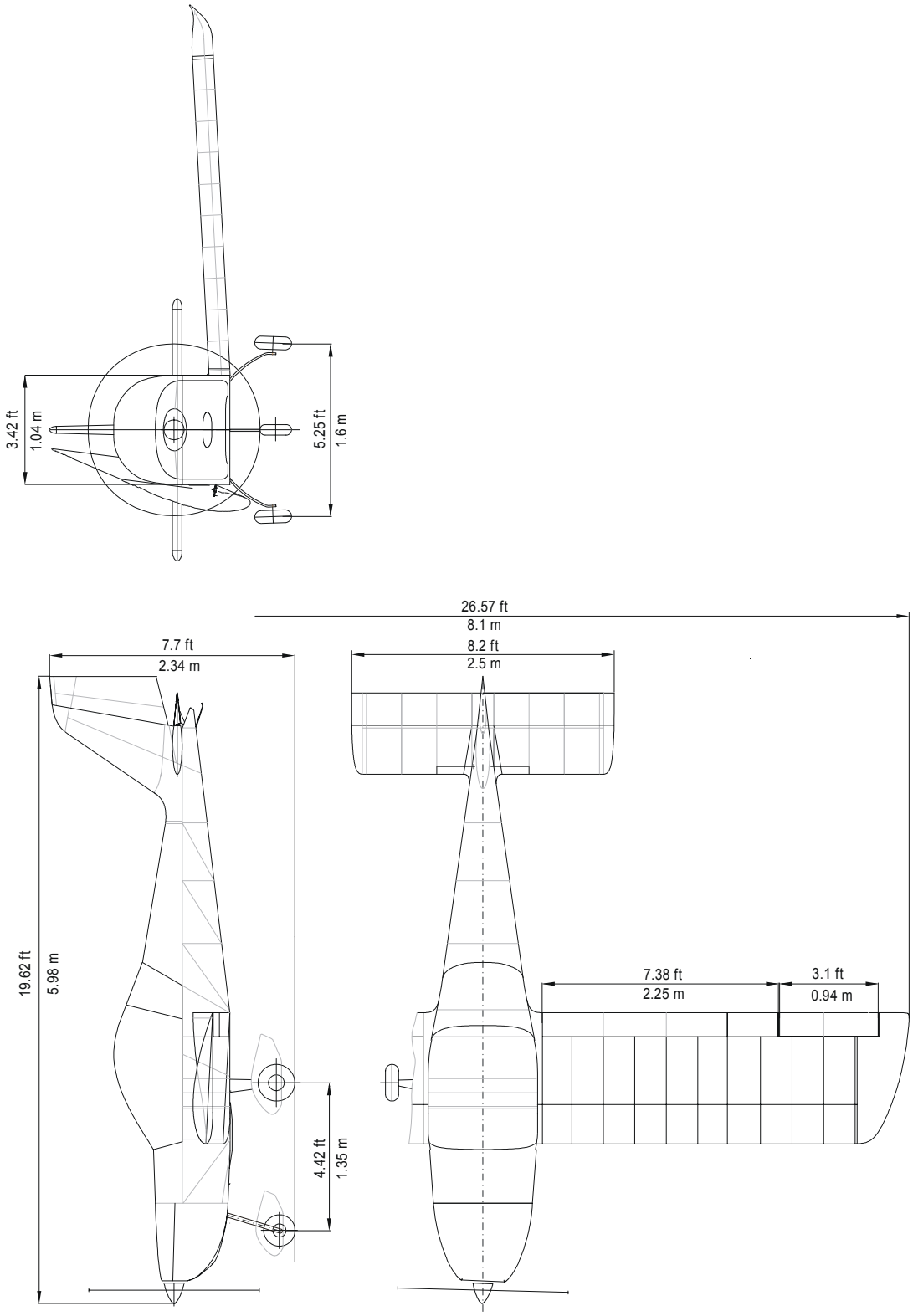
2.1.1 Designation

EV-97 „EUROSTAR“ model 2000 version R is an aircraft especially intended for recreational and cross-country flying with a limitation to non-aerobatic operation.

2.2 Basic technical data

2.2.1 Airplane views

2.2.2 Three-view drawing



2.2.3 Basic dimensions

Wing

span		26.57	ft	8.1 m
area		105.92	sq ft	9.84 m^2
MAC		4.10	ft	1.25 m
loading	(MTOW *992 lb*, 450 kg)	9.37	lbf/sq ft	45.7 kg/m^2
	(MTOW *1058 lb*, 480 kg)	9.99	lbf/sq ft	48,8 kg/m^2

Aileron

area		2.26	sq ft	0.21 m^2

Flap

area		5.60	sq ft	0.52 m^2

Fuselage

length		19.62	ft	5.98 m
width		3.41	ft	1.04m
height		7.68	ft	2.34m

HTU

span		8.20	ft	2.5 m
HTU area		20.99	sq ft	1.95m^2
elevator area		8.61	sq ft	0.8 m^2

VTU

height		4.07	ft	1.24m
VTU area		10.76	sq ft	1.0 m^2
rudder area		4.30	sq ft	0.4 m^2

Landing gear

wheel track		5.25	ft	1.6 m
wheel base		4.42	ft	1.35 m
main wheel diameter		14	in	350 mm
nosewheel diameter		14	in	350 mm

2.2.4 Weight

Empty weight (standard equipment)		*606*	*lbs± 3%*	275 kg ± 3%
Maximum Take-off weight	(Czech Rep., Germany..)	*992*	*lbs*	450 kg
	(Slovak Republic, Canada)	*1058*	*lbs*	480 kg
Maximum Landing weight	(Czech Rep., Germany..)	*992*	*lbs*	450 kg
	(Slovak Republic, Canada)	*1058*	*lbs*	480 kg
Maximum fuel weight	(65 l fuel tank - standard)	*104*	*lbs*	47 kg
Maximum weight in Baggage Compartment		*33*	*lbs*	15 kg

NOTE

Actual empty weight is stated on the placard "LOAD LIMITS," located on the cockpit canopy.

2.2.5 Center of gravity

Empty weight CG (standard equipment)	18±2	% MAC
Operating CG	20-34	% MAC

(MAC...Mean Aerodynamic Chord)

2.2.6 Operating limitations

Refer to the PILOT'S OPERATING HANDBOOK (POH), Section 2 for more details about the following operating limits:

- Airspeed limits
- Weight limits
- CG Range limits
- Approved maneuvers

Additional rules are of a more common character and result from generally valid flight regulations. It is in every user's interest to be familiar with these regulations, rules and restrictions.

2.3 Technical description of the airplane

2.3.1 General

The ***EV-97 „EUROSTAR“ model 2000 version R*** airframe is of semi-monocoque construction formed with metal reinforcements, bulkheads and duralumin cover. Pop-rivets are used for joints and some non-supporting parts are made from fiberglass.

2.3.2 Fuselage

The fuselage has a semi-monocoque construction formed with reinforcements and duralumin covers. The fuselage cross-section is rectangular in the lower section and elliptical in the upper one. The tail fin is an integral part of the fuselage. In the middle section of the fuselage there is a two-man cockpit which is accessible by unfolding the one-part perspex overlap canopy. The engine section in the nose is separated from the crew by a firewall which the engine mount is attached to.

2.3.3 Wing

The rectangular wing is of a monospar construction with an auxiliary spar for the ailerons and flap attachments. All the elements are riveted together. Fiberglass wing tips are riveted to the ends of the wings and the wings can be equipped with an optional folding mechanism for convenient storing.

2.3.3.1 Ailerons

The ailerons are of rectangular shape on each half of the wing and are attached to the wing with hinges. An aileron is formed with the ribs and cover, which forms a hollow section.

2.3.3.2 Flaps

Two-third 's of each half of the wing is fitted with a flap. The flaps are of rectangular shape and are formed with the ribs and cover, which forms a hollow section. The flap is attached to the wing with a hinge.

2.3.4 Horizontal tail unit

The rectangular HTU consists of a stabilizer and elevator with a trim tab. The semi-monocoque construction of the HTU consists of duralumin ribs, spar and cover. The width of 8.2 ft (2.5 m) enables transport without dismantling.

2.3.4.1 Stabilizer

The stabilizer is rectangular in shape and formed with a duralumin cover and ribs. The stabilizer is attached to the fuselage with two pins at the leading edge and secured with two screws at the stabilizer trailing edge.

2.3.4.2 Elevator

The elevator is rectangular in shape and formed with a duralumin cover and ribs. The elevator is attached to the stabilizer with a hinge. There is also a hinged trim tab at the elevator trailing edge.

2.3.4.3 Trim tab

Each elevator is equipped with the one trim tab of rectangular shape. The tab is formed with duralumin cover which forms a hollow section. The span of the trim tab is 2/3 of the elevator spanwise.

2.3.5 Vertical tail unit

The trapezoidal VTU consists of the fin and rudder. The rudder is attached on the fin by two hinges. The frame of the VTU is composed of a metal sheet spar and a duralumin cover.

2.3.5.1 Fin

The fin is an integral part of the fuselage rear section and is formed with a duralumin spar and cover. The fin tip is formed with a fiberglass cover, where the anticollision beacon can be installed. The fillet between the fin and rear upper fuselage part is formed with a fiberglass fillet cover.

2.3.5.2 Rudder

The rudder is of trapezoidal shape and formed with a duralumin spar and cover and attached by two hinges at the fin. The rudder upper tip is formed with a fiberglass cover.

2.3.6 Landing gear

2.3.6.1 General description

The aircraft is equipped with fixed nosewheel landing gear. The nosewheel is steerable, but as an option, can be casterable, only.

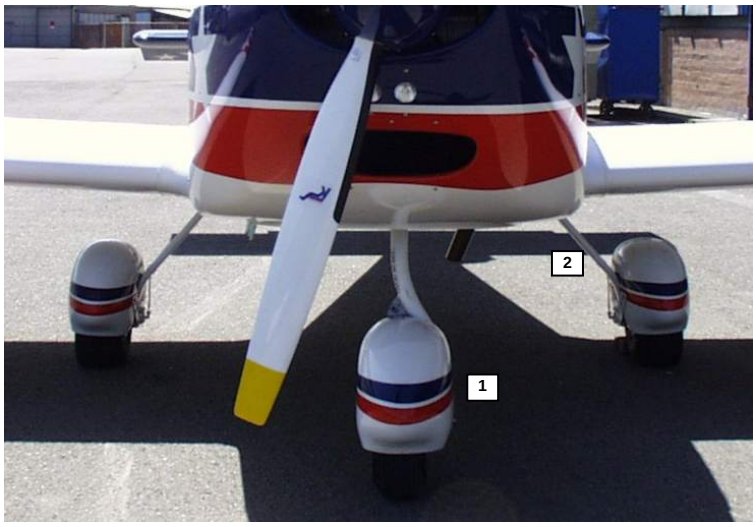


Fig. EV-97 Eurostar
1 - Steerable nosewheel
2 - Main landing gear

2.3.6.2 Main landing gear

2.3.6.2.1 Description

The main landing gear consists of the left and right landing gear legs. The legs are formed from fiberglass springs and are fixed by means of screws in the fuselage casing under the seats. Wheel axis is screwed at the lower part of the main landing gear legs. The main wheels on both legs are equipped with hydraulic disc brakes controlled with toe brake pedals mounted on the rudder pedals. The wheels can be covered with the fiberglass fairings (wheel pants) or mudguards.

2.3.6.3 Nosewheel landing gear

2.3.6.3.1 Description

Steerable nose landing gear consists of front landing gear leg, rubber rope suspension unit and suspension stop. The nose leg is made of a bended steel tube, attached to the firewall by two bearings. The axle, with wheel attached, is connected to the welded bushing in the bottom part of the leg. Two rods are used for the leg steering by the control pedals.

2.3.6.3.2 Main landing gear layout

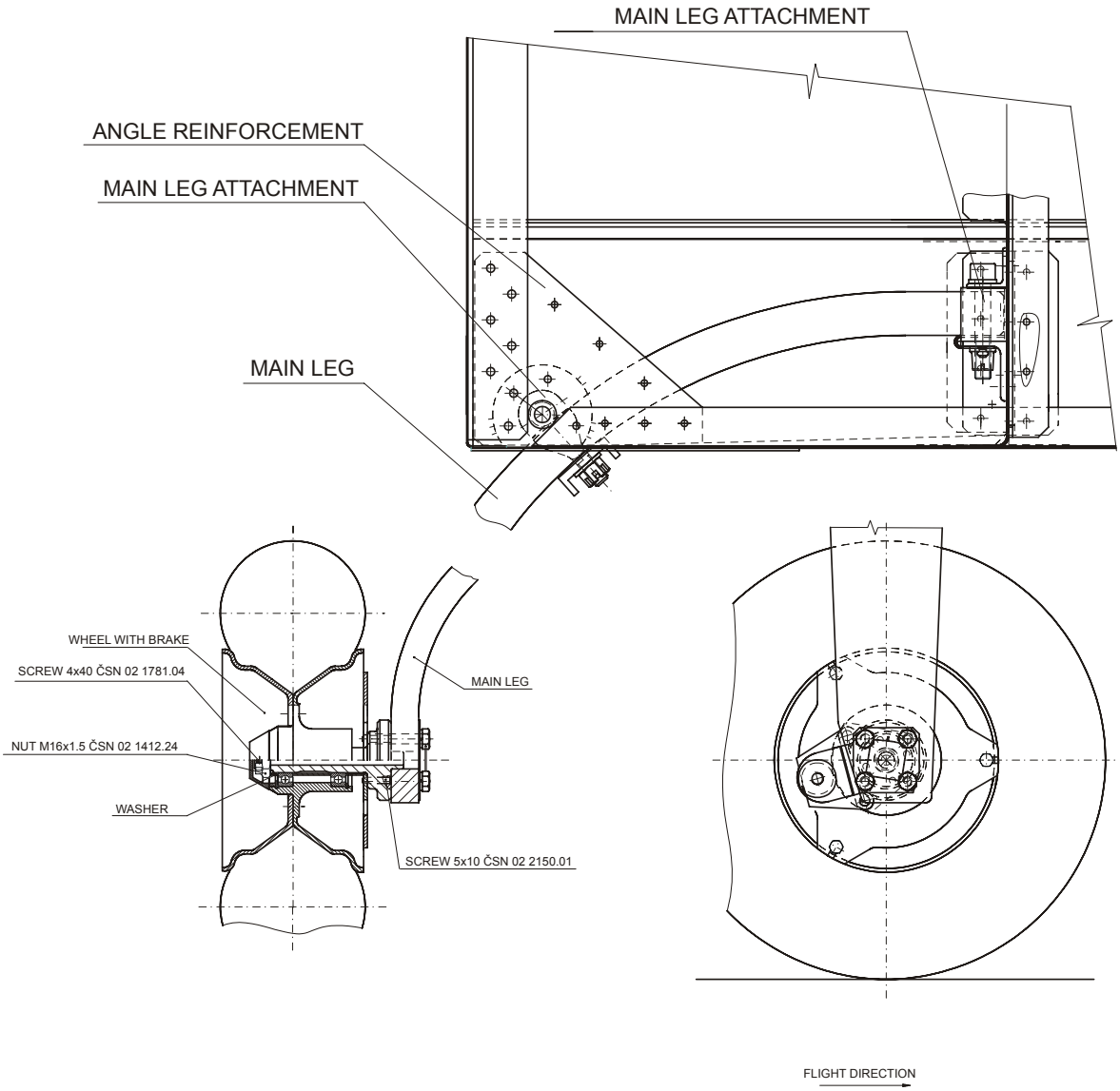
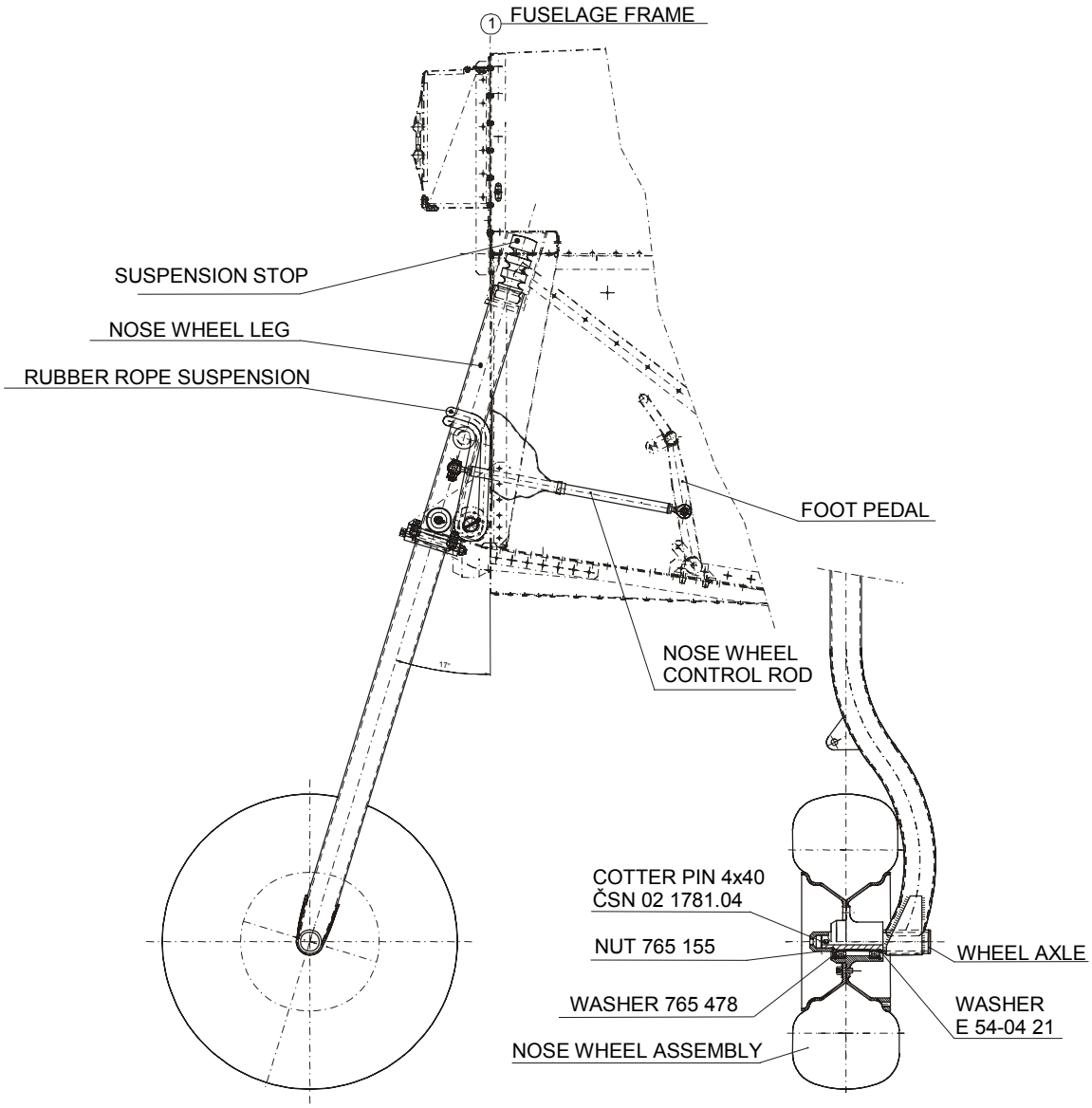


Fig. Main leg attachment into fuselage

2.3.6.3.3 Steerable nosewheel landing gear layout



2.3.6.4 Auxiliary tail skid

2.3.6.4.1 Description

The auxiliary tail skid is attached at the lower rear part of the fuselage and protects the aircraft from inadvertent damage during tail-down landing conditions. The tail skid is formed from composite material.

Fig. Auxiliary tail skid

2.3.6.5 Wheel brakes

2.3.6.5.1 Description

Both wheels on the main landing gear are equipped with hydraulic disc brakes. The brake system consists of the brake pedals (pilot standard, co-pilot as an option), hydraulic brake master cylinders, plastic hoses, brake caliper with the hydraulic brake cylinder, brake pads and the brake disc which is bolted onto the inner part of the rim.

Fig. The brake on the left wheel
1- brake caliper with the hydraulic cylinder,
2 - brake disc
3 - brake fluid hose,
4 - brake pad holder,
5 - terminal,
6 - air bleed screw

2.3.6.5.2 Brake control

The brakes on both wheels are controlled independently by toe brake pedals mounted on the pilot's rudder pedals (the brake pedals for the co-pilot are optional).

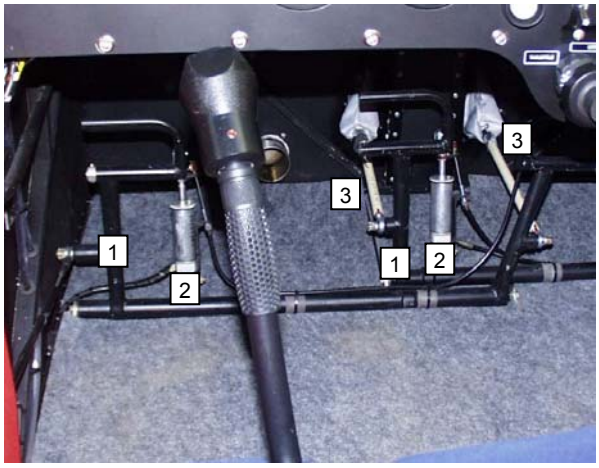
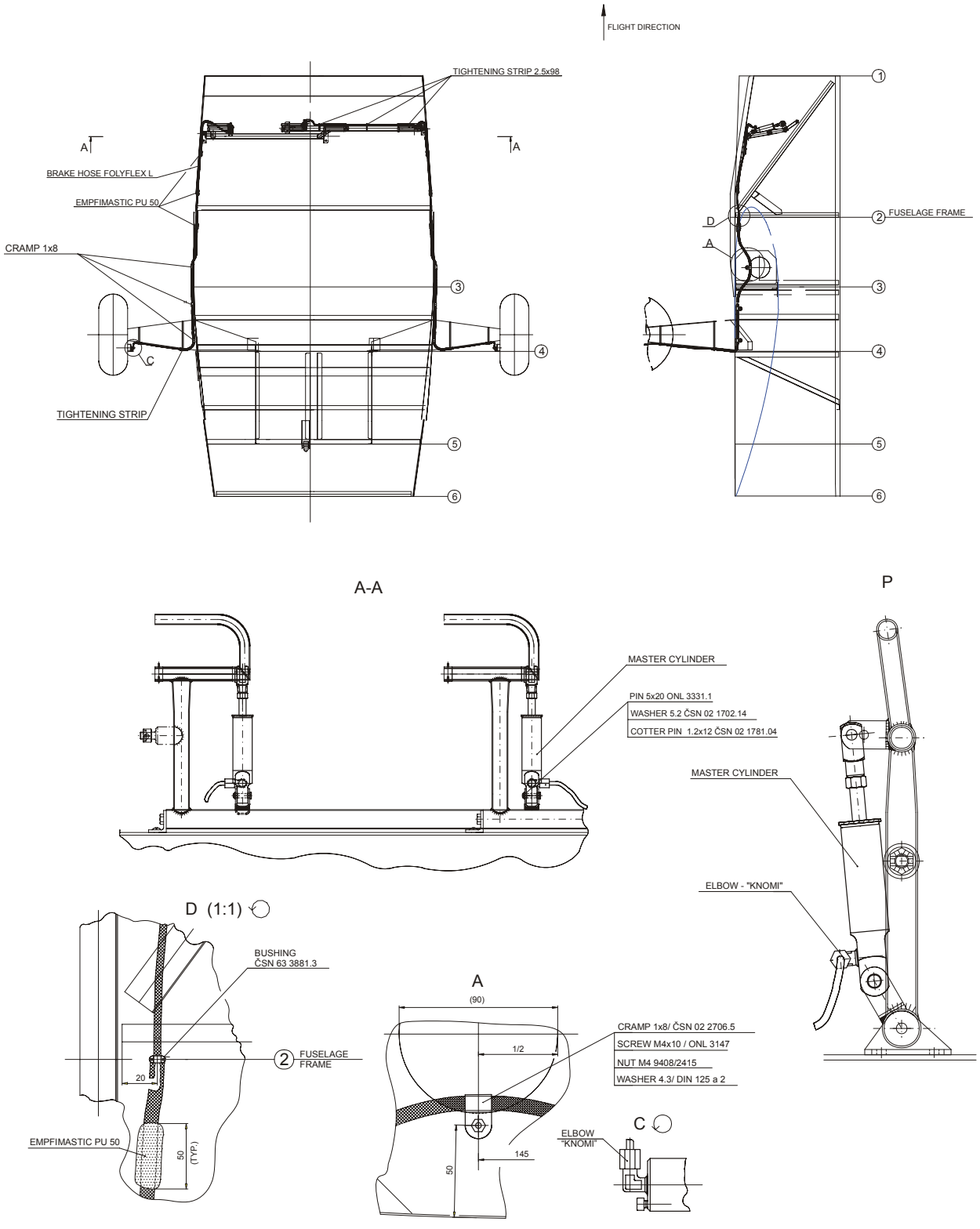


Fig. The brake control with toe brake pedals

1 – ruder pedals, 2 – brake cylinder, 3 – nose wheel steering rod

2.3.6.5.3 Brake system layout

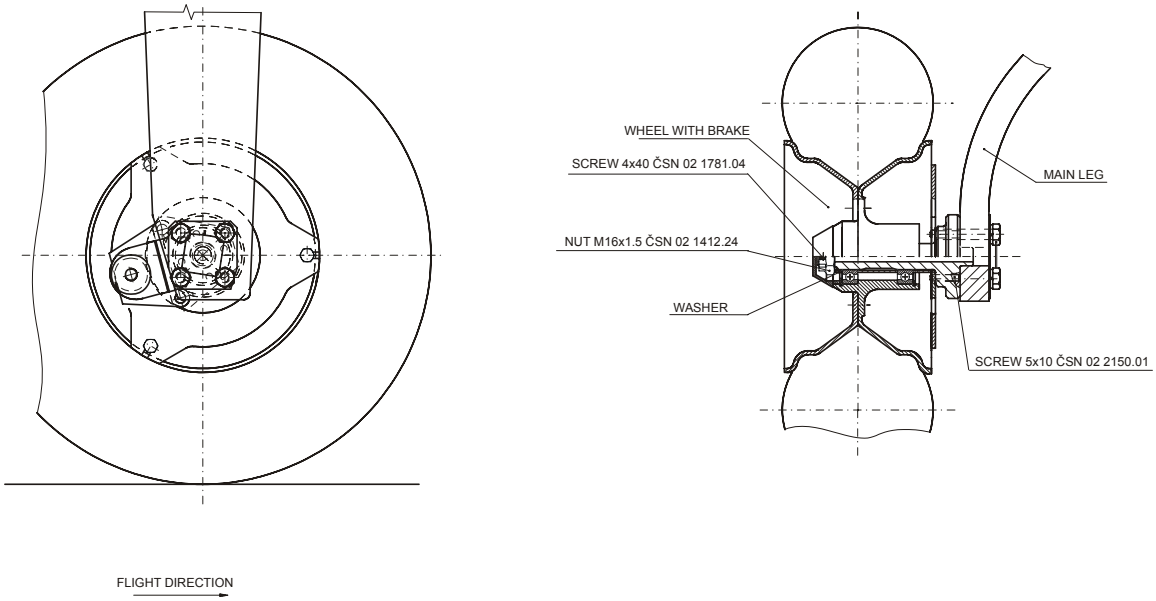


2.3.6.6 Wheels

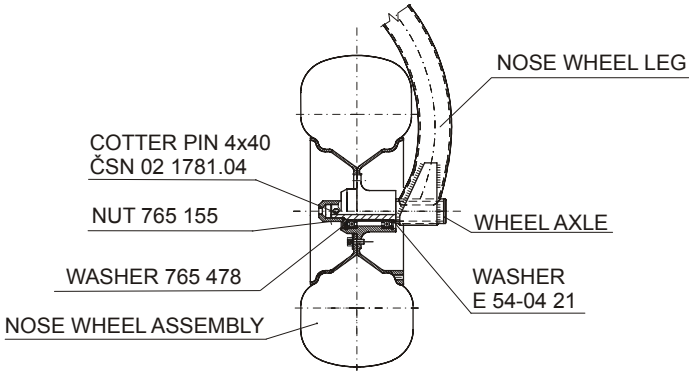
2.3.6.6.1 Description

All the wheels consist of a two-part casting rim with a tire and tube. The main wheels are on an axle attached to the main gear leg, fastened by the nuts.

2.3.6.6.2 Main undercarriage wheel layout



2.3.6.6.3 Nosewheel layout



2.3.6.7 Tires

Main landing gear ..... SAVA 14x4 (standard) or ptional:
DELI TIRE 15 x 6.00 - 6 (aircraft tire)
AIR TRACK 15 x 6.00 - 6 (aircraft tire)
TOST 15 x 6.00 – 6 (aircraft tire)

Nose landing gear ..... SAVA 14x4 (standard) or ptional:
DELI TIRE 15 x 6.00 - 6 (aircraft tire)
AIR TRACK 15 x 6.00 - 6 (aircraft tire)
TOST 15 x 6.00 – 6 (aircraft tire)

2.3.7 Cockpit

2.3.7.1 Description

The comfortable cockpit has a side-by-side, double control arrangement, which provides the crew with an excellent view and comfort. It protects the crew from adverse weather conditions, and allows easy access to the controls and instruments.
The instrument panel is located in front of the crew. The flap control lever, elevator trim tab lever and optional towing mechanism release lever are located on the quadrant between seats. A baggage compartment is situated behind the seats.
The cockpit floor is covered with a removable carpet and the seats are also covered with a thin upholstery. The inerior cockpit sides are covered with padded panels containing pockets.
The actual cockpit controls and instrument arrangement is described later.

Fig.: Cockpit of the Ev-97 “EUROSTAR” model 2000 version R

2.3.7.2 Cockpit controls

The standard cockpit control arrangement is shown in the following figure. A detailed instrument panel is shown in par. 2.3.8.3.

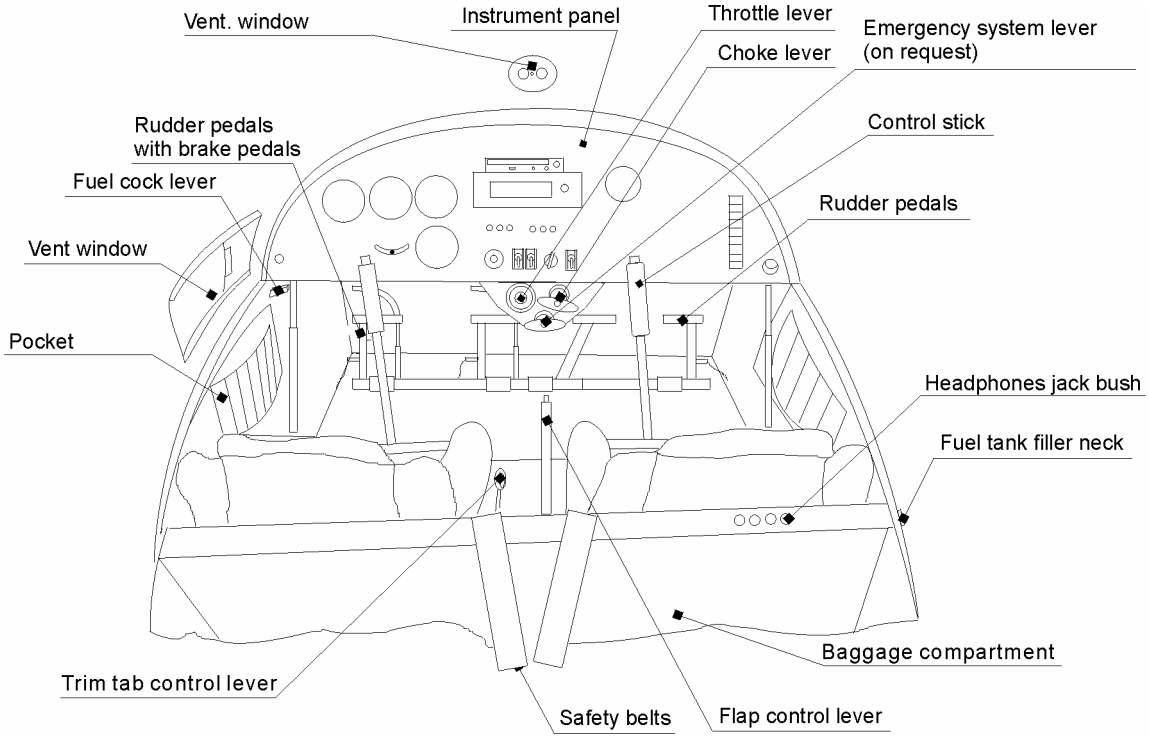


Fig. quadrant between the seats
1- flaps control lever, *2* - elevator trim tab control lever

2.3.7.3 Cockpit canopy

The semi-tear shape cockpit canopy consists of two parts. The front perspex portion can be tilted forward and is attached to a steel frame. The fixed rear portion can be made of either Al-alloy sheet or perspex. The canopy is attached to the nose section of the fuselage by two pins which make it possible for the canopy to be tilted forward. For easier manipulation, the weight of the canopy is counterbalanced by two gas struts which allow effortless opening. The lower frame has handles outside the canopy for lifting and the canopy is equipped with a lock on the upper rear section of the frame. Placards are attached to the canopy showing the lock/unlock directions of movement.

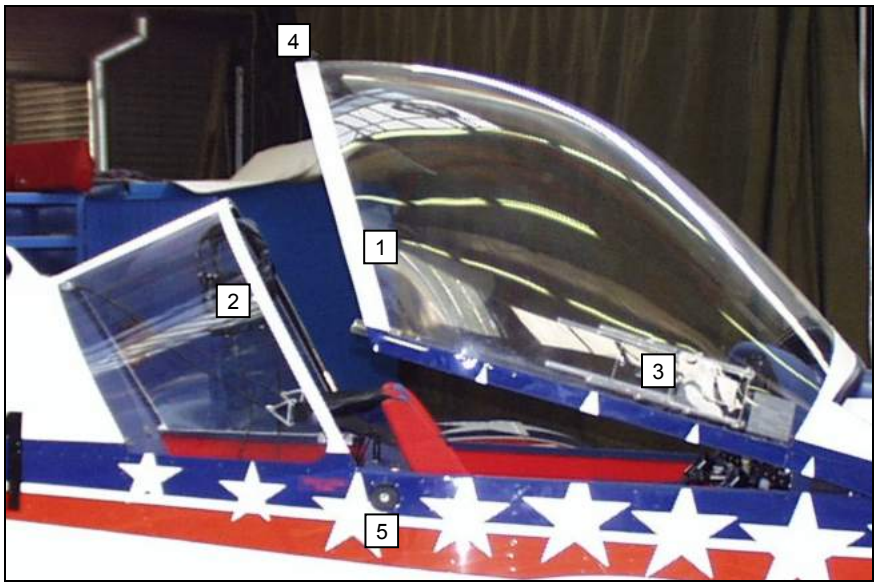


Fig. Two-parts cockpit canopy
1- front tilted canopy, *2* - rear fixed canopy, *3* - side vent window, *4* - canopy lock, *5* - fuel tank filler cap

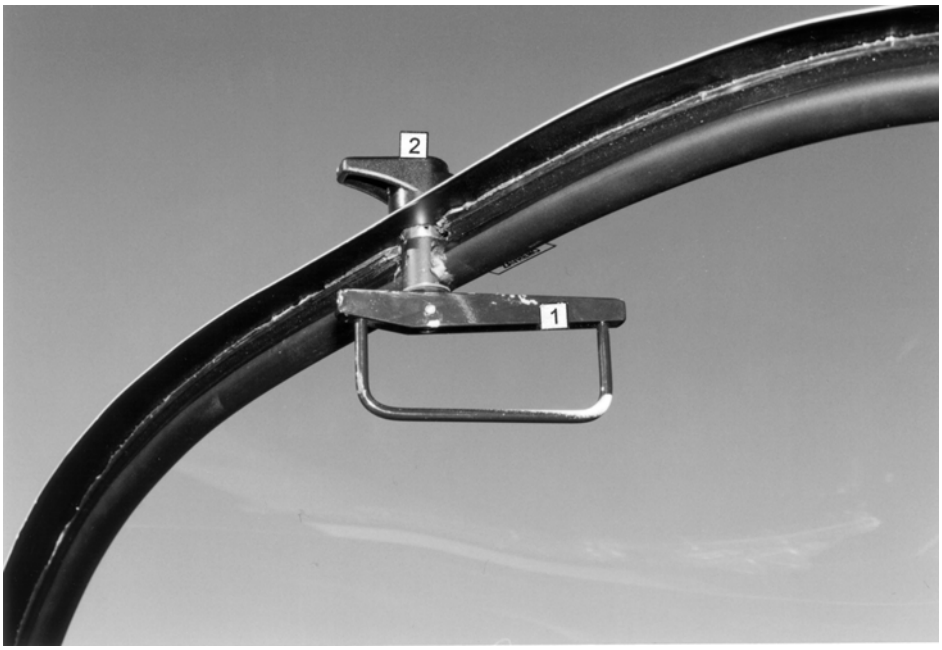


Fig. Cockpit canopy lock
1- inside lever
2 - outside lever (with a lock)

2.3.8 Equipment

2.3.8.1 Seats and safety harness

The plane has two side-by-side seats which are fixed and covered with upholstery, each equipped with seatbelts. Adjustable rudder pedals are optional. The seatbelts are attached alongside the seat and in the middle of the bulkhead behind the baggage compartment. The seatbelts can be provided as either three or four point safety belts.

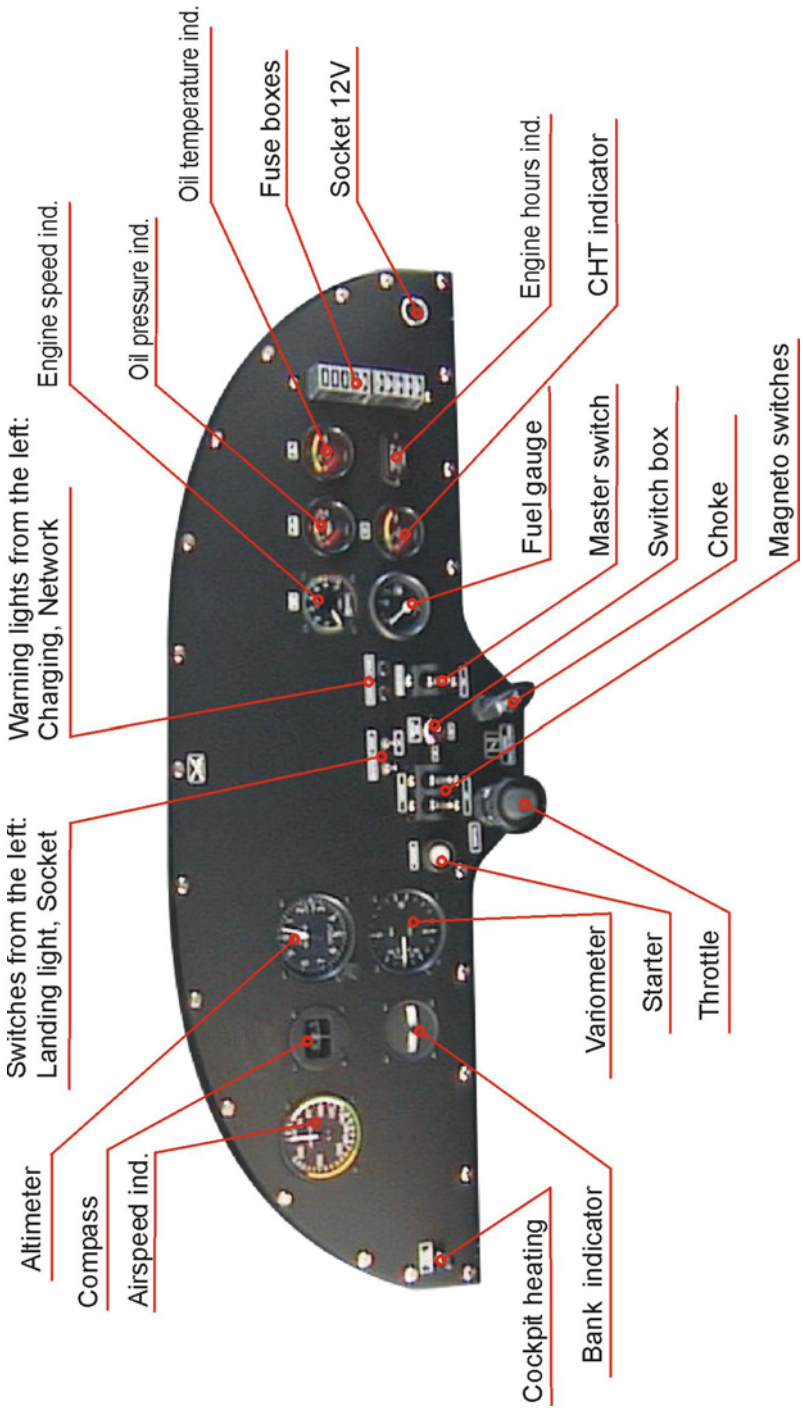
2.3.8.2 Baggage compartment

The baggage compartment is situated behind the seats.
Maximum baggage weight is stated on a placard located near the compartment.
There are pockets on both cockpit interior sides for small objects (maps, pencils, keys etc.).

Fig. Cockpit interior
1- seat upholstery, *2* - safety seat belts, *3* - baggage compartment

2.3.8.3 Instrument panel

The following instrument panel is installed in the ***EV-97 "EUROSTAR" model 2000 version R*** airplane, S/N : (example)



2.3.9 Avionics

The following instruments are installed in ***EV-97 "EUROSTAR" model 2000 version R*** airplane, S/N ……..: (example)

2.3.9.1 Flight and navigation instruments

1 Airspeed indicator .....
1 Altimeter .....
1 Compass .....
1 Variometer.....
1 Bank indicator .....

NOTE
Refer to the Manuals supplied with above listed instruments for operation.

2.3.9.2 Powerplant instruments

The analogous powerplant instruments are installed in the EV-97 aeroplane model 2000 version R:
1 Engine RPM indicator
1 CHT indicator
1 Oil temperature indicator
1 Oil pressure indicator
1 Engine hours indicator

The following powerplant instruments are installed in ***EV-97 "EUROSTAR" model 2000 version R*** airplane,S/N …….:

1 Electric Float Fuel Gauge ..... SW 13.803

2.3.10 *Additional equipment*

The following optional equipment is installed in the ***EV-97 „EUROSTAR“ model 2000 version R*** airplane, S/N :

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2.3.11 Control system

2.3.11.1 Longitudinal control system description

The airplane is equipped with a classic dual control system. The elevator is controlled by a control stick, with connecting rods and arms. A control stick push/pull movement is transmitted, by a rod inside the quadrant between the seats to the elevator through a two-armed lever located underneath the baggage compartment floor cover. The angular displacement of the two-armed lever is transferred by a longitudinal motion of two rods, connected with a single arm lever, inside the middle rear part of the fuselage. The rear rod is connected to the elevator single-arm lever.
A control stick motion is limited by two stops. Both control sticks have a common "push-down" stop on the center-section and each control stick has a "pull-up" stop formed with a reinforcement riveted on the front edge of each seat. The rods have adjustable ends and swivel bearings to adjust the elevator deflections.

2.3.11.2 Lateral control system description

The ailerons are controlled by control sticks, connecting rods and arms. A control stick lateral motion is transferred by a short rod in the cockpit to a longitudinal movement of a longer rod in the wing. This in turn transferrs to the angular displacement of a two-armed lever attached to the wing main spar. The two-armed lever angular movement is transferred to the ailerons by short rods. The rods have adjustable ends to adjust the aileron deflections. The control stick has a termination stop.

CAUTION
To adjust an aileron deflection, never use the adjustable end of the short rod which is accessible when wing fillet (covering the space between the wing and fuselage) is removed. See Figure in 2.3.11.7.1.

2.3.11.3 Directional control system description

The rudder control system is dual. The rudder is controlled by cables attached at the rudder pedals and guided alongside the fuselage sides to the rudder. The rudder pedals are attached to the cockpit floor. There are toe brake pedals on the pilot's rudder pedals to operate the main wheel brakes (co-pilot toe brake pedals optional). The cables are connected to the hinges in the lower part of the rudder leading edge. The cables are prestressed by means of bungees. One end is attached to the outer pedal with the second one attached to the firewall. The rudder control is connected to the nosewheel landing gear to control the nosewheel by the adjustable rods.

2.3.11.4 Flap control system description

The wing flaps are controlled by a control lever in a changing gate. The lever push/pull movement is transferred to a longitudinal movement of a rod guided inside the quadrant between the seats. Then to an angular displacement of a two-armed lever welded onto a tube connecting left and right flap. The flap control lever is located in the quadrant between the seats. When a lock button located on the upper end of the lever is pressed, the lock pin is pulled out of the groove in the changing gate. The flaps can then be extended to a position for takeoff or landing. The flap position is locked when the lock button is released.

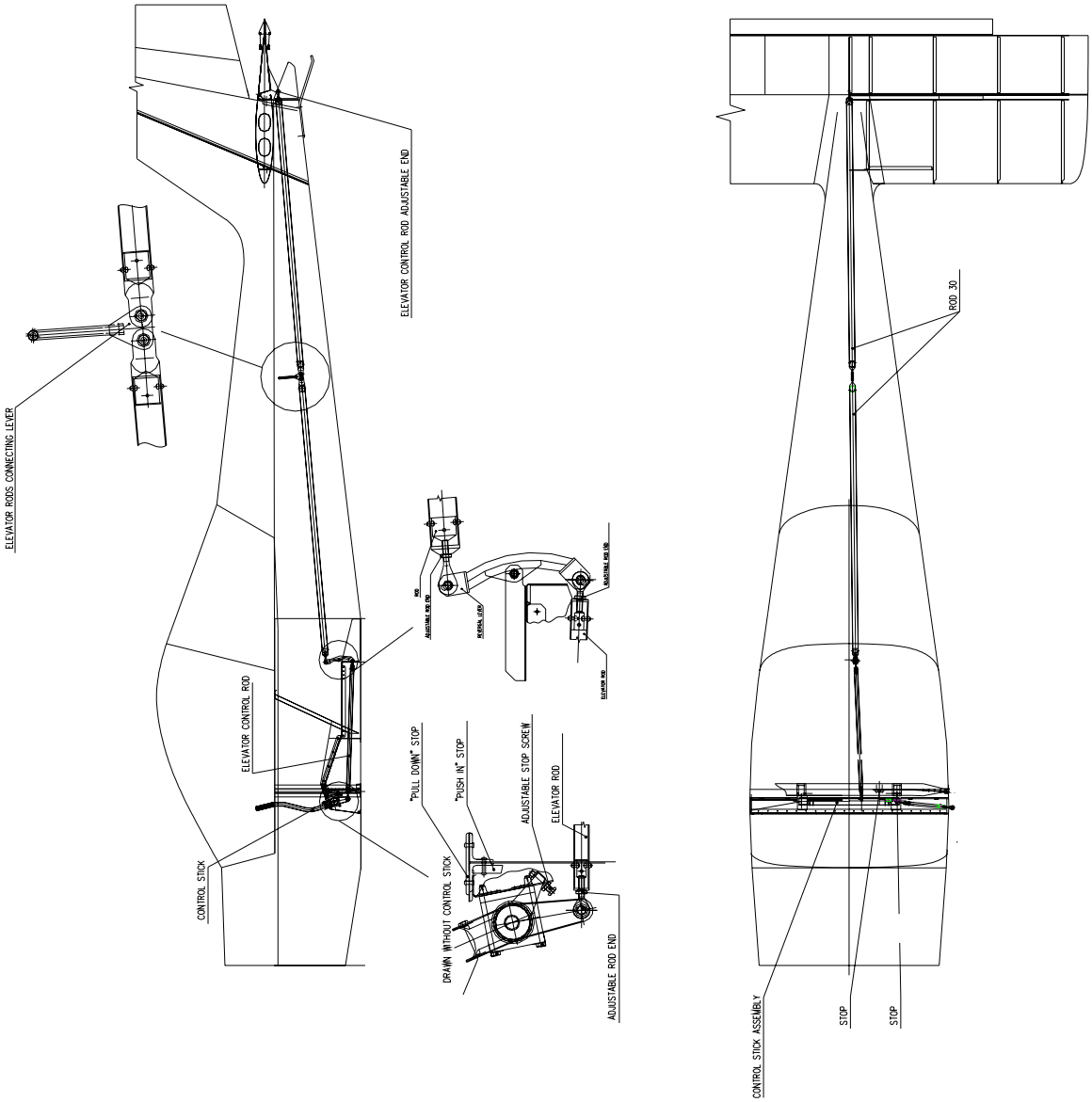
2.3.11.5 Trim tab control system description

The elevator trim tab is controlled by the control lever located in the quadrant between the seats. The trim tab control lever movement is transmitted to the trim tab displacement by bowden cables. Maximum trim tab deflections can be adjusted by means of adjusting screws on the upper or lower tab surface.

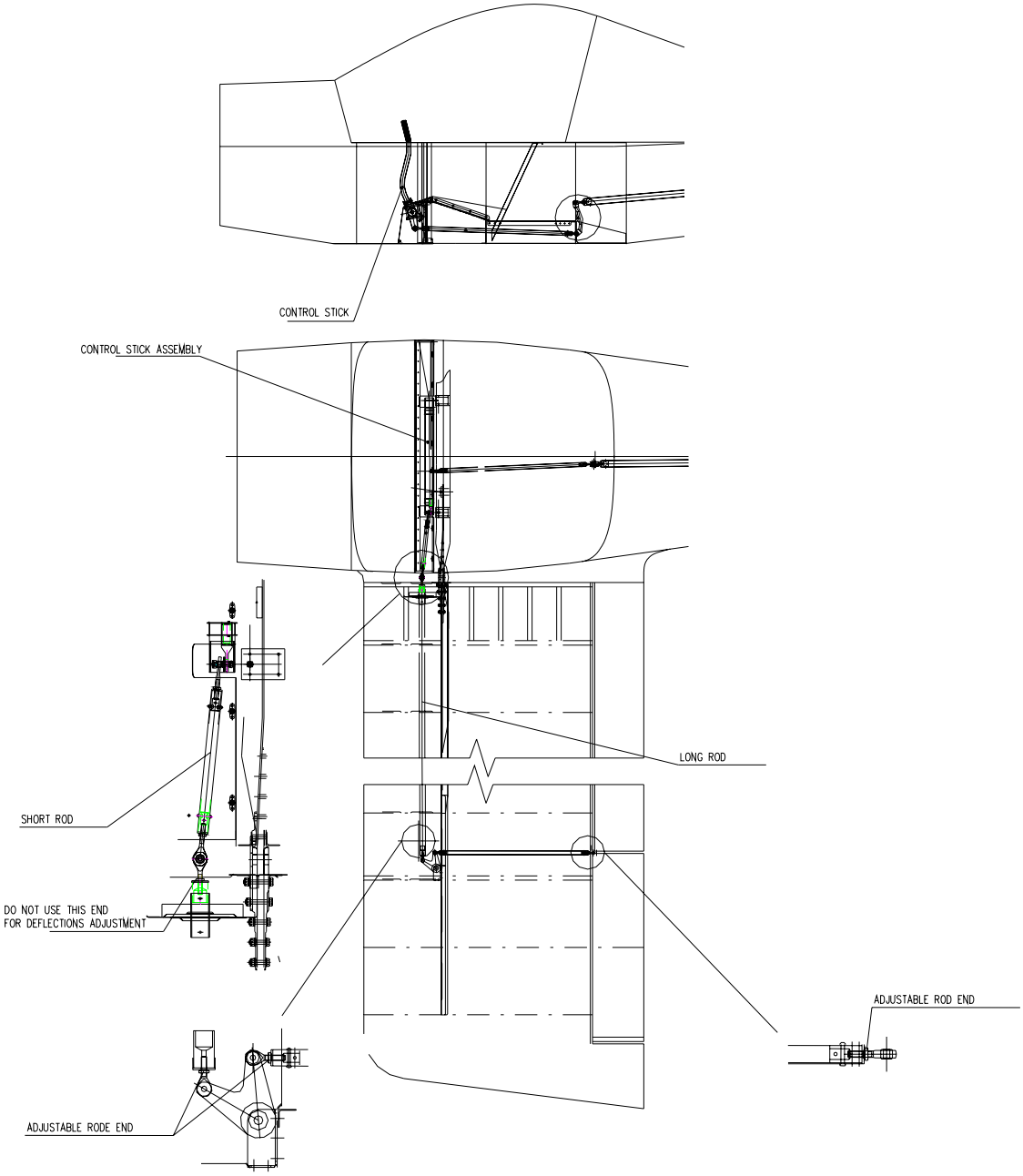
2.3.11.6 Nosewheel control system description

The aircraft is equipped with steerable nosewheel landing gear. In this case the wheel control system is connected to the rudder pedals. The nosewheel control system consists of the rods, connecting pedals and a two-armed lever welded to the nose landing gear strut.

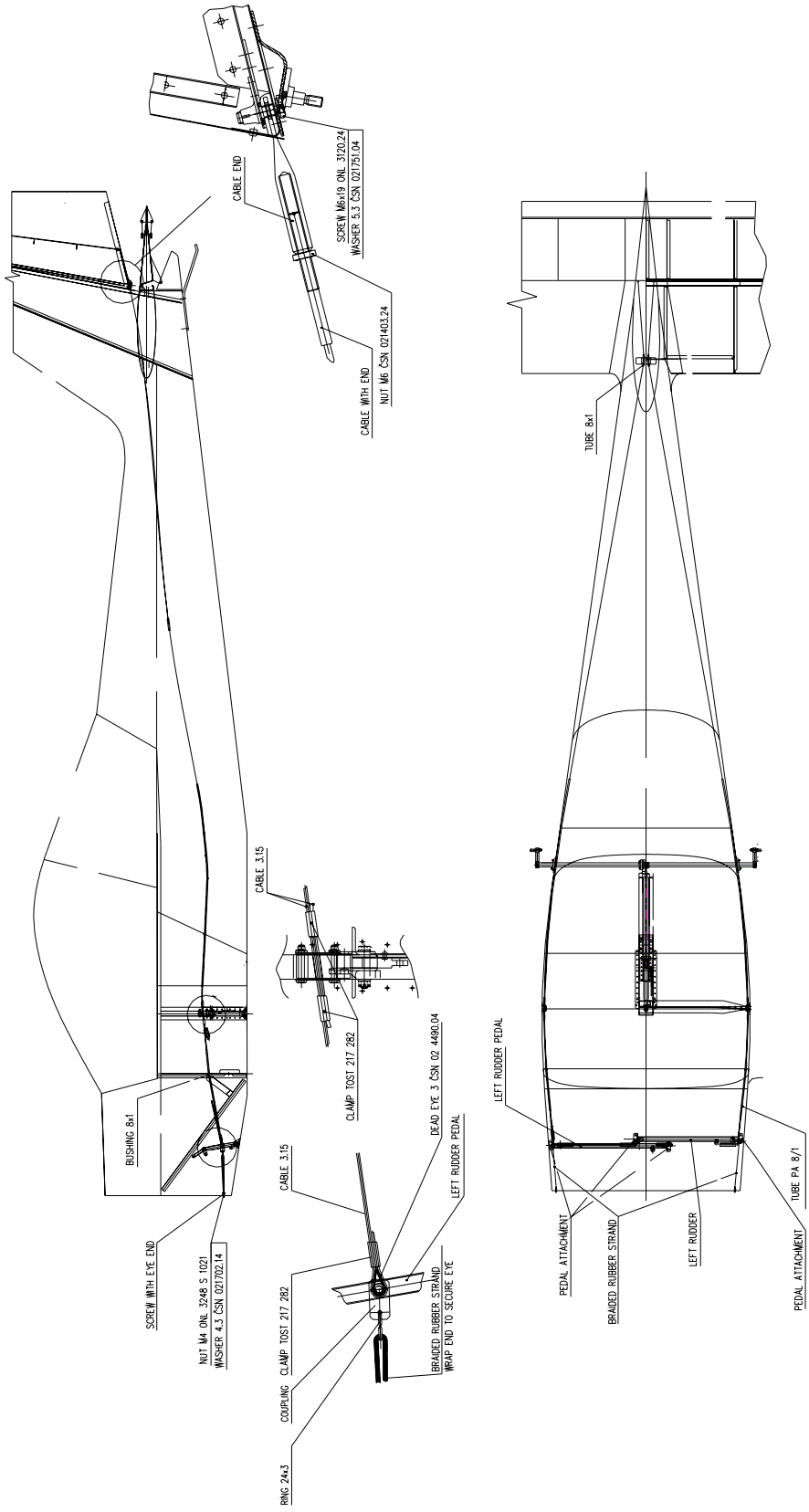
2.3.11.7 Longitudinal control system layout



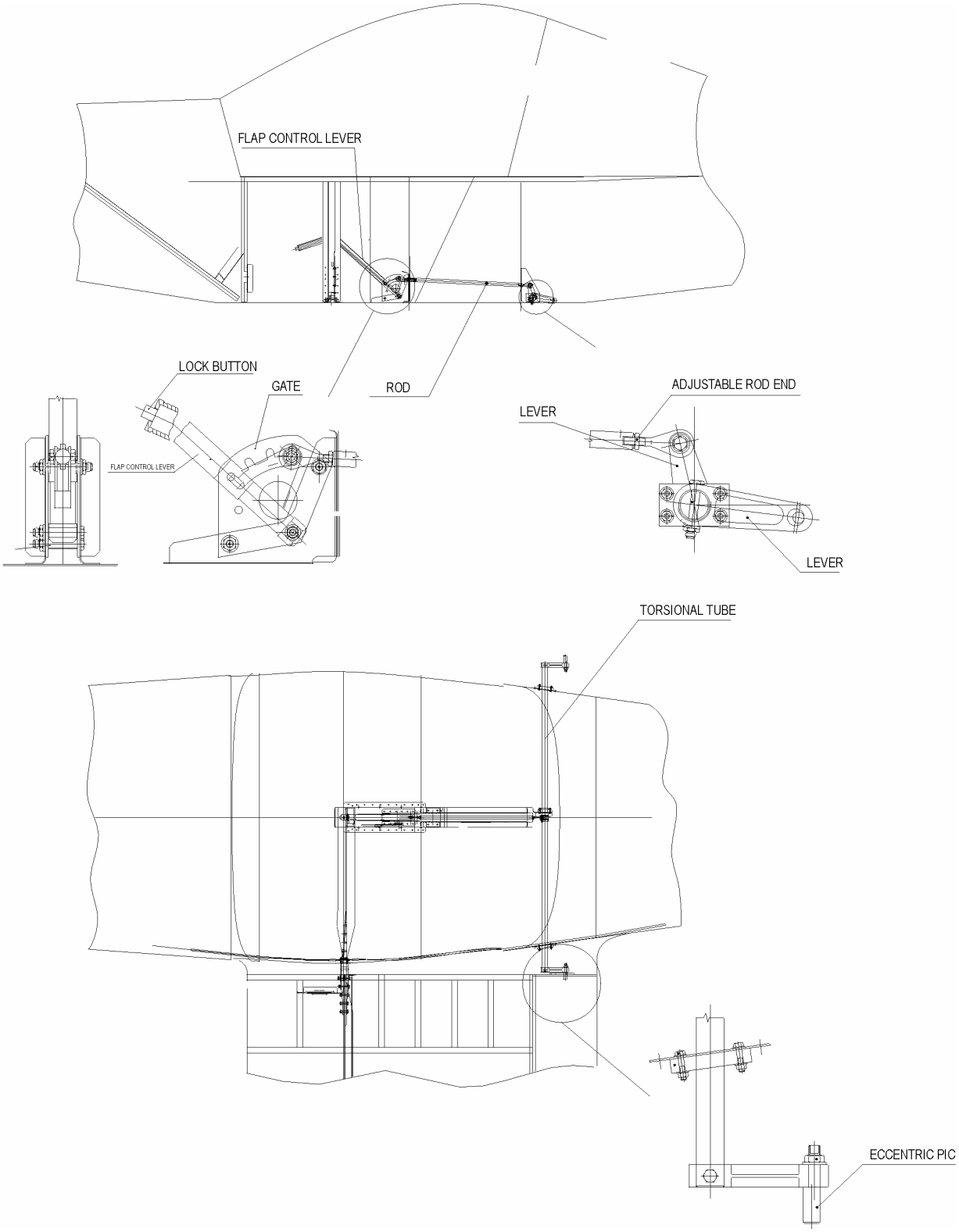
2.3.11.7.1 Lateral control system layout



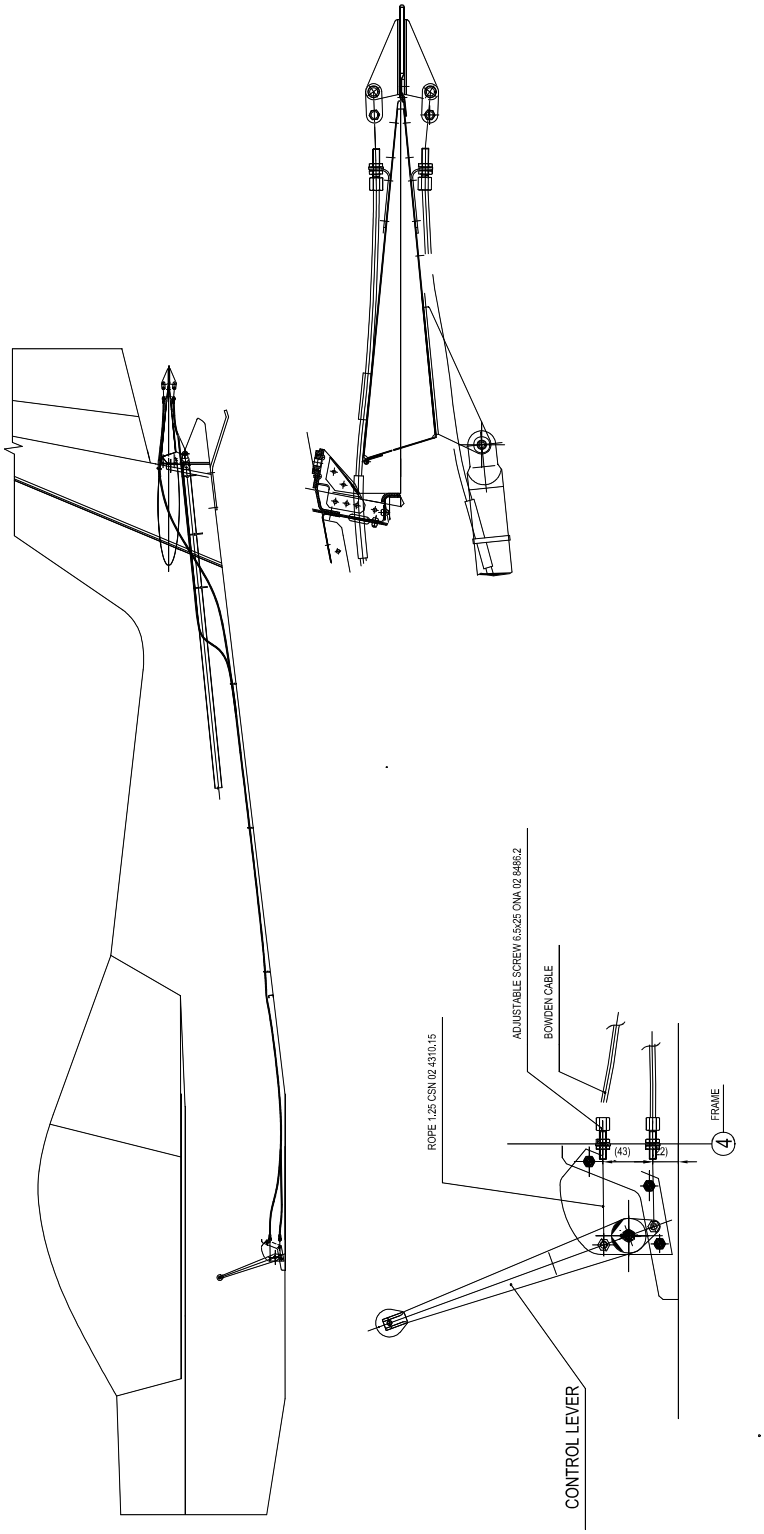
2.3.11.8 Directional control system layout



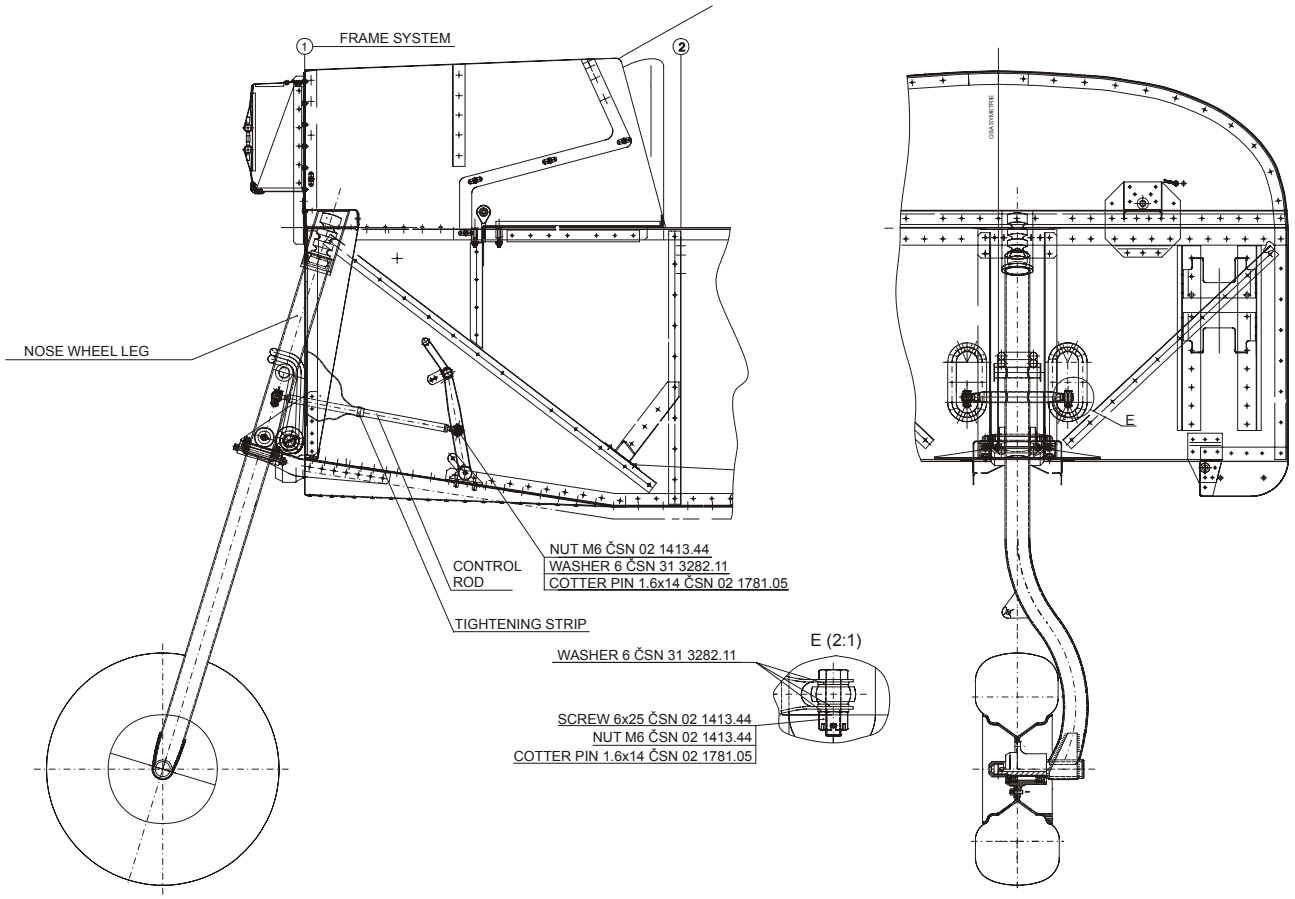
2.3.11.9 Flap control system layout



2.3.11.10 Trim tab control system layout



2.3.11.11 Nosewheel control system layout



2.3.12 Powerplant

2.3.12.1 Brief description

The standard powerplant consists of the four-cylinder, 4 stroke ROTAX 912 (A or UL version, 80 hp) engine and 2 blade, wooden, V 230 C fixed prop. The ROTAX 912 S resp. ULS (100 hp) engine may be installed as option. The engine data is scanned by an integrated digital engine instrument, the FLYdat, (standard equipment) or by analog instruments (optional).
Other props are optional.
The standard powerplant is shown in the following figure:

Fig. EV-97 Powerplant (standard powerplant)
1- ROTAX 912 engine (80 hp), *2* – V 230C prop, 3 - oil tank,
4 - coolant expansion tank, 5 - coolant overflow bottle

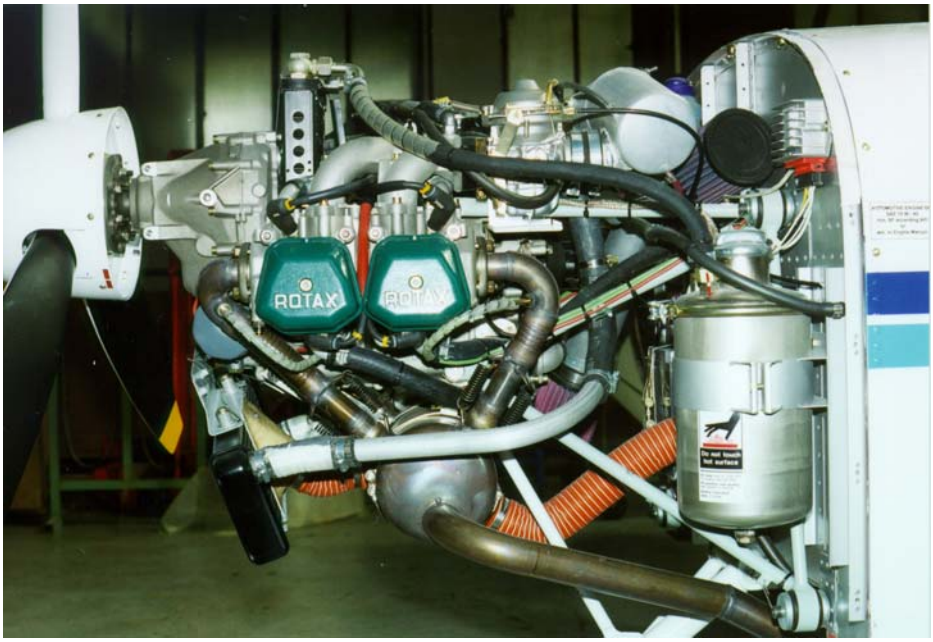


Fig. EV-97 Powerplant (ROTAX 912S 100 hp)
1- ROTAX 912 S engine, *2* - SR 200b prop, 3 - oil tank,
4 - coolant expansion tank, 5 - coolant overflow bottle

2.3.12.2 Engine

2.3.12.2.1 Description

The Rotax 912 S is a 4-stroke, 4 cylinder, horizontally opposed, spark ignition engine and has one central camshaft-push-rods-OHV.
Liquid cooled cylinder heads, ram air cooled cylinders.
Dry sump forced lubrication.
Dual breakerless capacitor discharge ignition.
The engine is fitted with electric starter, AC generator and mechanical fuel pump. Prop drive via reduction gear with integrated shock absorber.
Refer to the Rotax documentation for more details about different versions.

2.3.12.2.2 Technical data

ROTAX 912 engine is installed in the aircraft of S/N

Engine Model:		**ROTAX 912A** (or UL)		**ROTAX 912S** (or ULS)	
Engine Manufacturer:		Bombardier-Rotax GMBH			
Power	Max Take-off:	59.6 kW / 80 hp at 5800 rpm, max.5 min.		73.5 kW / 100 hp at 5800 rpm, max.5 min.	
	Max. Continuous:	37.7 kW / 50.6 hp at 5500 rpm		69 kW / 93.8 hp at 5500 rpm	
	Cruising:	53 kW / 71 hp at 4800 rpm		44.6 kW / 59.8 hp at 4800 rpm	
Engine speed	Max. Take-off:	5800 rpm, max. 5 min.			
	Max. Continuous:	5500 rpm			
	Cruising:	4800 rpm			
	Idling:	~1400 rpm			
Cylinder head temperature:	Minimum:	60 °C	*140 °F*	60 °C	*140 °F*
	Maximum:	150 °C	*302 °F*	135 °C	*275 °F*
Oil temperature	Minimum:	50 °C	*122 °F*	50 °C	*122 °F*
	Maximum:	140 °C	*284 °F*	130 °C	*266 °F*
	Optimum:	90 °C - 110 °C	*194 - 230°F*	90 °C - 110 °C	*194 - 230°F*
Oil pressure:	Minimum:	7,0 bar			
	Maximum:	1,5 bar			
	Optimum:	1,5-4,0 bar			
Fuel:		see 2.13			
Propeller and Manufacturer		V 230C VZLÚ Praha, Czech Republic			
Type:		two blade fixed wooden propeller			
Propeller diameter:		1625 $^{+2}_{-3}$ mm		*63.98$^{+0.008}_{-0.01}$ in*	
Propeller pitch:		18°20´ - 18°55´			

WARNING
The Rotax 912 UL has not been certified as an aircraft engine and its failure may occur at any time. The pilot is fully responsible for consequences of such a failure.

Fuel

- automotive premium grade gasoline, leaded, accord. to DIN 516000,Ö-NORM C 1103
- EUROSUPER RON 95 unleaded accord. to DIN 51607,Ö-NORM 1100
- AVGAS 100 LL
  Due to higher lead content in AVGAS, the wear of valve seats and deposits in the combustion chamber will increase. Therefore, use AVGAS only if you encounter problems with vapor lock or if other fuel types are not available.
- Refer to the Engine Operator's Manual and Service Information for more fuel brands

Oil

Automotive engine oil of a registered brand with gear additives, but not aircraft oil (refer to Engine Operator's and Manual Service Information). API classification "SF" or "SG".
Refer to para 4.6.1 and the Engine Operator's Manual and Service Information.

2.3.12.3 Propeller

2.3.12.3.1 Description

The standard powerplant consists of the four-cylinder, 4 stroke ROTAX 912 (80 hp) engine and V 230 C prop.
The V 230 C is a fixed, wooden, 2-blade propeller with leading edge protection. The prop is attached to the propeller hub with 6 bolts. A fiberglass spinner is used.
Refer to the manuals supplied with the prop for more information.

Propeller Technical Data

Diameter	$63.98^{+0.008}_{-0.01}$	*in*	1625^{+2}_{-3} mm
Pitch.....	18°20´- 18°55´		
Weight	*7.1±0.7*	*lbs*	3.2±0.3 kg
Propeller blade clearance measured from ground:.....	*12.6±1.2*	*in*	320±30 mm

Manufacturer ..... VZLÚ Praha - Letňany, Czech Republic

NOTE

The exact pitch/performance of the prop supplied with each aircraft may differ slightly, therefore the exact performance of your aircraft may be different.

2.3.12.3.2 Propeller pitch adjustment

Intentionally left blank.

2.3.12.4 The FLYdat engine instrument

The Rotax 912 engine parameters are monitored by the integrated digital instrument (FLYdat) (standard equipment).

The following parameters are displayed:

- Engine speed
- Engine hours
- Exhaust gases temperature
- Cylinder head temperature
- Oil temperature
- Oil pressure
- Fuel pressure (if the scanner and gauge are installed)
- Overrun of data limits

The permitted operational limits are shown in the following table:

Function		Minimum Limit	Normal Operating	Caution Range	Maximum Range
Engine speed [RPM]		1400	1400-5500	5500-5800	5800
Cylinder Head Temp. (CHT)	R 912 UL (80 hp)	60 °C *140 °F*	60-100 °C *140-212 °F*	100-150 °C *212-302 °F*	150 °C *302 °F*
	R 912 S (100 hp)			100-135 °C *212-275 °F*	135 °C *275 °F*
Exhaust Gases Temp. (EGT)					880 °C *1616 ° F*
Oil Temp.	R 912 UL (80 hp)	50 °C *122 °F*	90-110 °C *194-230 °F*	50-90 °C *122-194 °F* 110-140 °C *230-284 °F*	140 °C *284 °F*
	R 912 S (100 hp)			50-90 °C *122-194 °F* 110-130 °C *230-266 °F*	130 °C *266 °F*
Oil Pressure		1.5 bar *22 psi*	1.5-4.0 bar *22-58 psi*	4.0-5.0 bar *58-73 psi*	7.0 bar *102 psi* cold engine starting

The limits of operation for the Rotax 912 engine are stored in the FLYdat memory. The values of all limits are shown in the Operator's Manual for the FLYdat instrument. An exceeded limit is indicated by a flashing warning light and is stored in the FLYdat memory for further evaluation. Take care of these limits and do not exceed them. If the operational limits are exceeded, the message “SERVICE,“ will be indicated on the FLYdat display. Contact the nearest Rotax Service Center for assistance.

If analog engine instruments are installed (instead of the FLYdat instrument) then the instruments limit indicators should show the following:

Function		Minimum Limit	Normal Operating	Caution Range	Maximum Range
Engine speed [RPM]		1400	1400-5500	5500-5800	5800
Cylinder Head Temp. (CHT)	R 912 UL (80 hp)	60 °C *140 °F*	60-100 °C *140-212 °F*	100-150 °C *212-302 °F*	150 °C *302 °F*
	R 912 S (100 hp)			100-135 °C *212-275 °F*	135 °C *275 °F*
Exhaust Gases Temp. (EGT)					880 °C *1616 ° F*
Oil Temp.	R 912 UL (80 hp)	50 °C *122 °F*	90-110 °C *194-230 °F*	50-90 °C *122-194 °F* 110-140 °C *230-284 °F*	140 °C *284 °F*
	R 912 S (100 hp)			50-90 °C *122-194 °F* 110-130 °C *230-266 °F*	130 °C *266 °F*
Oil Pressure		1.5 bar *22 psi*	1.5-4.0 bar *22-58 psi*	4.0-5.0 bar *58-73 psi*	7.0 bar *102 psi* cold engine starting

The following analog powerplant instruments are generally installed:
Tachometer .....Rotax (Cat.No.: 950 141)
Oil Temperature Indicator .....Rotax (Cat.No.: 956 410)
Cylinder Head Temperature Indicator.....Rotax (Cat.No.: 956 410)
Oil Temperature Indicator .....Rotax (Cat.No.: 956 410)
Exhaust Gases Temperature Indicator .....Rotax (Cat.No.: 300 00 03)

2.3.12.5 Engine mount

2.3.12.5.1 Description

The engine mount is welded from chrome-molybdenum tubes and is attached to the firewall with 4 bolts. The bed is spring-mounted with four rubber silentblocks.

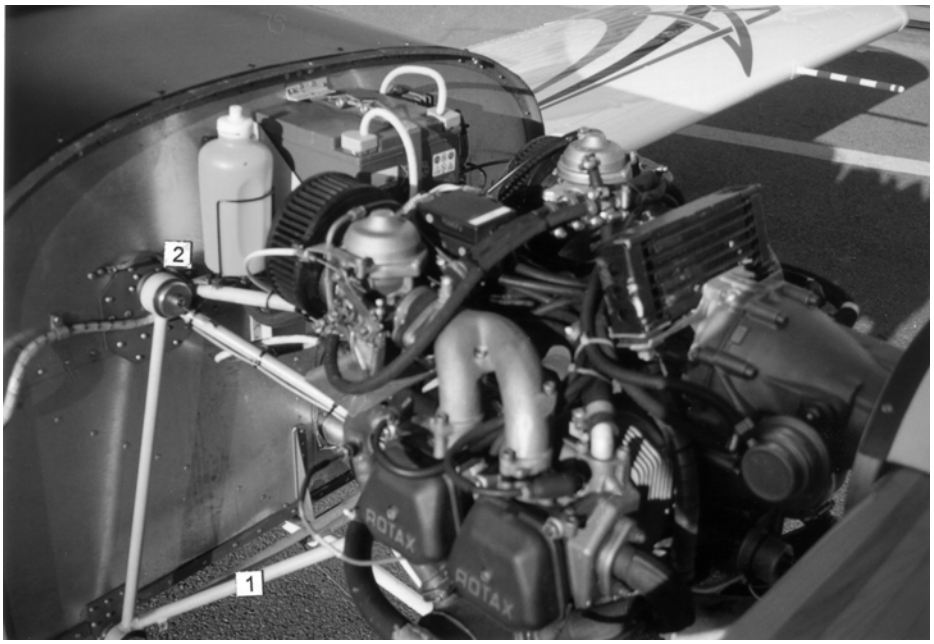
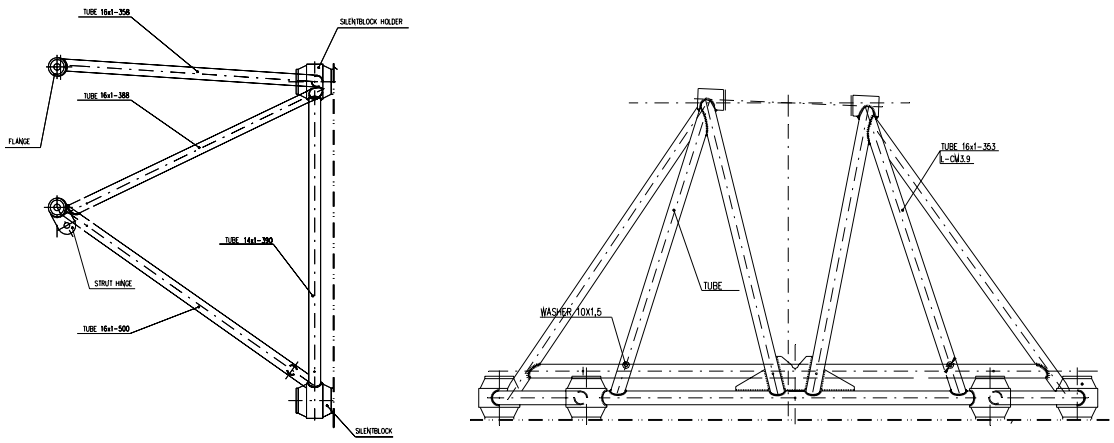


Fig. Engine mount
1- bed, *2* - rubber silentblock

2.3.12.5.2 Engine mount diagram



2.3.12.6 Engine cowlings

2.3.12.6.1 Description

There are two fiberglass cowlings (upper and lower) attached to the engine mount which cover the engine. The interior sides of the cowlings are protected with fireproof paint, exterior sides are painted with aircraft paint. The upper cowling is connected to the lower cowling with quick-closing locks and is usually removed for preflight inspections.
Removed engine cowlings are shown in the following figure.

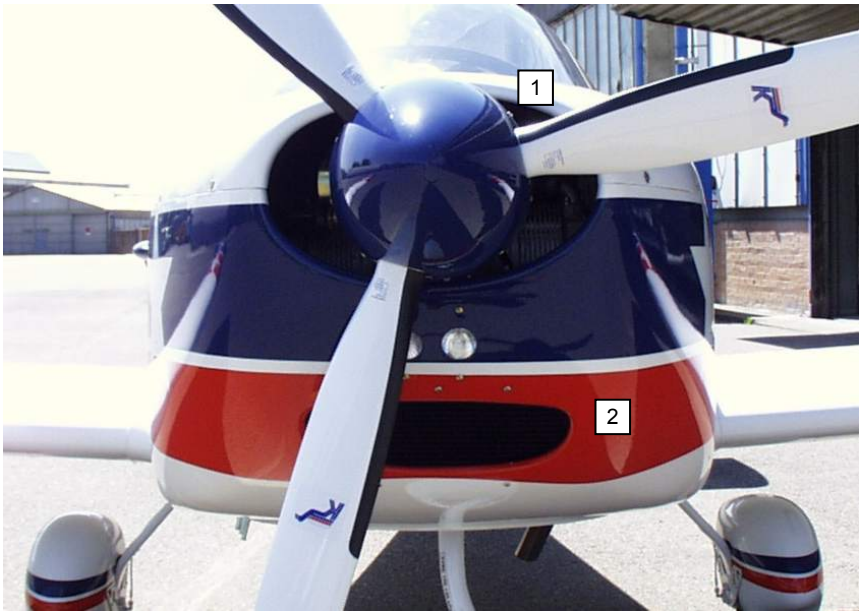


Fig. Rotax 912 engine cowlings
1- upper cowling, *2* - lower cowling with cut-out for the radiator

2.3.12.6.2 Engine cowlings disassembly and assembly

- The upper cowling: The disassembly and assembly are both easy -just release the quick-closing locks (or screws). The upper cowling is usually removed during engine pre-flight inspection to check the engine compartment, operating fluids quantity (oil, coolant) and to check the engine installation.
- The lower cowling: To remove it, un-screw the attachment screws connecting the cooler to the cowling face side, then un-screw the attachment screws connecting the cowling to the firewall border.

It is highly recommended to protect the removed cowlings so as to prevent them from inadvertent damage.
The cowling assembly is the reverse of disassembly.

2.3.13 Fuel system

2.3.13.1 Fuel system description

The standard fuel tank volume is *17.2 USGAL* (65 liter). The tank is located behind the seat backrests. Fuel is pulled from the fuel tank through the fuel valve located inside the cockpit on the left-hand side, below the instrument panel. Then through the fuel filter to the engine fuel pump and on to carburetor.
The fuel tank is equipped with a drain pocket and drain valve. The draining outlet is located under the fuselage. The drain valve is located near the right flap root and accessible when the flaps are fully extended. The fuel tank filler neck is placed on the right-hand (starboard) side of fuselage near the rear canopy.
Fuel quantity is indicated by an electric fuel gauge.

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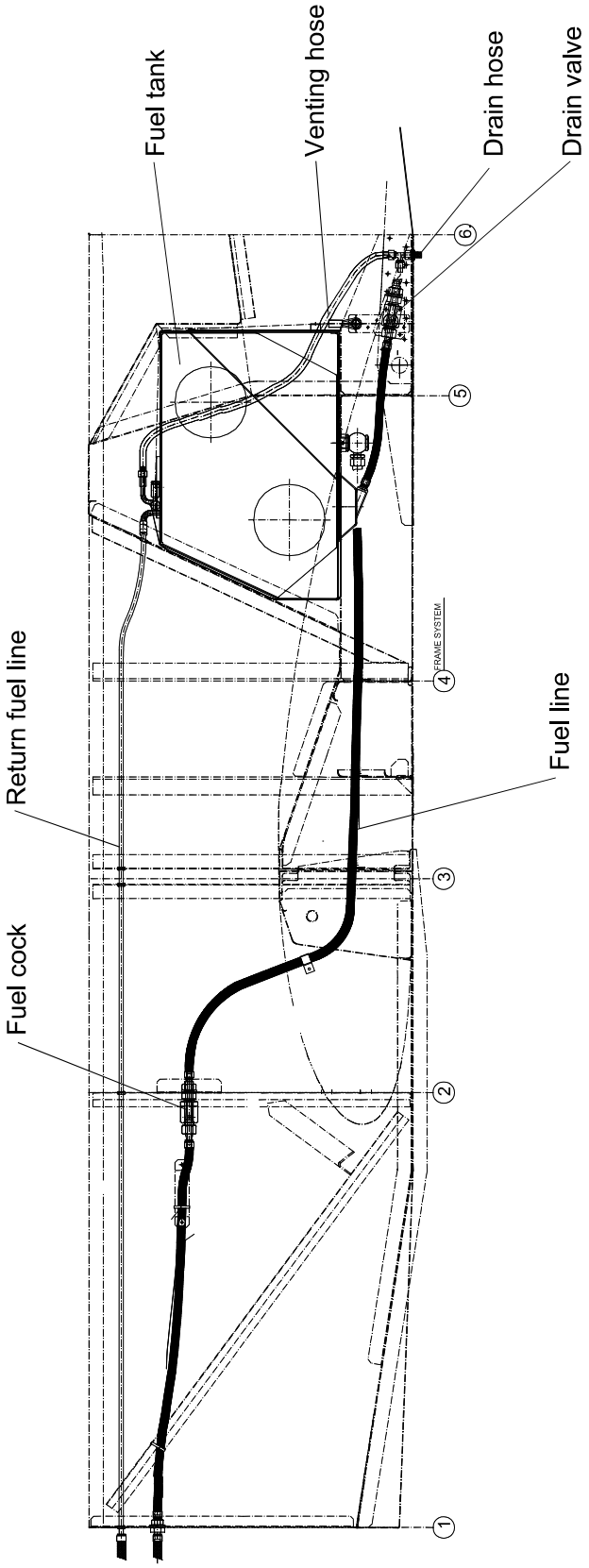
2.3.13.2 Standard fuel system layout



2.3.13.3 Fuel tank draining

The objective is to drain any water and/or contaminates in the fuel tank.

Draining procedure:

1. Place a suitable bottle or transparent cup below the draining outlet hose (near the right-hand flap root)
2. Open the fuel drain valve
3. Drain off a small quantity of fuel
4. Close the drain cock
5. Repete untill no water or contaminants are visable

WARNING
Do not smoke or have open any flame during draining!

Fig. Fuel tank draining
1- drain valve near the right-hand flap root, *2* - draining hose outlet under the fuselage

2.3.14 Engine lubrication system scheme

2.3.14.1 Lubrication system description

The Rotax 912 engine is provided with a dry sump forced lubrication system. The oil pump pulls the motor oil from the oil tank attached to the firewall via the oil cooler. Then forces it through the oil filter to the lubrication points in the engine.
The surplus oil emerging from the lubrication points accumulates on the bottom of the crankcase and is forced back to the oil tank by the blow-by gases.
The oil tank is equipped with a vent hose.
The engine lubrication system is further described in documentation supplied with the engine.

2.3.15 Cooling system description

2.3.15.1 Cooling system description

The cooling system uses two forms of cooling. The cylinder heads are liquid cooled and the cylinders ram air cooled. The radiator is located in the front of the lower engine cowling. The coolant is forced through the radiator by a water pump, driven from the crankshaft to the cylinder heads. From the top of the cylinder heads the coolant passes on to the expansion tank which allows for coolant expansion. The expansion tank is closed by a pressure cap with an excess pressure valve and return valve. When the temperature rises the coolant creates excess pressure, a relief valve opens and the coolant flows through a thin hose to the overflow bottle mounted on the firewall.

The engine cooling system is more completely described in documentation supplied with the engine.

2.3.16 Heating

A cockpit heating system is optional.

2.3.16.1 Description

Air is preheated by the passage through a water cooler. Then an air collector leads the preheated air to the heat exhanger in the exhaust system, then through the valve at the firewall directly to the cockpit. The heating valve is operated with a cable from the heating valveflap to a small push/pull handle located on the instrument panel. Pull the handle to open the heating valve and bring hot air into the cockpit.

The cockpit heating system is shown in the following figure:

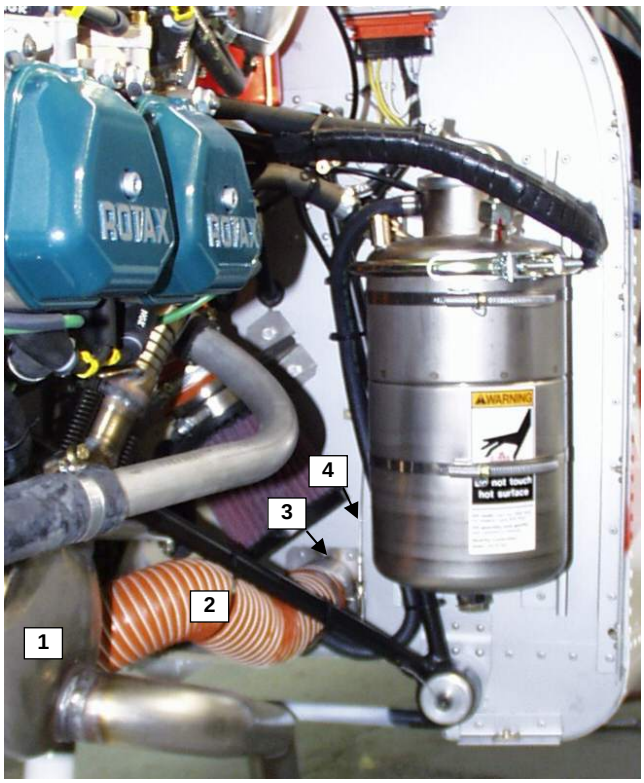


Fig. Cockpit heating system
1- muffler, *2* - air hose,
3 - on the firewall mounted heating flap, *4* - heating valve control cable

2.3.17 Ventilation

2.3.17.1 Description

Canopy vent windows:

- The windshield air window - turn the window to open two air holes. It is highly recommended to close the window while the aircraft is parked to prevent water seeping onto the instrument panel.

- The side sliding window (standard on the left-hand side of the canopy, optional on the right-hand side) with a vent air flap. This window may be equipped with a lock. The nut should be tightened slightly from time to time to prevent the air flap from accidently opening at high airspeeds.

Fig. Windshield air window

Fig. Side sliding window
1 - window, *2* - vent air flap,
3 - window lock

2.3.18 Wiring

2.3.18.1 Wiring description

The electric system is a single-wire type with the negative side connected to the chassis. The power source is a single-phase generator integral to the engine. A 12V/16Ah maintenancefree battery is located on the firewall. The system is protected by the main 25 Amp circuit breaker or fuse. Separate appliances have separate switches. The circuits of the particular sections are each guarded individualy by fuses.
The dual engine ignition is a separate part of the electrical system. Each ignition circuit has its own on/off switch.

2.3.18.2 Wiring diagram

The wiring system will vary and depends on the instrumentation, electronic equipment, and electric acessories of your aircraft.

2.3.18.3 Fuses

The fuse box is generally located on the right-hand side of the instrument panel.

(example)

Fuse No.10	Unused
Fuse No.9	Unused
Fuse No.8	Unused
Fuse No.7	Unused
Fuse No.6	Landing light 10.0 A
Fuse No.5	Fuel gauge 1.0 A
Fuse No.4	Main fuse 25.0 A
Fuse No.3	Instruments 1.0A
Fuse No.2	RPM indicator 1.0 A
Fuse No.1	Generator 40.0 A

2.3.19 Pitot-static system

2.3.19.1 Pitot-static system description

The Pitot-static tube, located under the left wing near the aileron root, provides both dynamic and static air pressure.
Pressure distribution to individual instruments is done through flexible plastic hoses.
Keep the system clear to ensure its correct function.
Both hose systems (Dynamic and static) are equipped with dirt pockets. The dirt pockets are located inside the cockpit in front of the pilot's seat.
If water gets inside the system, unscrew the covers from the dirt pockets and slightly blow into the Pitot-static head. Then screw the covers back and check the packings.

CAUTION
Avoid blowing into the Pitot-static system with the dirt pocket cover is closed - it may cause an instrument malfunction.

Fig. Pitot-static tube under the left wing

2.3.20 Placards

A new aircraft is equipped with placards supplied by the airplane manufacturer.
These placards explain the purpose of controls, instruments, airspeed limits, weight limits, etc. Placards with supplemetal information such as a direction of handles are also supplied.
The placards are usually attached to the appropriate instruments and controls. Limitation placards are attached to the canopy, external placards are attached on the appropriate aircraft part, however placards may vary slightly from plane to plane.

CAUTION
The owner (aircraft operating agency) of the aircraft is responsible for the readability of placards during the aircraft service life.

In case of placard damage or unreadibility, it is permissable to copy placards enclosed in the Appendices of this manual (copy on suitable adhesive tape) to replace the damaged placard.

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3. OPERATION

3.1 Operation outlines

During operation of the ***EV-97 „EUROSTAR“ model 2000 version R*** it is required to have in the following documentation in the plane:

- Airplane Technical Description, Operating, Maintenance and Repair Manual ***EV-97 "EUROSTAR" model 2000 version R***
- Airplane Flight Manual for ***EV-97 "EUROSTAR" model 2000 version R***
- Engine Operator's Manual
- Propeller Operator's Manual
- Additional documents supplied with instruments or equipment

The airworthiness and operational readiness of the airplane depends upon the careful adherence to the recommended procedures and regulations. Climate, aerodrome conditions, dustiness, manner of hangaring and other factors, such as the corrosive effects of industrial or seaside areas, should be considered.
The procedures given in this manual suit average operational conditions, more harsh environments may require more frequent maintenance intervals.

3.2 Airplane assembly

3.2.1 Wing

3.2.1.1 Wing assembly

The wing assembly procedure for an aircraft not-equipped with the optional wing folding mechanism is the following (3 persons are needed to accomplish this task):

3.2.1.1.1 Necessary tools

- a hammer to move the wing suspension pins
- a screwdriver to attach wing fillets
- wrenches to tighten the rear wing suspension bolt nut
- lubricant to preserve the wing suspensions

3.2.1.1.2 Wing-to-fuselage assembly procedure

The assembly procedure of one half of the wing is the following.
The procedure for both halves is similar.

1. Thoroughly clean and lubricate all the wing suspensions and bolts with a suitable lubricant before the assembly. Also lubricate the flap root groove.
2. The first person holds the wing tip, the second person holds the wing root leading edge, while the third holds the wing root trailing edge.

CAUTION
Take care of the pitot-static tube when handling the left wing.

3. The person holding the wing tip lifts it to shoulder level.
4. Carefully place the wing suspensions into the fuselage ones, so that the wing upper suspension hook will insert into the fuselage. Be carefull of the pitot-static hoses on the left wing and the wires of the position lights (if lights are installed).
5. The person holding the wing tip lowers it down to waist level.
6. Adjust the window position so the wing and fuselage suspension pinholes are perfectly coaxial.
7. The person holding the wing root near the leading edge inserts a pin into the lower wing suspension (with the head of the pin in the flight direction) and tap the pin to connect the wing-fuselage suspensions (slightly lift and lower the wing tip to make the pin insertion easier)
8. The person holding the wing root near the leading edge inserts a pin into the upper wing suspension (with the head of the pin in the flight direction) and taps the pin to connect the wing-fuselage suspensions
9. The person near the wing trailing edge sets the cut out in the flap root at the flap control pin (to allow this to slightly move the wing tip forward)
10. Insert and tap the bolt into the rear wing suspension
11. Secure all the pins with safety pins
12. Join the aileron control rod and secure the connections
13. Connect the wires of the position lights and pitot-static hoses
14. Position and attach the wing fillets

3.2.1.2 Wing disassembly

3.2.1.2.1 Necessary tools

- a hammer to tap the wing suspension pins out
- a screwdriver to unscrew the wing fillet connection screws
- wrenches to unscrew the rear wing suspension bolt nut
- a drift made from duralumin round or other suitable material (diameter 10-12 mm) to drive out the wing suspension pins

3.2.1.2.2 Wing-from-fuselage disassembly

1. Be sure to disconnect pitot-static hoses in the cockpit if disassembling the left wing
2. Remove the screws and remove the wing fillets
3. Disconnect the position lights wire connectors (if lights are installed)
4. Disconnect the aileron control rod
5. First person holds the wing tip, the second person holds the wing root leading edge with the third holding the wing root trailing edge.
6. Disconnect the rear wing suspension.
7. Remove the safety pins from the main wing suspensions.
8. Slightly lift the wing tip.
9. Tap the upper and lower wing suspension pins out with a hammer and metalic drift.
10. The person holding the wing tip should lift it to eye level to unhook the upper main wing suspension and then slightly pull the wing away from the fuselage.
11. Place the disassembled wing away the airplane to avoid accidental damage.

3.2.1.3 Wing folding

The aircraft may be equipped with the optional wing folding mechanism. The wing folding is convenient to reduce the parking area required in a hangar and to allow transportation on a suitable trailer.
The wings may be folded in 15 minutes by 2 people.

3.2.1.3.1 Wing folding mechanism scheme

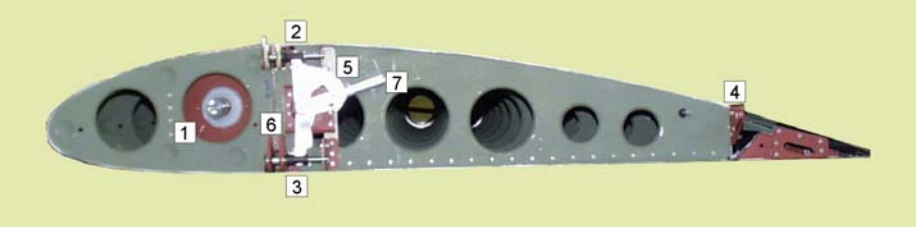

Fig. Wing folding mechanism
1- disconnected aileron rod,
2 - upper wing suspension pin,
3 - lower wing suspension pin,
4 - rear auxiliary wing suspension,
5 - upper pin shift in-out lever,
6 - lower shift in-out lever,
7 - assembly lever supply tube,

3.2.1.3.2 Tools necessary for wing folding

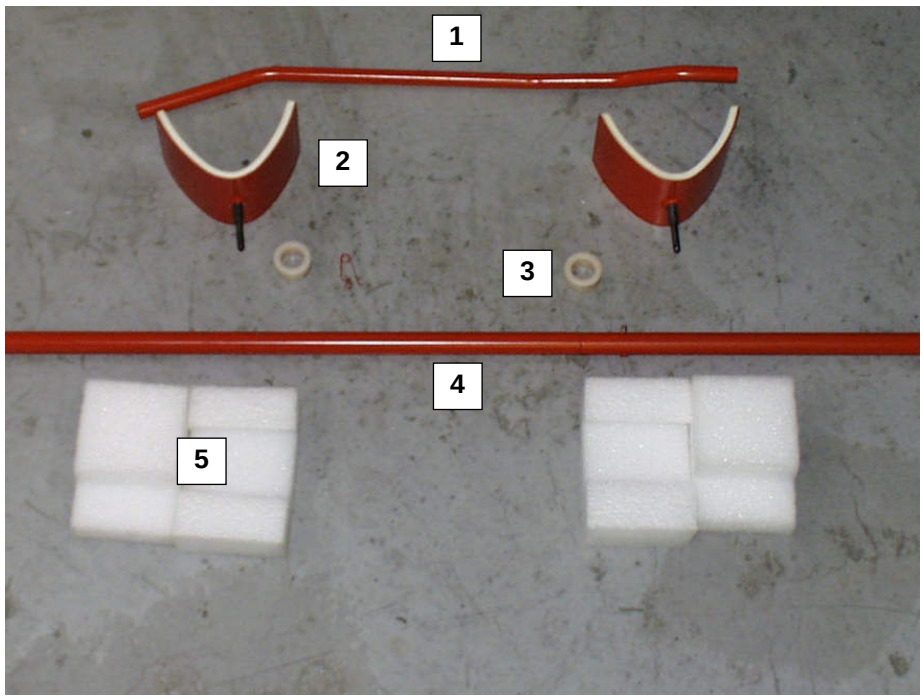

Fig. Tools necessary for wing folding
1- assembly lever to shift in-out wing suspensions pins,
2 - wing rest,
3 – palstic O-ring,
4 – comlete strut,
5 - soft pad

3.2.1.3.3 Wing folding procedure

1. Insert plastic O-ring onto right strut, secure by cotter pin. Insert the right strut into the hole provided in the rear right side of fuselage. Put the strut through the fuselage (the holes in fuselage sides are covered with spring loaded covers). ⇨ see **FIG. 1**
2. Insert plastic O-ring onto left strut, secure by cotter pin. Insert the left strut into the right tube from the left side of fuselage.
3. Insert a retaining pin to connect left and right struts and secure the pin by a safety pin ⇨ see **FIG. 2**
4. Put the wing rests onto both left and right struts, secure by cotter pin. ⇨ see **FIG. 3**
5. Set the flaps to the "LANDING" position
6. Unscrew and remove the wing/fuselage fillets⇨ see **FIG. 4**
7. Disconnect the aileron control rods in the space between the wing and fuselage ⇨ see **FIG. 4**
8. Release the safety pin securing the rear pin nut ⇨ see **FIG. 5**
9. Release the safety pin securing the main lower pin
10. Release and remove wing folding mechanism securing wire ⇨ see **FIG. 6**
11. An assistant should hold the wing tip and slightly lift the wing ⇨ see **FIG. 7**
12. Unscrew the rear pin nut and push the pin out of the hinge
13. Put the assembly lever on the wing folding mechanism lock⇨ see **FIG. 7**
14. Push the assembly lever down to push-out the pins from the upper and rear wing suspensions
15. Push the assembly lever forward and then lift it to push-out the pin from the lower wing suspension
16. Pull the assembly lever out
17. The assistant should lift the wing tip up to eye level ⇨ see **FIG. 8**
18. The assistant, with the assistance of a second person, may run the wing slightly out the fuselage⇨ see **FIG. 8**
19. The assistant holding the wing tip should turn the wing vertically (L.E. down) - (Use caution with the pitot-static tube when folding the left wing) ⇨ see **FIG. 9**
20. The assistant gradually folds the wing towards to the fuselage - using care not to damage the wing leading edge during folding ⇨ see **FIG. 10**
21. Insert a soft pad between the leading edge and landing gear leg ⇨ see **FIG. 11**
22. The assistant next lifts the wing on the wing rest⇨
23. Lock the wing using the rubber rope and wing anchor hinge see **FIG. 12**
24. Secure the flap in the open position – by the self adhesive tape, for instance
25. Use the same procedure to fold the other half of the wing

see the Figures on the next pages for the Wing folding procedure

NOTE

Wing folding is a procedure intnded to reduce the required parking place. It is not recomended to transport the aircraft over long distances when the wings are folded. Be carefull when transporting the folded wing aircraft along grassy and rough roads.

3.2.1.3.4 Wing reassemble procedure

1. An assistant holds the wing tip
2. Another person removes the safety pin securing the wing on the wing rest hinge
3. The assistant removes the wing from the rest (using caution with the pitot-static tube under left wing)
4. The assistant lifts the wing off the fuselage rear
5. The assistant turns the wing to the horizontal position
6. The assistant lifts the wing tip up to the eye level
7. The assistant pushes the wing towards the fuselage while the second person guides wing suspensions onto the fuselage. The small hook of the wing suspension must fit in the groove of the fuselage upper main suspension
8. After insertion the assistant lowers the wing tip
9. The second person connects the flap control
10. Put the assembly lever on the wing folding mechanism supply tube
11. Push the assembly lever down to move the pin into the lower wing suspension. Check the connection (the assistant slightly moves the wing tip forward and backward to make the pin movement easier)
12. After moving the lower pin, the assembly lever springs back
13. Lift the assembly lever up and check proper location of the upper pin
14. Push the assembly lever forward, insert the wing folding mechanism securing wire and lock it (put the wire end through the wing cover)
15. Secure the main lower pin with a cotter pin
16. Put the rear pin in the wing hinge
17. Screw the nut on the rear pin. Secure the nut.
18. Join the aileron control rod and secure the connection
19. Check all of the wing suspensions for proper attachment. Inspect proper locking of the wing folding mechanism
20. Locate and attach wing fillets using screws removed
21. Use the same procedure for the other half of the wing
22. Remove the pin connecting the left and right wing rests
23. Remove wing rests from the fuselage

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3.2.2 Horizontal tail unit

3.2.2.1 HTU-from-fuselage disassembly

1. Unscrew attachment bolts of HTU-fuselage fiberglass cover.
2. Disconnect the trim tab control cables.
3. Disconnect the elevator control rod.
4. Remove the safety pins securing the castle nuts on the bolts of the stabilizer rear suspensions. Screw off the nuts and remove the washers.
5. Draw the HTU out of the fuselage.
6. Put connecting components in a safe place to avoid loosing them .

3.2.2.2 HTU-to-fuselage assembly

3.2.2.2.1 Necessary tools

- a wrench to tighten M8 nuts
- a screwdriver to attach the tail unit/fuselage fairing

3.2.2.2.2 HTU-to-fuselage assembly

1. Make the connecting components ready, clean and lubricate HTU suspensions.
2. Insert the HTU from the rear into the fuselage as far as the stabilizer will go into the two pins in the front, and the two bolts (M8) in the rear. Take care of the trim tab control cables.
3. Put the washers on the M8 bolts. Screw on the castle nuts, and secure with a safety pins.
4. Attach the HTU/fuselage fairing using screws.
5. Insert the M8 bolt to connect the elevator control hinge with the control rod. Put on a washer, and self-locking nut.
6. Connect trim tab control cables.
7. Adjust the elevator and trim tab deflections (see 3.4.6)

3.2.3 Vertical tail unit

3.2.3.1 Assembly and Disassembly of the rudder

3.2.3.1.1 Necessary tools

- a wrench to tighten/remove the M5 nut

3.2.3.1.2 Rudder-from-fuselage disassembly

1. Disconnect the rudder control cables, attach the ends of the cables together to keep the cables from sliping inside the fuselage.
2. Remove the safety pin from the lower suspension bolt. Remove the castle nut and washer.
3. Lift and remove the rudder from suspensions

3.2.3.1.3 Assembly procedure

1. Put the rudder on the fin suspensions from above. Use care not to move the spherical bearings in the rudder suspensions.
2. If necessary insert a washer to adjust lower suspension clearance.
3. Put the washer on the lower suspension bolt, tighten the castle nut and secure with a safety pin.
4. Attach the rudder control cables. Use tab washers to secure bolt heads.

3.2.4 Landing gear

3.2.4.1 Tire replacement

1. Support the aircraft to lift a wheel with punctured tire.
   Main wheel:
   It is recommended to support the aircraft near the main gear leg entry into the fuselage (a reinforced area)
   Nose wheel:
   a) Push the rear of the fuselage down and support the aircraft under the nosewheel leg-fuselage attachment or
   b) Load the fuselage rear near the fin (bags with a load) to lift the nosewheel above the ground
2. Remove the wheel pant if installed
3. Deflate tire
4. Remove cotter pin securing castle nut from the axle
5. Remove the castle nut and washer from the axle
6. Cut the safety wire securing the bolts which join together the rim halves
7. Remove the 3 bolts connecting the rims together
8. Remove outer rim from the axle
9. Remove tire
10. Replace the tire and/or tube
11. Insert a new tube into a tire and pump up slightly(put easy powder on the tube and/or in the tire)
12. Put the valve insert of the tyre with tube into the hole on the rim
13. Put the rim with tube and tire onto the axle and adjust the wheel hub
14. Insert the 6 connecting screws and slightly tighten
15. Adjust the tire and tube positions to avoid puncturing the tube
16. Tighten the screws
17. Put the washer on the axle
18. Screw the castle nut on the axle
19. Insert a new cotter pin and lock the castle nut
20. Secure the screw heads with safety wire – pull the safety wire through the holes in two screw heads and twist together.
21. Remount the wheel pant

3.2.4.2 Assembly and disassembly of wheel pants

An airplane may be equipped with the optional aerodynamic fiberglass wheel pants.

3.2.4.2.1 Main landing gear wheel pants

The main wheel pant is attached with 8 screws to the main leg from the inner side and with 1 screw from the outer side. The removal and replacement is easy and straightforward.Remove/replace the screws to remove/replace the wheel pant.

3.2.4.2.2 Nosewheel pant

Nose wheel pant is attached to the landing gear leg by means of bracket by 8 bolts .It is also attached to the wheel axis by means of two bolts on the sides.

Fig. Main wheel pants

Fig. Nosewheel pant

3.2.4.3 Removal and replacement of mudguards

The aircraft may be equipped with optional mudguards to prevent wing lower surface pollution.

3.2.4.3.1 Main landing gear wheel mudguard

Demounting

1. Support the airplane to lift the main wheel with the mudguard to be removed
2. Cut the safety wire securing the screw that attaches the mudguard to the axle (1 screw) and the safety wires securing the screws at internal leg side (two screws)
3. Remove the screw attaching the mudguard to the axle
4. Remove the two screws attaching the mudguard to the main landing gear leg from the inner side
5. Remove washers
6. Remove the mudguard

Mounting

Use the following procedure to mount a mudguard on an airplane not equipped with mudguards by the manufacturer (use the opposite procedure to the demounting one (see above) to mount a mudguard back on a main leg)

1. Support the airplane to lift a main wheel on which a mudguard would be mounted on
2. Remove the cotter pin securing the castle nut on the wheel axle
3. Remove the castle nut and washer
4. Put the new castle nut, supplied with mudguards, on the wheel axle, tighten
5. Secure the nut with a cotter pin
6. Mount a mudguard on the wheel
7. Attach the mudguard to the main leg with two screws (use washers), slightly tighten.
8. Use the screw and washer to attach the mudguard to the castle nut on the axle, slightly tighten
9. Adjust mudguard position
10. Tighten all the screws
11. Secure all screws with safety wires
12. Remove the airplane from supports

3.2.4.3.2 Nose- wheel mudguard

1. Lift and support aircraft middle section
2. Dismount and remove the nosewheel axle
3. Remove the nosewheel
4. Unscrew the screws attaching the mudguard to the nosewheel fork

3.2.5 Cockpit canopy

3.2.5.1 Canopy demounting

The front portion of the canopy can be removed, while the rear portion is fixed. The front section of the canopy is attached to the fuselage with two screws. The weight of canopy is counter-balanced with two gas struts inside the cockpit. Use the following procedure to remove the front canopy:

1. Unscrew and remove canopy cover sheets (this cover has an "ear" shape)
2. Lift the canopy
3. Remove the screws from the console that holds the gas strut ball ends at the canopy frame
4. Carefully remove the canopy attachment screws. Caution: The front canopy hinges, and the canopy should be supported
5. Remove the canopy

3.2.5.2 Canopy mounting

The Mounting procedure is the reverse.

3.2.6 Installation and reinstallation of instruments

The installation procedure will depend on the instrument being installed. Follow the manufactors recommendations.
Ordinarily, there is no need to remove the instrument panel when installing or removing an instrument. Remove the instrument attaching screws and remove the instrument from the back of the instrument panel (after disconnection of appropriate wires or hoses). If it is necessary to gain access to the instrument wiring, remove the sheet cover over the instrument panel.

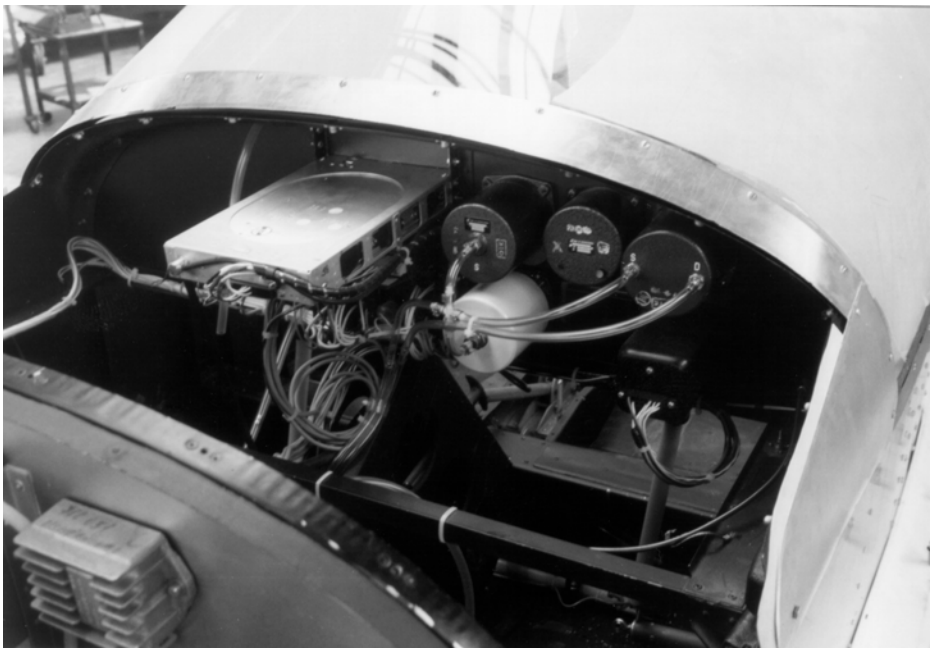


Fig. Access to the instruments mounted on the instrument panel after removal of the cover

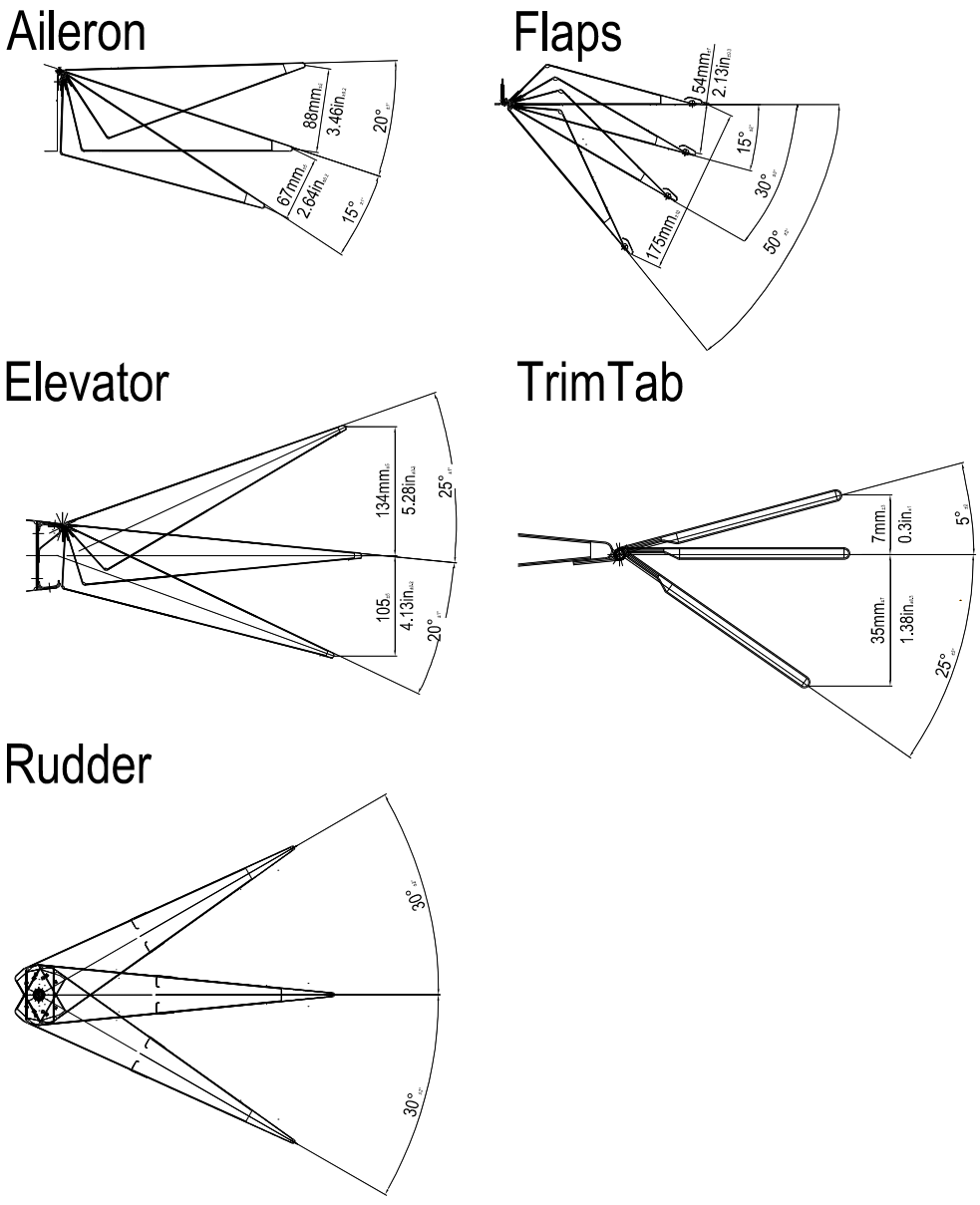
3.3 Leveling

Leveling is used to check the airframe alignment.
First set the aircraft in a horizontal position (use boards) according to leveling points. The leveling points are the rivets on the aircraft which are (generaly) marked with red paint. The location of the points is shown in the Leveling Record. Use the leveling points 1(3) and 2(4) to set the airplane in a horizontal position in longitudinal direction, and leveling points 5 and 7 in lateral direction.
Boards, under the main or nosewheel, may be used to level the airplane. The best way to measure a leveling point height is to use a level. Alternatively a running meter is sufficient for approximate measurement.
Measured values should be recorded in the Leveling Record (see Appendices). Height differences between corresponding leveling points have to be calculated. A check must than be carried out to prove that any differences do not exceed the tolerances permitted in the Leveling Record.
If any difference exceeds the permitted tolerance, the aircraft assembly, plays in hinges and eventual permanent deformations, should be inspected.
The aircraft manufacturer should be contacted in serious cases.

3.4 Measurement of control surfaces deflections

3.4.1 Required deflections

The deflection of the control surfaces are specified in the Control Surfaces Deflection Record (see Appendices of this Manual) and in the following Figure. A protractor with deflecting hand is used by the airplane manufacturer to measure deflections. The protractor is attached to a control surface with a hand clamp. There are also alternative procedures in the following text.



3.4.2 Aileron deflection measurement

Measurement procedure:

1. Attach a protractor with a deflection hand at the aileron upper surface by means of a clamp.
2. Set the aileron in neutral position (the aileron must fit the wing profile).
3. Zero the protractor - starting position for measurement .
4. Deflect the aileron fully down/up and note the deflections.
5. Compare, the measured deflections with the ones specified in the Control Surfaces Deflection Record. If required - adjust the aileron deflection according to par. 4.11.1 .

If a protractor is not available, the following procedure may be substituted:

1. Insert a stiff cardboard sheet of paper in the space between the aileron and the flap. Hold the drawing paper against the flap.
2. Put the aileron in a neutral position. Then trace its profile (upper surface from the hinge to the trailing edge). Then trace the profile of the aileron fully deflected in both directions.
3. Remove the drawing paper and measure the deflection from the neutral position using a protractor.
4. Compare the measured deflections with those specified in the Control Surfaces Deflection Record. If required - adjust the aileron deflection according to 4.11.1.

3.4.3 Flap deflection measurement

The wing flaps can be set in 4 positions: RETRACTED, TAKEOFF, LANDING (2 positions).
Measurement procedure:

1. Cut a strip of aluminium sheet 2 inches (50 mm) wide. The strip is than attached to the flap lower surface with two bolts (somewhere in the middle of the flap span where two nuts are riveted). The strip should overhang the flap trailing edge.
2. Attach a protractor with a deflection hand at the strip using a clamp.
3. Zero the protractor - this will be the starting position for measurement with the flaps retracted
4. Extend the flap to the required position using the flap control lever and read the deflection
5. Compare the measured deflections with those specified in the Control Surfaces Deflection Record. If required - adjust flap deflection according to the par. 4.11.2.

If a protractor is not available, the following procedure may be substituted:

1. Insert a stiff cardboard sheet of paper in the space between the aileron and the flap. Hold the drawing paper against the flap.
2. Trace the profile of the retracted flap on the lower surface from the hinge to the flap trailing edge)
3. Move the flap to an extended position and trace the lower surface profile again
4. Remove the drawing paper and measure the deflection from the "RETRACTED" position using a protractor
5. Compare the measured deflections with those specified in the Control Surfaces Deflection Record. If required - adjust flap deflection according to the par. 4.11.2.

3.4.4 Elevator deflections measurement

The starting position to measure the elevator deflections is the neutral positions of the control stick and elevator. The neutral position of the control stick is set by the aircraft manufacturer, by means of a jig. The distances between the control stick and instrument panel, and between the control stick and fuselage side can be used to set the neutral position. When the elevator is in the neutral position, the chord of the Horizontal tail unit will be parallel to upper edge of the fuselage side (lower frame of the cockpit).

Measurement procedure:

1. Attach a protractor with a deflection hand at the elevator trailing edge
2. Set the elevator to the neutral position
3. Zero the protractor
4. Fully pull or push the control stick to deflect the elevator and read the deflection
5. Compare the measured deflections with those specified in the Control Surfaces Deflection Record. If required - adjust elevator deflection according to the par. 4.11.3.

If a protractor is not available, the following procedure may be substituted:

1. Support the airplane under the tail skid and firewall and set the airplane in a horizontal position (a level set on the canopy lower frame can be used to set the airplane in horizontal position)
2. Stand a suitable staff close to the elevator trailing edge and mark the neutral position of the elevator.
3. Move the control stick and fully deflect the elevator. Mark the positions of the elevator while fully deflected
4. Measure the distances between marks on the staff
5. Compare the distances with those specified in Fig. 3.4.1. If required - adjust elevator deflection according to the par. 4.11.3.

3.4.5 Rudder deflection measurement

The rudder deflections are set by the aircraft manufacturer and cannot be adjusted or changed.
A measuring instrument is used by the aircraft manufacturer to measure the rudder deflections. The instrument is put on the vertical tail unit and a rudder deflection may be read directly.
The rudder deflection may be measured however, when the set the rudder is set to the neutral position. Stand a suitable staff at the ruder trailing edge and mark lower edge of the rudder. Fully deflect the rudder and measure using a ruler between the mark on the staff and the lower edge of the rudder. Compare the measured distance with that specified in 3.4.1.

3.4.6 *Trim tab deflections measurement*

The trim tab deflection is measured from the neutral position. When the trim tab profile does not protrude from the elevator profile with elevator set in neutral position.
Measurement procedure:

1. Attach a protractor with a deflection hand at the trim tab
2. Neutralize the trim tab and the elevator
3. Zero the protractor
4. Set the trim tab in maximum lower or upper position using the trim tab control lever and read the deflection from the protractor scale.
5. Compare the deflection with that specified in the Control Surfaces Deflection Record. If required - adjust trim tab deflection according to the par. 4.11.5.
6. Check tension of trim tab control cables according to the following procedures:
   Block elevator against to movement and trim tab control lever set to the neutral position. Apply a load of 20 N (preferably according to a dynamometer) to the trim tab trailing edge. The trim tab deflection musn't exceed value of 5^{+2} mm from the original position. If the trim tab deflection exceeds this value, then it is neccessary to adjust trim tab cable preload by adjusting screws.

If a protractor is not available, the following procedure may be substituted:

1. Insert a stiff cardboard sheet of paper in the space between the elevator and the trim tab and hold the cardboard against the elevator.
2. Trace the profile of the neutralized trim tab
3. Move the trim tab to the maximum (both directions) using the trim tab control lever and trace the profile again
4. Remove the carboard and measure the deflection from the neutral position using a protractor
5. Compare the deflection with that specified in the Control Surfaces Deflection Record. If required - adjust trim tab deflections according to the par. 4.11.5.
6. Check tension of trim tab control cables according to the following procedures:
   Block elevator against to movement and trim tab control lever set to the neutral position. Apply a load of 20 N (preferably according to a dynamometer) to the trim tab trailing edge. The trim tab deflection musn't exceed value of 5^{+2} mm from the original position. If the trim tab deflection exceeds this value, then it is neccessary to adjust trim tab cable preload by adjusting screws.

3.5 Permissible Tolerances

The following table indicates the permissible tolerances for critical parts of the airplane. These values should not be exceeded in operation.
It is expected that an operator will take steps if excessive plays are found on/in part not listed below.

System	Procedure to find a play	Procedure to remedy a play	Max. product. play	Max. operat. play
Ailerons control system	Block ailerons up to the wing and move the control stick to the left and right	Check condition of bearings and replace if needed	*0.08 in* 2 mm	*0.2 in* 5 mm
Elevator control system	Block elevator up to the stabilizer, pull and push the control	Check condition of bearings and replace if needed	*0.08 in* 2 mm	*0.2 in* 5 mm
Flaps control system	Set the flaps in all position by degrees and then handle the flap trailing edge near the flap root, move the trailing edge up/downward to find possible plays	Check the part with oval hole for the control pin in the flap root rib and replace the worn-out pin or the part with oval hole.	*0.08 in* 2 mm	*0.2 in* 5 mm
Trim tab control system	Block the tab up to the elevator, move the trim tab control lever to find a play in a control system	Check cable tension	*0.08 in* 2 mm	*0.2 in* 5 mm
Wing-Fuselage attachment	Move the wing tip and note play in wing suspensions	Check wing suspensions, replace pins	0	*0.08 in* 2 mm
HTU attachment	Move the stabilizer tip forward-rearward	Replace bearings in suspension points and bearings in control system	0	*0.08 in* 2 mm
Rudder hinges	Lift the rudder	Change swivel bearing or insert a washer under the lower hinge pin	*0.04 in* 1 mm	*0.08 in* 2 mm
Nose wheel	Push the rear part of the fuselage down (use a weight) to lift the nosewheel, then move the wheel forward- rearward	Remove the wheel, remove the rim and tire and replace the bearings	*0.04 in* 1 mm	*0.12 in* 3 mm
Main landing gear	Lift the wing tip (hold the wing under the main spar) to lift a main leg, then move the wheel forward-rearward and note play in bearings or leg attachment	Check the leg attachment, wheels attachment, replace the bearings, if necessary	*0.04 in* 1 mm	*0.12 in* 3 mm

3.6 Weighing the airplane and C.G. calculation

WARNING

Never exceed the maximum takeoff weight and c.g. range for any configuration of crew, fuel and baggage as shown in the flight manual.

The removal or addition of equipment may result in changes to the center of gravity and empty weight of the aircraft. The permissible useful load can also be affected. In such case a new weight and balance is necessary to determine the new empty weight and center-of-gravity position. The new empty weight and C.G. position should be recorded in the Flight Manual, Section 6., Weight and Balance Record / Permitted Payload Range. Then a new permitted crew weight for fueling and baggage must be computed and recorded. The cockpit placard "Load Limits" should also be up-dated.

3.6.1 Empty weight determination

The empty weight of an aircraft includes all operating equipment that has a fixed location and is actually installed in the airplane. It includes the weight of the painted airplane, accumulator, standard and optional equipment, full engine coolant, hydraulic fluid, brake fluid, oil. The aircraft is weighed without crew, fuel and baggage.

The following weighing procedure is recommended:

1. Remove excessive dirt, grease, moisture from the airplane before weighing
2. Weigh the airplane inside a closed building to prevent errors due to wind
3. Place the scales, calibrate zero
4. Place the airplane on the scales (use boards to run on the scales or lift the airplane - see airplane jacking)
5. Place the airplane in a level flight position (use suitable rests under the wheels)
6. Check the configuration for weighing (e.g. empty weight);
7. Weigh the airplane and record the values in Weight and Balance Record (make a copy of standard Record included in section 6 Appendices).
8. Compute the weight and C.G. position according to the formula Weight and Balance Record
9. Compute and record permitted crew weight for fueling and baggage - see Pilot's Operating Manual par. 6.2.
10. Up-date the placard "Load Limits" (make a new one) and attach in the cockpit.

3.6.2 *Operating C.G. Range calculation*

On the basis of knowledge of arms, weights of items, airplane empty weight and the C.G. position it is possible to calculate weight and C.G. position according to below given formula:

Item	**Arm to the Datum (Leading edge) C.G.$_i$**		**Weight W$_i$**	**Moment M$_i$**
	[in]	**[mm]**	***[lbs]*** or **[kg]**	***[lbs.in]*** or **[kg.mm]**
Empty airplane				
Crew	*19.69*	500		
Fuel (*6 lbs/ USGAL*) (0.72 kg/ltr.)	*36.22*	920		
Baggage	*50.00*	1270		
			Total Weight TW=ΣW$_i$ *[lbs]* or [kg]	**Total Moment TM=ΣM$_i$ *[lbs.in]* or [kg.mm]**

C.G. position from Datum (Leading edge):

$$C.G. = \frac{Total\ Moment}{Total\ Weight} = \frac{\qquad\qquad}{\qquad\qquad} = \ldots\ldots\ldots\ldots\ldots [in]\ or\ [mm]$$

C.G.position in % MAC

(MAC ...Mean Aerodynamic Chord = *49.2 in* i.e.1250 mm):

$$\overline{C.G.} = \frac{C.G.}{MAC} \cdot 100 = \frac{\qquad\qquad}{\qquad\qquad} \cdot 100 = \ldots\ldots\ldots\ldots\ldots\ [\%\ \mathrm{MAC}]$$

C.G.range limits

Empty weight C.G. range (standard equipment) ..... 18±2 % MAC
Operating C.G. range..... 20÷34 % MAC

3.7 Ground handling

3.7.1 Towing the airplane

It is easy to tow the airplane a short distance by holding the blade root, since the empty weight of this airplane is relatively low.
Suitable surfaces to hold the aircraft airframe, are the rear part of the fuselage before the fin and the wing roots.
A tow bar may be used to tow the aircraft over long distances. Steerable nose wheel is equipped with the stops, it is impossible to turn it around.

CAUTION

Avoid excessive pressure on the aircraft airframe - especially at the wing tips, elevator, rudder, trim etc.
Handle the propeller by holding the blade root - never the blade tip! When starting the engine manually - always handle the propeller on a blade surface i.e. do not hold only an edge.

3.7.2 Parking the airplane

It is advisable to park the aircraft inside a hangar, or eventually inside a other weather proof space (such as a garage) with stable temperature, good ventilation, low humidity and a dust-free environment.
It is necessary to tie-down the aircraft when parking outside.
On occations when the plane must be tied-down outdoors for extended periods, it is advisable to cover the cockpit canopy, and if possible, the entire aircraft using a suitable cover.
The space requirements in the case of long-term hangaring may be reduced by removing or folding the wings. We recommend removing the battery in winter and storing it at indoor temperature.

3.7.3 Tieing-Down

The airplane is usually tied-down when parked outside a hangar. The tie-down is necessary to protect the aircraft against possible damage caused by wind gusts.
For this reason the aircraft is equipped with tie-down eyes, located on the wing lower surface.
Procedure:

1. Check: Fuel valve off, Circuit breakers and Master switch off, Switch box off.
2. Block the control stick up e.g. by means of safety harness or connect the control stick with rudder pedals by means of a suitable rope.
3. Shut all the ventilation windows
4. Close and lock cockpit
5. Tie-down the aircraft to the ground by means of a mooring rope passed through the eyes, located on the wing lower surface. It is also necessary to tie the nose wheel landing gear and the tail skid to the ground.

NOTE

It is advisable to cover cockpit canopy, if possible the whole airplane, by means of a suitable covering material attached to the airframe for long term outside parking..

3.7.4 Jacking the airplane

Because the empty weight of this airplane is relatively low it is easy to lift the airplane using 2 people.
First prepare two suitable jacks to support the aircraft.
The aircraft should be lifted by the following parts:

- Press-down on the rear of the fuselage in front of the fin to lift the front and then support under the firewall. The fuselage nose may be supported under the horizontal tube of the engine mount or under the nose landing gear attachement to the fuselage.
- To jack the rear of the fuselage, grab the fuselage near the auxiliary tail skid, lift it upward and then support.
- To lift the wings, push on the wings lower surface at the main spar. Do not lift by the wing tips.

3.8 Road transport

The aircraft may be transported in a suitable trailer. It is necessary to remove (or to fold) the wings before loading. The aircraft and removed wings should be fastened down securely to ensure against possible damage.

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4. MAINTENANCE

4.1 Overall maintenance survey

Aircraft maintenance is required to mantain its airworthiness. Periodical events are performed (periodical and pre-flight inspections) along with irregular events e.g. a repair of a damage as required.

4.2 Pre-flight inspection

A pre-flight inspection is performed prior to the beginning of each flight. A pre-flight inspection should be repeated prior to each flight even during the same day.
The Pre-flight inspection is a visual check of the aircraft for deformations, surface damage, fuel and oil system leaks, prop damage, released locks, covers and cowlings etc.
Any damage or failure should be repaired immediately if the airworthiness is affected or when the aircraft can not be put out of operation.
It is important to perform a pre-flight inspection carefully to prevent problems from arising.
Refer to the Pilot's Operating Handbook for more details.

4.3 Post-flight inspection

Post-flight inspection is performed at the end of each flight day; the post-flight inspection events are the same as the preflight ones. If possible failures, damages and malfunctions should be recorded and repaired immediately. It is recommended to clean and/or wash the airplane and check that the fuel and oil consumption are in the normal range.
Lastly record all hours flown and other data in appropriate documentation (Log Book etc.).

4.4 Periodical inspections

4.4.1 Periodical inspection intervals

The periods for overall checks and contingent maintenance will depend on the conditions of the operation and the overall condition of the airplane. The manufacturer recommends maintenance checks and periodic inspections in the following periods:

1) after the first 25 ± 2 flight hours
2) after every 50 ± 3 flight hours
3) after every 100 ± 5 flight hours or annual inspection

Refer to the Rotax 912 Operator's Manual for engine maintenance.
The propeller is maintained according to its condition.

4.4.2 Periodical inspections Sign off sheets

The following Periodical maintenance Sign off Sheets are intended for copying and serve as the Maintenance Records. It is also recommended to include small repairs, damages and their remedy or replacement.
Some parts of the airplane (engine, propeller etc.) may have special time limits - refer to the appropriate manuals.

4.4.3 Periodical inspections - events

Model:	S/N.:	Hours flown:	Date of inspection:
EV -97 EUROSTAR	Registration:	No. of Takeoffs:	

Event #	Event description	Inspection ☑ Tick off performed inspection			Carried out by:	Inspected by:
		after the first 25 hrs.	every 50 hrs.	every 100 hrs.		
1.	Prior to the inspection clean and wash the airplane surfaces, if needed.	☒	☒	☒		
2.	**ENGINE**	see engine manufacturer´s instructions				
3.	**ENGINE COMPARTMENT**					
3.1.	**Fiberglass engine cowlings**					
3.1.1.	Check condition of cowlings and quick closing locks - repair any damage			☒		
3.1.2.	Remove engine cowling	☒	☒	☒		
3.1.3.	Visually check inside fireproof primer paint - Repaint if needeed - White color T 50, Norm V1000 N 56582		☒	☒		
3.2.	**Engine mount**					
3.2.1.	Visually check condition, attachment, security of attachment bolts: engine-engine mounting, engine mounting-firewall	☒	☒	☒		
3.2.2.	Visually check condition of rubber silentblocks - replace those cracked and excessively deformed			☒		
3.3.	**Suction system**					
3.3.1.	Visually check condition, attachment and security of air filter at carburetor inlet - clean filter acc. to the engine manual	☒	☒	☒		
3.3.2.	Visually check condition of suction tubing	☒	☒	☒		
3.3.3.	Check carburetor - condition, control cables attachment, lubricate cables at inlet to the bowdens	☒	☒	☒		
3.4.	**Battery**					
3.4.1.	Visually check attachment and security		☒	☒		
3.4.2.	Check charging - charge if needed			☒		
3.4.3.	Visually check condition and attachment of wire leads - replace those damaged	☒	☒	☒		
3.5.	**Wiring**					
3.5.1.	Visually check condition and integrity of wires, connections, security of wires	☒	☒	☒		
3.6.	**Fuel system**					
3.6.1.	Visually check condition, integrity, attachment and security of hoses - replace those damaged	☒	☒	☒		
3.6.2.	Visually check fuel filter condition - replace stopped up filter	☒	☒	☒		
3.6.3.	Visually check system for leaks	☒	☒	☒		

Event #	Event description	Inspection ☑ Tick off performed inspection			Carried out by:	Inspected by:
		after the first **25** hrs.	every **50** hrs.	every **100** hrs.		
3.7.	**Cooling system**					
3.7.1.	Visually check radiator for condition and leaks			☒		
3.7.2.	Visually check condition, attachment of hoses; check system for leaks	☒	☒	☒		
3.7.3.	Tighten hose clips if needed		☒	☒		
3.7.4.	Check coolant quantity in the expansion tank - add or change coolant acc. to the engine manual if needed	☒	☒	☒		
3.7.5.	Visually check condition and attachment of ovewflow bottle on the firewall			☒		
3.8.	**Lubrication system**					
3.8.1.	Visually check condition and attachment of oil tank			☒		
3.8.2.	Check oil cooler for condition, attachment and leaks	☒	☒	☒		
3.8.3.	Visually check hoses for condition, leaks, attachment and security - replace damaged hoses	☒	☒	☒		
3.8.4.	Check oil quantity - add or change oil acc. to the engine manual if needed	☒	☒	☒		
3.9.	**Exhaust system**					
3.9.1.	Visually check exhaust system for condition, cracks, deformations or damage - repair / replace	☒	☒	☒		
3.9.2.	Visually check condition and attachment of the muffler - repair / replace	☒	☒	☒		
3.9.3.	Check joint security	☒	☒	☒		
3.10.	**Heating**					
3.10.1.	Visually check hose leading hot air into the cockpit - check hose for condition, integrity, attachment and security		☒	☒		
3.10.2.	Check condition, function and control of the heating flap		☒	☒		
3.11.	**Reinstall lower engine cowling** Reinstall Upper engine cowling when the inspection is completed and engine test run performed	☒	☒	☒		
3.12.	**Lubricate per Lubricating Chart**	☒	☒	☒		
4.	**PROPELLER**	see manufacturer instructions +				
4.1.	**Blades**					
4.1.1.	Inspect blades for abrasions, cracks, paint damage, condition of blades leading edges and tips - repair according to the proppeler manual	☒	☒	☒		
4.2.	**Spinner**					
4.2.1.	Visually check spinner for condition, abrasions, cracks, paint damage - repair large damage		☒	☒		
4.2.2.	Remove spinner		☒	☒		
4.3.	**Propeller**	see manufacturer instructions +				
4.3.1.	Check prop attachment, security of bolts		☒	☒		
4.3.2.	Check run-out			☒		
4.3.3.	Install spinner			☒		

Event #	Event description	Inspection ☑ Tick off performed inspection			Carried out by:	Inspected by:
		after the first 25 hrs.	every 50 hrs.	every 100 hrs.		
4.3.4.	Pitch change mechanism (if controllable pitch prop is mounted) Check condition and function according to the prop manufacturer´s instructions	see manufacturer instructions				
5.	**LANDING GEAR NOSEWHEEL LANDING GEAR**					
5.1.	**Nosewheel leg**					
5.1.1.	Check condition and attachmnet of the nosewheel leg (lift airplane nose)	☒	☒	☒		
5.2.	**Wheel pants**					
5.2.1.	Visually check wheel pants or mudguards condition - repair damages and cracks		☒	☒		
5.2.2.	Remove fairing (reinstall when nosewheel inspection is completed)			☒		
5.3.	**Rubber rope and rubber suspension stop**					
5.3.1.	Visually check rubber rope a suspension stop for deformation, cracks, excessive wear - replace if needed		☒	☒		
5.4.	**Tire**					
5.4.1.	Check tires for condition, cuts, uneven or excessive wear and slippage - replace if needed		☒	☒		
5.4.2.	Check pressure - inflate to required pressure	☒	☒	☒		
5.5.	**Wheel**					
5.5.1.	Visually check for cracks, permanent deformations - if damaged, replace			☒		
5.5.2.	Check valve condition around the hole in the rim			☒		
5.5.3.	Check condition of bearings, wheel free rotation, play			☒		
5.6.	**Joints**					
5.6.1.	Check torque and security of fixed joints	☒	☒	☒		
5.6.2.	Check nosewheel free rotation inside the leg - the rotation should not be too free to prevent shimmy		☒	☒		
5.7.	**Nosewheel control system**					
5.7.1.	Check control rods condition, rod ends security		☒	☒		
5.7.2.	Check condition of nosewheel control lever covers for wear through - repair damage			☒		
5.8.	**Lubricate per Lubricating Chart**	☒	☒	☒		
6.	**LANDING GEAR MAIN LANDING GEAR**					
6.1.	**Fiberglass legs**					
6.1.1.	Visually check condition of fiberglass legs - repaint damaged areas, contact airplane manufacturer if cracks were found	☒	☒	☒		
6.1.2.	Inspect leg attachment into the fuselage (no play) - Lift the landing gear and move a leg forward-backward, upward-downward; at the same time check wheel play on the axle - tighten attachment bolts if the leg has a play		☒	☒		
6.1.3.	Check torque and security of fixed joints	☒	☒	☒		

Event #	Event description	Inspection ☑ Tick off performed inspection			Carried out by:	Inspected by:
		after the first 25 hrs.	every 50 hrs.	every 100 hrs.		
6.1.4.	Check cloth cover which covers the leg-fuselage input hole			☒		
6.2.	**Wheel pants or mudguards**					
6.2.1.	Visually check wheel pants / mudguards condition - repair damage and cracks		☒	☒		
6.3.	**Tires**					
6.3.1.	Check tires for condition, cuts, uneven or excessive wear and slippage - replace if needed	☒	☒	☒		
6.4.	**Wheel**					
6.4.1.	Visually check wheel rims for cracks, permanent deformations - replace wheel rim in case of cracks			☒		
6.4.2.	Check valve condition around the hole in the disc			☒		
6.4.3.	Check condition of bearings, wheel free rotation, play		☒	☒		
6.5.	**Brakes**					
6.5.1.	Check attachment of brake system plastic hoses to the main leg			☒		
6.5.2.	Visually check condition of pads - steady and symmetry abrasion of pads - replace pads if needed		☒	☒		
6.5.3.	Check wear of the disc			☒		
6.5.4.	Check brake system for leaks - add brake fluid and bleed the system if a brake pedal has soft movement	☒	☒	☒		
7.	**WING**					
7.1.	**Wing**					
7.1.1.	Visually check condition - no loose rivets, deformations, cracks or any other damage - contact the airplane manufacturer	☒	☒	☒		
7.1.2.	Check play of wing suspensions - move the wing tip upward-downward, frontward-rearward			☒		
7.1.3.	Check condition and attachment of fiberglass wing tips			☒		
7.2.	**Aileron**					
7.2.1.	Visually check condition	☒	☒	☒		
7.2.2.	Check free movement	☒	☒	☒		
7.2.3.	Check aileron hinge	☒	☒	☒		
7.2.4.	Check play		☒	☒		
7.2.5.	Check security of control rod ends	☒	☒	☒		
7.2.6.	Lubricate per Lubricating Chart	☒	☒	☒		
7.2.7.	Remove inspection covers from the lower wing surface to inspect security of control system joints			☒		
7.2.8.	Lubricate per Lubricating Chart and reinstall covers	☒	☒	☒		
7.3.	**Flap**					
7.3.1.	Fully extend the flaps and visually check condition	☒	☒	☒		

Event #	Event description	Inspection ☑ Tick off performed inspection			Carried out by:	Inspected by:
		after the first 25 hrs.	every 50 hrs.	every 100 hrs.		
7.3.2.	Check flap hinge	☒	☒	☒		
7.3.3.	Check play		☒	☒		
7.3.4.	Check condition of flap control pin and wear of the groove at the flap root			☒		
7.3.5.	Lubricate per Lubricating Chart	☒	☒	☒		
7.4.	**Pitotstatic tube**					
7.4.1.	Check pitotstatic tube attachment			☒		
7.4.2.	Check pitostatic system for leaks - the airplane manufacturer uses KPU 3 instrument			☒		
7.5.	**Wing suspensions**					
7.5.1.	Remove wing fillets	☒	☒	☒		
7.5.2.	Visually check condition of wing suspensions (wing folding mechanism), cleanness of folding system, lubrication	☒	☒	☒		
7.5.3.	Check wear, corrosion			☒		
7.5.4.	Check security of joints	☒	☒	☒		
7.6.	**Lubricate per Lubricating Chart**	☒	☒	☒		
8.	**FUSELAGE**					
8.1.	**Fuselage surface**					
8.1.1.	Visually check condition - no loose rivets, deformations, cracks or any other damage - repair small damage or contact the airplane manufacturer	☒	☒	☒		
8.1.2.	Visually check rivets near the landing gear attachment			☒		
8.1.3.	Check condition and attachment of equipment - antenna, beacon etc.			☒		
8.1.4.	Check tail skid attachment			☒		
8.1.5.	Visually check condition, attachment and operation of towing mechanism (if installed)		☒	☒		
8.1.6.	Visually check condition of fiberglass wing fillets			☒		
8.2.	**Cockpit canopy**					
8.2.1.	Visually check canopy condition for - cracks, scratches, any other damage - drill end of cracks	☒	☒	☒		
8.2.2.	Check canopy lock for condition and operation	☒	☒	☒		
8.2.3.	Check vent windows for condition and operation			☒		
8.2.4.	Check gas struts operation - replace those functionless			☒		
8.2.5.	Check canopy rubber packing			☒		
9.	**HORIZONTAL TAIL UNIT**					
9.1.	**Visually check condition** - no loose rivets, deformation, cracks, scratches and any other damage - contact the airplane manufacturer	☒	☒	☒		
9.2.	**Visually check condition and attachment of fiberglass tips**			☒		
9.3.	**Check elevator free movement**	☒	☒	☒		
9.4.	**Check elevator hinge**	☒	☒	☒		

Event #	Event description	Inspection ☑ Tick off performed inspection			Carried out by:	Inspected by:
		after the first 25 hrs.	every 50 hrs.	every 100 hrs.		
9.5.	**Check play** - move the stabilizer frontward-rearward, upward-downward - contact the airplane manufacurer if play exceeded tolerances		☒	☒		
9.6.	**Check security of joints at control column**	☒	☒	☒		
9.7.	**Trim tab**					
9.7.1.	Visually check condition		☒	☒		
9.7.2.	Check hinge		☒	☒		
9.7.3.	Check control cables condition			☒		
9.7.4.	Check tension of trim tab control cables and check securing the adjusting screws			☒		
9.8.	**Lubricate per Lubricating Chart**	☒	☒	☒		
10.	**VERTICAL TAIL UNIT**					
10.1.	**Visually check condition** - no loose rivets, deformation, cracks, scratches and/or other damage - contact the airplane manufacturer	☒	☒	☒		
10.2.	**Visually check condition and attachment of fiberglass tips**			☒		
10.3.	**Check rudder free movement**	☒	☒	☒		
10.4.	**Check rudder suspensions**	☒	☒	☒		
10.5.	**Check play** - move rudder upward-downward			☒		
10.6.	**Check joints security**	☒	☒	☒		
10.7.	**Lubricate per Lubricating Chart**	☒	☒	☒		
11.	**COCKPIT**					
11.1.	**Instrument panel**					
11.1.1.	Visually check condition and attachment of the instrument panel		☒	☒		
11.1.2.	Check condition and attachment of individual instruments		☒	☒		
11.1.3.	Check function of instruments			☒		
11.1.4.	Check throttle and choke levers free movent and lock	☒	☒	☒		
11.1.5.	Inspect completeness and readability of placards			☒		
11.2.	**Seats**					
11.2.1.	Visually check seat upholstery, remove upholstery			☒		
11.2.2.	Visually check seats and backrests condition			☒		
11.2.3.	Check for loose rivets or any other damage on the seats			☒		
11.2.4.	Visually check main landing gear legs attachment inside the fuselage			☒		
11.3.	**Safety harness**					
11.3.1.	Visually check condition, attachment and security			☒		
11.4.	**Hand control**					
11.4.1.	Remove aileron rod covers inside the cockpit		☒	☒		
11.4.2.	Check hand control free movement	☒	☒	☒		
11.4.3.	Check play	☒	☒	☒		
11.4.4.	Check joints security	☒	☒	☒		
11.4.5.	Check control column stops for condition			☒		

Event #	Event description	Inspection ☑ Tick off performed inspection			Carried out by:	Inspected by:
		after the first 25 hrs.	every 50 hrs.	every 100 hrs.		
11.4.6.	Pitostatic system drainage, see 2.3.19			☒		
11.4.7.	Lubricate per Lubricating Chart	☒	☒	☒		
11.5.	**Rudder control**					
11.5.1.	Check stiffness of movement	☒	☒	☒		
11.5.2.	Check joints security	☒	☒	☒		
11.5.3.	Check stops at pedal control cables			☒		
11.5.4.	Check condition and security of cables	☒	☒	☒		
11.5.5.	Check hydraulic brake system, add brake fluid if needed	☒				
11.5.6.	Lubricate per Lubricating Chart	☒	☒	☒		
11.6.	**Flap and trim control, Towing mechanism control**					
11.6.1.	Check free movement of levers	☒	☒	☒		
11.6.2.	Check opperation of flap control lever lock (push button)		☒	☒		
11.6.3.	Lubricate per Lubricating Chart	☒	☒	☒		
11.7.	**Complete lubricating per Lubricating Chart**	☒	☒	☒		
11.8.	**Install seats upholstery and covers**					
11.9.	**Engine Test Run (see POH)** • idling • throttle and choke levers operation • acceleration - deceleration • r.p.m. drop with either magneto switched off • max.r.p.m. • test brake system efficiency	☒	☒	☒		
11.10.	**Test flight**	☒	☒	☒		
11.11.	**Clean the airplane surface** (only for service station)	☒	☒	☒		

Date of Issue: 06/2001 **Revision:** MB1

4.4.4 List of periodical inspections of Rotax 912 UL engine

Engine: Rotax 912	S/N:	Hours flown:	Date of inspection:
Airplane: EV-97	S/N:	Registration:	

Event description	after the first 25 hr.	Every 100 hr. (after 100 hr. or 1 year, whichever occurs first)	every 200 hr.	Every 1000 hr. (after 1000 hr. or 10 years, whichever occurs first)	every 2 years	every 5 years	every 10 years	see. Rotax 912 F Operator's Manual	Carried out by	Inspected by
1. Engine cleaning	☒	☒	☒					12.3.1		
2. Engine inspection	☒	☒	☒					12.3.2		
3. Leak-proof check	☒	☒	☒					12.3.3		
4. Suspension check	☒	☒	☒					12.3.4		
5. Check of equipment	☒	☒	☒					12.3.5		
6. Check of reducer	☒	☒	☒					12.3.6		
7. Oil quantity check	☒	☒	☒					12.3.7		
8. Oil change	☒	☒	☒					12.3.8		
9. Oil filter replacement	☒	☒	☒					12.3.9		
10. Cooling system inspection	☒	☒	☒					12.3.10		
11. Cooling sytem flush			☒					12.3.10. 1		
12. Liquid coolant change					☒			12.3.10. 2		
13. Air filter check	☒	☒	☒					12.3.11		
14. Carburetor check	☒		☒					12.3.12		
15. Check of cabling	☒	☒	☒					12.3.13		
16. Auxiliarry generator V-belt tension check (if installed)	☒	☒	☒					12.3.14		
17. Replacement of spark-plugs			☒					12.3.15		
18. Ignition wires on spar plugs attachment			☒					12.3.16		
19. Check of compression			☒					12.3.17		
20. Engine test run	☒	☒	☒					12.3.18		
21. Replacement of rubber parts						☒		12.3.19		
22. Overhaul				☒			☒	12.3.20		

4.5 Fluids

The fluids are: fuel, engine oil, liquid coolant and brake fluid.
Filling locations can be seen in the Figure below. Fuel and Brake fluid filling locations are described in 4.6.4.3 and 4.6.3.2 respectivly.

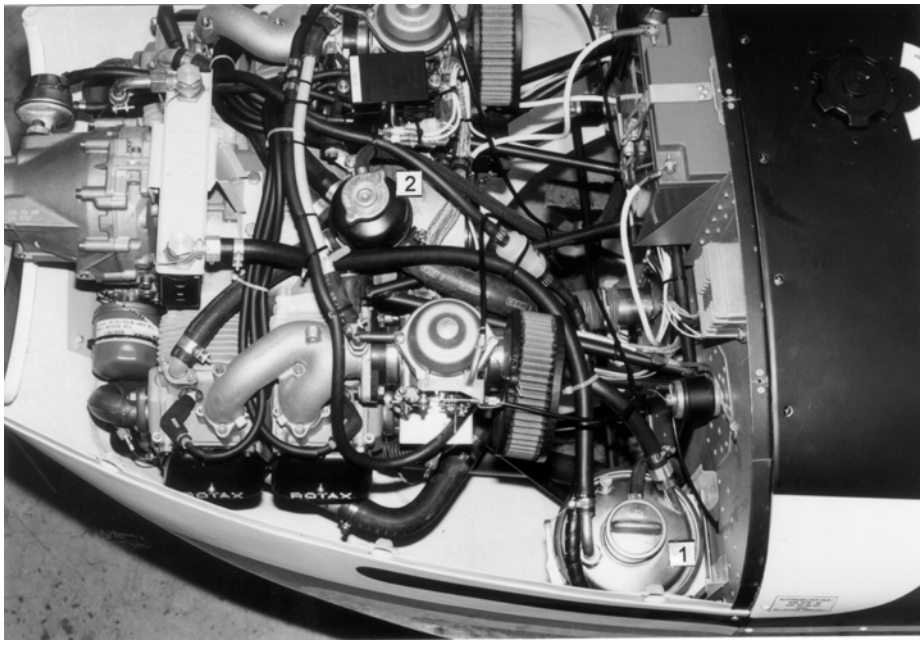


Fig. Filling locations in engine compartment
1- oil tank,
2 - liquid coolant tank

4.5.1 Engine oil

4.5.1.1 Recommended brands

The recommended oil brands are listed in Service Information 18 UL 97-D/E, Jan. 1998, which is enclosed with this Manual.

4.5.1.1.1 Table of oils

see Engine Operator's manual for suitable oil grades.

4.5.1.2 Oil quantity

The total oil quantity in the Rotax 912 lubricating system amounts to 0.9 USGAL (3.5 liters). Prior to oil check, turn the propeller by hand (ingition switched off!) several times to pump oil from the engine into the oil tank, or leave the engine idle for 1 minute. The oil level in the oil tank should be between the min. and max. marks and should not be below min. mark.

4.5.1.3 Oil filling

The oil tank is located in the engine compartment and is accessible when engine upper cowling is removed. Oil quantity is measured by wire-gauge in the oil tank - see previous paragraph.

4.5.1.4 Oil emptying

Unscrew the plug located on the bottom of the oil tank to empty out the oil.
To empty oil from the engine, unscrew the plug located on the bottom of the engine, close to the oil return hose.
It is recommended to empty oil when the engine is warm.

4.5.2 Coolant

4.5.2.1 Recommended types

Refer to the Rotax 912 Operator's Manual for recommended coolant types. The "BASF Glysantin Anticorrosion", "FRIDEX G 48" or "Glysantin Protect Plus (produced by BASF)" is recommended by the engine manufacturer. The engine manufacturer also recommends the use of antifreeze concentrate during cold weather operation.

4.5.2.2 Coolant quantity

Total coolant quantity is about *1.6 USQTS* (1.5 liters).

4.5.2.3 Coolant refilling

The expansion tank located in the engine compartment is used for filling. In addition to that, an overflow bottle is attached on the firewall to absorb coolant in the case of engine overheating.

4.5.2.4 Coolant emptying

Disconnect the hose going from the radiator into the pump (on the lowest part of the cooling system) to empty coolant into a suitable container.

4.5.3 Brake fluid

4.5.3.1 Recommended types

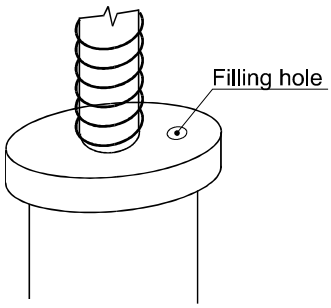
Only brake fluid of J 1703c classification should be used for hydraulic brake system (type for middle hard or hard operation).

Czech Rep.	Foreign
• Syntol HD 205 or • Syntol HD 260	• ATE Blau • STOP SP 19 • MOBIL Hydraulic Brake Fluid 550 • BP Brake Fluid • PENTOSIN Super Fluid • AGIP F. 1 Brake Fluid Super HD • NAFTAGAS AT-2 • INA UK-2.

These brake fluid types may be blended as required and refilled in any mixing proportion.

4.5.3.2 Brake fluid refilling

Brake fluid refilling is necessary when a low brake system efficiency occurs due to a fluid leak.
A brake fluid filling hole is drilled in the brake master cylinder attached to a brake pedal (see fig.). It is recommended to use a hypodermic needle to refill the brake cylinder. See table for the suitable brake fluid types to use for refilling the brake system. Step repeatedly on the pedal during refilling. Bleed the system after refilling.



4.5.3.3 Brake fluid emptying

Brake fluid thickens during aircraft operation and absorbes water. This condition causes brake system failures. It is not possible to determine when this may occur. The best way to prevent trouble is to change the brake fluid every year.

4.5.4 Fuel

4.5.4.1 Recommended brands

Refer to enclosed Service Information 18 UL 97 D/E, Issue Jan. 1998 for recommended fuel brands.

4.5.4.2 Fuel quantity

The standard aircraft is equipped with a *17.2USGAL* (65 liters) fuel tank.
An additional fuel tank optional. Keep the maximum permitted take-off weight in mind when adding fuel to a larger tank.

4.5.4.3 Fueling

Precaution
The following precautions should be maintained during fueling to prevent fire.

WARNING

- No smoking or open flames during fueling!
- Fire extinguisher should be within reach!
- Under no circumstances add fuel with the engine running!
- Connect the aircraft to ground prior fueling.
- No person in the cockpit during fueling!

A fuel tank filler is located on the right hand side of the fuselage, close to the rear cockpit canopy (see photo). If the additional fuel tank is installed (in the front part of the fuselage), both tanks are connected. The filler neck is located on the upper surface of the additional fuel tank (not on the right hand side of the fuselage) and tanked fuel flows into the standard fuel tank first of all. From here the fuel is sucked up by the engine pump.

Fig. Fuel tank filler neck

A gasoline can and a funnel with a flexible end may be used to fill the fuel tank or a device described below may be constructed. It consists of a gasoline can and tire-pump. A gasoline can funnel is set on the gasoline can - a tire valve is brazed on the funnel and a hose on the tire-pump (compressor) is connected to the valve. A suction tube with a filter is welded to the gasoline can funnel. The tube is inserted into the gasoline can. A flexible hose is attached to the funnel. Pressure in the gasoline can will increase during pumping. Gasoline is then forced through the hose into the fuel tank. An advantage of fueling with this device is easy handling by 1 person.
Close the fuel tank filler using the lockable cap when the tank filled up.
Clean the aircraft surface if stained with gasoline.

CAUTION

It is highly recommended to pour gasoline through a filter if it was not tested for water content. After fueling, allow 20 min. for water to settle out on the bottom. Drain off some fuel and look for water.
Avoid getting gasoline on the rear cockpit canopy which will run the the perspex canopy!!!

4.5.4.4 Fuel emptying

Precaution
Use the same precautions as during fueling.

Draining procedure
1. Connect the airplane to the ground
2. Open the main fuel valve
3. Fully extend the flaps
4. Put an empty gas can under the drainage hose (on the bottom of fuselage close to the right hand flap root)
5. Open the drain valve (under the right wing fillet, close to the right hand flap root)
6. Close the drain valve when desired quantity of fuel is reached
7. Close the main fuel valve
8. Retract the flaps

NOTE

Remove the fuel tank filler cap to speed up draining.

4.6 Lubrication

4.6.1 Lubrication fundamentals

There are some generally inaccessible joints and control system parts inside the wings and fuselage, which have been cleaned and lubricated during airplane assembly. Lubrication of these will be performed during a periodic inspection.
There are some parts, e.g. landing gear, which are exposed to external conditions and to varying loads. These parts will be inspected during pre-flight and during periodical inspections. These should be lubricated as is necessary, but at least in the intervals specified below.

4.6.2 Recommended lubricants

4.6.2.1 Greases

Greases are mineral oils thickened with calcic, sodium, lithium or any other thickeners of aliphatic acids.
The greases do not SAE classification and their usage is recommended by manufacturer. Grease may be applied all the year round.

The following greases are recommended:

- **Greases of LV series** are of waxy, semi-solid or butyraceous consistency and water resistant. They are used at very low temperatures (*-58 °F,* -50 °C) and at high temperatures (*302 °F*, 150 °C). There are two types marked LV2 and LV3. Each type has specific characteristics determining its use.

Czech BENZINA BENZINOL	Foreign
LV 2	CASTROL
	Castrolease LM
	MOBIL
LV 3	Mobilgrease MP
	SHELL
	Retinax A
	LITOL 24

4.6.2.2 Lubricating points

Unit	Lubricating point	after the first 25 hrs.	Every 50 hrs.	Every 100 hrs	Lubricant
Prop	• Adjustable props acc. to Prop Manual				
Engine	• oil change acc. to Engine Manual				
	• carburetor control cable at inlet into the bowden (in engine compartment)	x	x		oil
	• choke control cable at inlet into the termination (in engine compartment)	x	x		oil
Nosewheel landing gear	• landing gear leg in the area of bushing	x	x	x	oil
	• bearings in pull rod terminals of landing gear control	x	x	x	oil
Main landing gear	• pins of brake pads holders		x		LV2, LV3, foreign greases
Wing	• all movable joints of wing folding mechanism (if mounted)	x	x	x	LV2, LV3, foreign greases
Ailerons	• hinges		x		oil
	• control hinge pin			x	LV2, LV3, foreign greases
	• two-armed aileron control levers inside the wing			x	LV2, LV3, foreign greases
	• hinge joint of rods under the wing fillet			x	LV2, LV3, foreign greases
Flaps	• hinges	x	x		oil
	• all movable joints under the quadrant cover between the seats			x	LV2, LV3, foreign greases
	• All movable joints under the baggage compartment bottom cover			x	LV2, LV3, foreign greases
	• Flaps control pins (at a flap root)		x		LV2, LV3, foreign greases
HTU	• elevator hinge		x		oil
	• swivel bearing in the elevator control rod termination			x	LV2, LV3, foreign greases
VTU	• rudder suspensions			x	LV2, LV3, foreign greases
	• rudder control cables at attachment to the rudder			x	LV2, LV3, foreign greases
Trim tab	• trim tab hinge	x	x		oil
	• control cables at inlets inot the terminations			x	LV2, LV3, foreign greases
Stick control	• All movable joints in the cockpit			x	LV2, LV3, foreign greases
Rudder control	• All movable joints in the cockpit			x	LV2, LV3, foreign greases
	• The passages of rudder control cables			x	LV2, LV3, foreign greases
	• Brake system control cables at inlets in the bowdens (at brake pedals)			x	LV2, LV3, foreign greases

4.7 Mechanism adjustments

4.7.1 Torque moments

Metric thread		Strengh class									
		4D	5D	4S	6E	5S	5R	6S	8G	10K	12K
M4	N.m					1,67					
	kg.m					0,17					
M5	N.m					3,45					
	kg.m					0,35					
M6	N.m	4,31	4,90	5,39	5,88	6,86	7,84	8,33	9,80	13,72	16,67
	kg.m	0,44	0,50	0,55	0,60	0,70	0,80	0,85	1,00	1,40	1,70
M7	N.m	5,88	7,84	8,82	9,80	10,78	11,76	12,74	14,70	20,59	25,49
	kg.m	0,60	0,30	0,90	1,00	1,10	1,20	1,30	1,50	2,10	2,60
M8	N.m	8,33	10,78	12,74	13,72	15,69	17,65	19,61	22,55	32,36	38,24
	kg.m	0,85	1,10	1,30	1,40	1,60	1,80	2,00	2,30	3,30	3,90
M10	N.m	16,18	21,57	24,51	27,45	31,38	34,32	37,26	44,12	61,78	73,54
	kg.m	1,65	2,20	2,50	2,80	3,20	3,50	3,80	4,50	6,30	7,50
M12	N.m	27,45	36,28	42,16	47,07	52,95	58,83	63,74	74,53	104,93	125,52
	kg.m	2,80	3,70	4,30	4,80	5,40	6,00	6,50	7,60	10,70	12,80
M14	N.m	43,14	58,83	66,68	73,54	78,54	93,16	98,06	117,67	164,75	196,13
	kg.m	4,4	6,00	6,80	7,50	8,00	9,50	10,00	12,00	16,80	20,00
M16	N.m	60,80	78,45	93,16	98,06	107,87	127,48	131,29	164,75	225,55	274,58
	kg.m	6,20	8,00	9,50	10,00	11,50	13,00	14,00	16,80	23,00	28,00
M18	N.m	88,25	117,67	137,29	156,90	171,61	196,13	205,93	245,16	343,23	411,87
	kg.m	9,00	12,0	14,00	16,00	17,50	20,00	21,00	25,00	35,00	42,00
M20	N.m	117,67	156,90	176,51	196,13	225,55	245,16	274,58	313,81	441,29	539,36
	kg.m	12,00	16,00	18,00	20,00	23,00	25,00	28,00	32,00	45,00	55,00
M22	N.m	147,09	196,13	225,55	245,16	284,39	313,81	333,42	392,26	558,97	676,65
	kg.m	15,00	20,00	23,00	25,00	29,00	32,00	34,00	40,00	57,00	69,00
M24	N.m	205,93	274,58	313,81	353,03	392,26	441,29	470,71	549,17	755,11	970,85
	kg.m	21,00	28,00	32,00	36,00	40,00	45,00	48,00	56,00	77,00	99,00
Ultimate strength		37	50	37	-	50	-	60	80	100	120
ϑ in %		25	22	14	-	7	-	8	12	8	8
Yield point		21	28	32	36	40	45	48	64	90	108

Torque moment formula (valid for all bolt sizes):

$$M_{kmax} = 1{,}065 \times \frac{d \cdot \sigma \cdot S}{m}$$

Legend:

Mk.....	torque moment	kg.cm
d.....	bolt shank diam.	cm
σ	min.yield point	kg/cm^2
m.....	safety factor	(m=1,25 for σ<50 kg/mm^2; m=1,43 for σ>50 kg/mm^2)
s.....	lead of helix	cm

4.8 Necessary maintenance tools

No special tools are needed for the ***EV-97 „EUROSTAR“ model 2000 version R*** maintenance. Tools used for automobile maintenance are suitable.
If the aircraft is equipped with the wing folding mechanism, the following additional tools are supplied with that aircraft: assembly lever, rests for folded wings and others as required.

4.9 Access holes

The following are the inspection and access holes:

- Screw caps on the wings lower surface - access to the aileron control rods and levers and to the pitot/static installation in the left half of the wing
- Screw cap on the fuselage lower surface under the baggage compartment close to the fuel tank - access to the fuel tank installation
- Screw cap on the fuselage lower surface in the middle of the rear section - access to the elevator control rods and a lever
- Wing fillets which cover space between the fuselage and wing - access to the wing-fuselage suspensions (wing folding mechanism)
- Cover sheet of Control stick system in the cockpit

4.10 Brake system efficiency adjustment

4.10.1 Brake pad replacement

Brake pad replacement is only performed when a pad is worn-out.

CAUTION
Due to the possibility of brake fluid leaking, it is advisable not to loosen the hose cup nut during brake pad removal. In the case of a leak in the brake system, filling and bleeding is necessary.

Brake pad replacement procedure

1. Jack the airplane
2. Remove the cotter pin, unscrew the slotted nut M16, remove the washer from the axle
3. Bend small tabs on the 3 washers and unscrew the M6 screws connecting the brake disc to the wheel rim
4. Remove the wheel and the distance ring from the axle
5. Take the brake disc off (leave the brake on a main leg)
6. Remove the cotter pins, shift the pins out and remove the brake pads
7. Mount a new brake pad, secure the pins with cotter pins (diam. 2mm, length14mm) (part.No.: 039 300)
8. Put the brake disc on the wheel
9. Put the distance ring and the wheel on the axle (adjust the distance ring between bearings)
10. Set tab washers (6x2, Nom. 038160) on the screws, apply Loctite and attach the brake disc to the inner part of the rim. Bend the tabs of the washers to secure the screw heads.
11. Put the washer on the axle, tighten the slotted nut and secure with a cotter pin

List of parts necessary for Brake pad replacement procedure:

1. Brake pads	4	pcs		Nom.	765 195
2. Cotter pin	4	pcs	ϕ 2x14mm	Czech Stand.	021781.04
				Nom.	039 300
3. Cotter pin	2	pcs	ϕ 4x40mm	Czech Stand.	021781.04
				NOM.	040 350
4. Washer	6	pcs	6x2mm	Works Stand.	3288.2
				Nom.	038 160

5. LOCTITE (e.g. 638) to secure the screws

4.10.2 Bleeding

It is important to thoroughly bleed the brake system. Otherwise the system function may be unreliable and the brakes may fail. There are two main reasons for air entering the brake system:

1. Disconnected or loose hoses
2. Insufficient quantity of brake fluid

Fig. Brake system bleeding
1- Bleeding screw

Procedure:

1. Loosen the bleeding screw in the brake cylinder
2. Step repeatedly on the pedal to bleed the brake system
3. Tighten the screw
4. Repeat several times or until the pedal offers resistance against motion (feels firm)

NOTE
If the brake efficiency remains unsatisfactory after bleeding or if the pedal motion is excessive, fill with brake fluid and bleed the system again. Continue untill all the air is out of the system

4.11 Control surfaces deflection setting

Control surface deflections of a new aircraft are set by the manufacturer. Deflections are adjusted to values specified in the Control Surfaces Deflection Record enclosed in this Manual.
A neutral position of the control surfaces and controls is used as a base for adjustment of deflections.

4.11.1 Aileron deflection adjustment

A range of deflections are set using adjustment screws on lateral control stops located on a console of the left/right control column.
The aileron neutral position can be adjusted with the adjustable end of the short rod inside the cockpit. Or with the adjustable end of the longer rod inside the wing (when the screw cap on the lower wing surface is removed).
The aileron differential (difference between a lower and upper deflection) can be adjusted with the adjustable end of the rod connecting the two-armed lever inside the wing and the hinge. The adjustable rod end is accessible when an aileron is deflected upward.

4.11.2 Flap deflection adjustment

Flaps deflection is determined by cut-outs in a changing gate on the flap control lever inside the cockpit. Use the following procedure to adjust flap deflection:
Remove the baggage compartment bottom cover which will allow access to a rod connecting the flap control lever and a lever welded on a tube connecting the left and right flap. Use the adjustable rod end to adjust the flaps in the position "RETRACTED" .
The deflection of the left flap against the right flap may be adjusted when the eccentric pin is moved a slight amount. The pin is inserted into the groove of the flap root.
See Figure in par. 2.3.11.9 for adjustment points.

4.11.3 Elevator deflection adjustment

The range of elevator deflection is determined by the control stick stops setting. The "Push in" stop may be adjusted with an adjustment screw. To adjust an upward deflection, fully pull the control stick rearward. Then measure the elevator upward deflection and compare it with the one specified – if required adjust the deflection with the adjustable end of the elevator control rod or with the adjustable end of the rod below the baggage compartment cover. When the upward elevator deflection is adjusted, fully push the control stick and adjust a stop so that an elevator downward deflection corresponds with that one specified.
See Figure in par. 2.3.11.7 for adjustment points.

4.11.4 Rudder deflection adjustment

The rudder deflections are set by the aircraft manufacturer and can not be adjusted.

4.11.5 Trim tab adjustment

The following may be adjusted:

a) The position of the trim tab control lever may be adjusted against a groove in the quadrant between the seats.
Loosen the bolts which clamp the ends of the trim tab control cables on the upper/lower tab surface. Then set the trim tab control lever in the desired position, slightly tighten the cables and retighten the bolts.

b) Trim tab deflections
Trim tab deflection may be adjusted using the adjustment bolts which clamp the ends of the control cables. The adjustment bolts are on the upper/lower tab surface – see Fig. in par. 2.3.11.10.

4.12 Steerable nosewheel landing gear adjustment

A steerable nosewheel adjustment is necessary so the airplane tracks during straight taxiing with engine idling (to eliminate prop turning moment) with rudder pedals held in neutral position (no crosswind!).

Procedure:

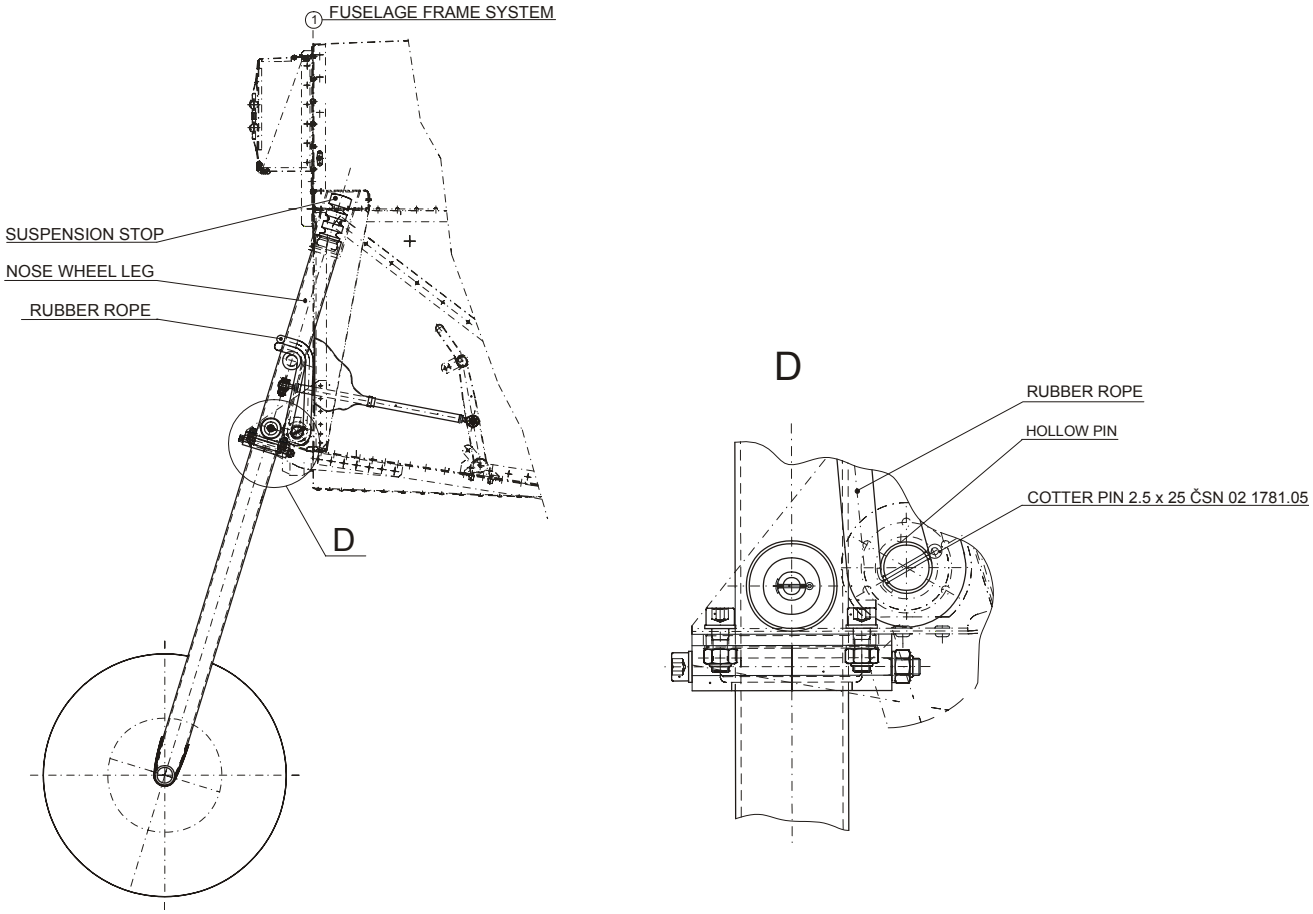
1. Lift the nosewheel and neutralize wheel and rudder pedals
2. Adjust the nosewheel control cables using the adjustable cable ends located close to the control levers under the fuselage

4.12.1 Rubber shock absorber replacement

Perform rubber shock absorber replacement when rubber rope is damaged or excessivelly worn down.

Procedure:

1. Remove cotter pins from the hollow pin attaching a rubber rope
2. Remove hollow pin
3. Remove rubber rope from the nose landing gear leg
4. Mount a new rubber rope at the leg, sling over pin
5. Insert a hollow pin back
6. Secure by two cotter pins



4.13 Engine idle adjustment

Because the engine idle is adjusted on a running engine, use extreme caution near the propeller. The aircraft should be tied down. Use the adjustment screw on the carburetor of the Rotax 912 engine to adjust the idle. Idle engine speed is approximately 1400 r.p.m..

4.14 Tire inflation pressure

Main and nose wheel pressure:

Sava tire 14x4	*26.5 + 3*	*psi*	180 + 20	kPa
Deli tire or Air track 15 x 6.00 – 6 or				
GOOD YEAR	*11.6 + 3*	*psi*	80 + 20	kPa
TOST 15 x 6.00 – 6	*17,4 + 3*	*psi*	120 + 20	KPa

Tire Pressures are noted on placards located on the airdraft. Use the adapter supplied with the aircraft to inflate the nosewheel. A car tire pump or compressor, or pressure bottle may be used for inflating the wheels.

4.15 Cleaning and care

4.15.1 Airplane care outlines

Use mild detergents to clean the exterior surfaces. Oil spots on the surfaces (except the canopy!) may be cleaned with gasoline or strong detergents such as 409.
Upholstery covers can be removed from the cockpit, brushed or washed in lukewarm water with a laundry detergent. Dry the upholstery before reinstalling.

4.15.2 External surfaces cleaning

The external metal surfaces and fiberglass parts (cowlings, wingtips etc.) of the airplane are protected with weather-proof paint. Wash the airplane surface with lukewarm water and car wash type detergents. Then wash the airplane with water and sponge dry. It is recommended to protect painted external surfaces twice a year, by appling an automotive type polish. Use only on a clean and dry surface, and polish with a soft flannel rag.

CAUTION

- Never wipe a dry surface – the surface may be scratched by dusts and dirt
- Never apply any chemical solvents
- Repair a damaged painted surface (see par. 5.6) as soon as possible to prevent corrosion

4.15.3 Interior cleaning

Keep in mind the following:

- Remove any loose objects from the cockpit
- Vacuum the interior, upholstery and carpets
- Wipe the upholstery using a rag with in lukewarm water and mild laundry detergent. Then dry or remove the seat upholstery, side panels, carpets and clean with lukewarm water and/or carpet cleaners, upholstry cleaners. Dry thoroughly before reinstallation.
- Clean the cockpit canopy interior surface (see par.4.15.4) below.

4.15.4 Cockpit canopy cleaning

The canopy may be cleaned by washing it with lukewarm water and car or laundry type detergents. Use a clean, soft cloth. Then use a suitable polisher on the canopy such as Meguire's plastic polish.

CAUTION

- Never clean dry canopy
- Never apply gasoline or chemical solvents!
- Cover the canopy with a cover sheet

4.15.5 Engine maintenance

Refer to the engine manufacturer's instructions for engine maintenance.

4.15.6 Propeller maintenance

Refer to the propeller manufacturer's instructions for engine maintenance.

4.15.7 Winter operation

4.15.7.1 General

It is considered a winter operation, if the outside temperature falls below *41 °F* (+5 °C) .

4.15.7.2 Preparing the aircraft for winter operation

Aircraft airframe

- Lubricate the aircraft per Lubricating Chart (100 hr. Inspection) if the last inspection was not within 6 months
- Check and adjust rudder control cable prestress
- Check cockpit canopy rubber packing – replace if damaged
- Check fuel tank venting
- Check attachment of wing, ailerons, flaps and tail units; lubricate per Lubricating Chart

Engine

Refer to the Engine Manual for more details.
The following should be done:

- Add Anti-freeze to the cooling system as reuired (usually a 50/50 mix.)
- Change the oil (see Engine Manual or Service instructions)
- Check the spark plug gaps

If low cylinder head or oil temperatures occur during operation under low outside temperature, then do the following:

- Cover a portion of the radiator face using an alluminum sheet, insert it between the radiator and the lower engine cowling.
- Cover the oil cooler face (or a part of face) using an alluminum sheet attached with a suitable adhesive tape.
- Cover the Reduction gear unit by means of a car engine cover
- Cover the oil filter (see instructions for Reduction gear unit)
- Cover the oil tank (see instructions for Reduction gear unit)

4.15.7.3 Operation

Preflight inspection

In addition to the Pre-flight inspection described in the POH, the following must be done:

- Remove all snow from the airplane surfaces,and remove any icing using hot air
- Check the control surfaces for free movement. Check the slots of the control surfaces and flaps. Remove any snow or ice
- Check the fuel tank vent hose for openness before each flight
- Check fuel system (fuel filter) for debbris, if found, empty the fuel tank and refill with fresh fuel
- Check fuel for water prior filling (use of water separator/filter is recommended)
- Drain fuel tank (drain valve is located close to the right flap root), drain a small quantity of fuel; check for water and other contaminates untill fuel is clean and clear.

Pre-heating engine and oil
It is permissible to start an engine without pre-heating if the outside air temperature is not below *41 °F* (+5 °C). Pre-heat the engine and oil if air temperature falls below *41 °F* (+5 °C). Use suitable air heater or a dryer.

WARNING
Never use open fire to pre-heat an engine!

Blow hot air from the front into the hole around the prop (engine covered with fiberglass cowlings). The temperature of the hot air should not exceed *212 °F* (100 °C) at air heater output. Warm up the oil tank along with the oil in the engine. Pre-heat until cylinder head and oil temperatures exceed *68 °F* (+20° C).

Engine starting
1. Turn the propeller by hand (ignition switched off!)
2. Open the fuel valve
3. Set throttle lever to idle
4. Open the choke
5. Master switch to "ON"
6. Turn the key in the switch
7. Switch on ignition switches to "RUN"
8. Push the starter button to start the engine
9. Adjust engine RPM after starting
10. Close the choke
11. Warm up the engine

CAUTION
If the cylinder head and oil temperatures fall during parking. Start and warm up engine from time to time between flights. Do not open choke when starting a hot engine.

Parking and taxiing
Check wheel brakes for freezing when parked outside and temperature is below zero. Check wheels free rotation prior to taxiing (Grasp the propeller and pull the airplane). Heat the brakes with hot air (to melt snow or ice). Frozen materials should not be removed by forced towing.

After winter operation
- Clean the airplane thoroughly (hinges, especially the suspensions)
- Lubricate airplane per Lubricating Chart
- Check and adjust the control system

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5. REPAIRS

5.1 Repair guidelines

Dents, cracks and loose rivets are the most frequent problems encounered during the operation of an all-metal airplane.

The following guedelines should be considered during repair:

1. The Strength in any cross-section of a repaired area/part should at a minimum be the same as the strength of the original undamaged area or part. The repaired area is not considered as a whole, but is considered as:
   - an original assembly unit
   - a stiffener or a patch

   If the cross-section of a repaired area is (in all directions), at a minimum, the same as the one on the original part, or if a stiffener or a patch shape and design is similar, and the same materials are used, then there is no risk that the airframe will be weakened.
2. The load has to be transferred by a stiffener or a patch from one side of a repaired area to another.
   In some cases, the load transmission should be carried out at another structural member and then back to an undamaged part of the original structure.
3. The length of overlapping between a reinforcement (stiffener or patch) and theundamaged part of the original structure should be enough to assure a good joint. The length of overlapping is very important to assure no load concentration.

5.2 Damage classification

Various types of damage may occur during aircraft operation. It is important to correctly classify damage according to its character, size and especially, which part of the airplane has been damaged.
The important parts are the engine, engine mount, propeller, wing spar, wing box in the fuselage, elevator, control system, and landing gear.
Minor damage may be repaired by the airplane operator/owner but major structural damage, especially of the above listed parts, should be repaired only by an approved Service center. The damaged parts may be replaced in their entirty by new ones supplied by the aircraft manufacturer.
Any damage and its repair should be recorded in the Log Books.

5.3 Materials used

5.3.1 List of skin sheets

Duralumin sheets plated by aluminum are used for skins.
The following thickness are used: *0.016 in* 0.4 mm
0.020 in 0.5 mm
0.024 in 0.6 mm
Material qualities are shown in the following table :

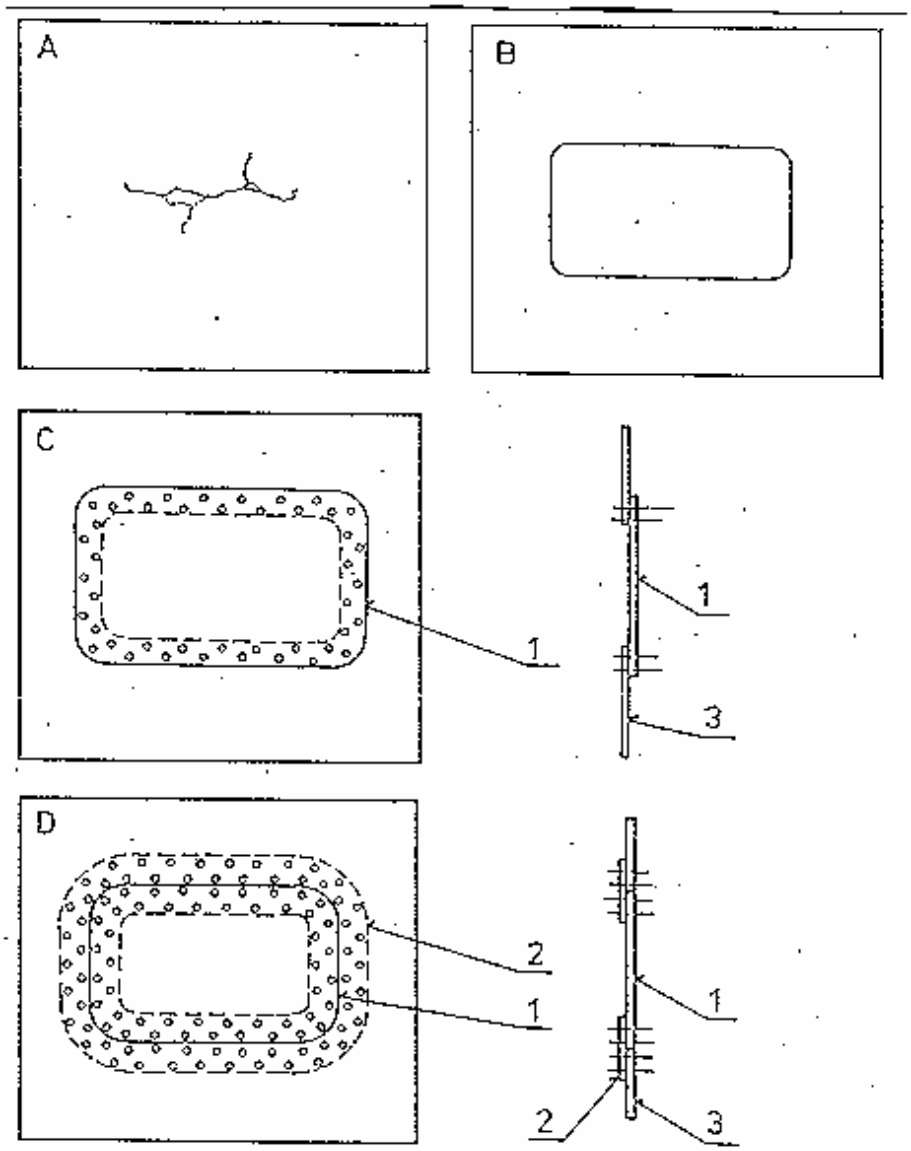
1	Czech Standard		Z 424253.61 Al Cu Mg 1
2	Mechanical properties (Minimum)	Yield Point Rp 0.2	260 MPa
		Tensile strength Rm	400 Mpa
		Ductility [%]	15 (A10)
3	Chemical composition [%]	Al	rest
		Fe max	0.3
		Mn	0.3 - 0.9
		Cu	3.8 - 4.9
		Zn max	0.3
		Mg	1.2 - 1.8
		Si max	0.3
4	Half-finished products		by Al plated Sheets Al thickness max.10 mm *0.4 in*
5	Equivalent Standards	U.S.A. Germany Russia	Alclad 2024 DIN 1745 D16AT

List of Skin Sheets:

Thickness	Airplane part
0.016 in 0.4 mm	Elevator skin (except L.E.) Trim tab skin Rudder skin
0.024 in 0.5 mm	Wing skin between rib 7 and 11 Flaps skin Ailerons skin Skin of upper rear part of a fuselage
0.024 in 0.6 mm	Wing skin between rib 1 and 7 Elevator leading edge Skin of front, rear and bottom part of the fuselage

5.4 Skin repair

1. To prevent small cracks (*0.2 in,* max.5 mm) from spreading, drill a hole of *0.06 in* (1.5 mm), diameter at the end of the crack
2. Small holes (up to ∅ *0.16 in*, 4 mm) can be repaired by a "POP"rivets
3. More extensive skin damage is repaired by means of a patch. A flush or surface patch may be used, as shown on the following figure:



1-Patch
2-Frame
3-Skin

A-Punctured skin
B-Cutting out and deburring the damaged spot
C-Surface patch repair
D-Flush patch repair

Use a sheet of the same thickness as the repaired skin for the patch and frame.
A single row of rivets is used to rivet a small patch (patch area max. *7.75 sq in*. 50 cm^2). Use a double row for greater patches as shown on figure. Paint the internal side of the patch and frame before riveting and then paint the outside surface.

5.4.1 Riveting

"POP" rivets are used in most cases. Aircraft solid rivets are used only on wing spars, the wing box inside the fuselage and on some airplane parts with high loads. These parts must be repaired only by an authorized repair station.
When pop-rivets are to be removed, weaken the rivet head by drilling and then carefully drive out with a drift. Use a drill of the same size as the rivet. The rest of the rivet may be removed from the underside.
Pop-rivet joints are sealed. If necessary use a suitable knife to cut out the sealing compound and then clean the joint. Warm up the sealed joint using hot air gun (*572 °F*, 300 °C) to make it easier to cut.

In the following table lists the pop-rivets and characteristics:

AVEX Pop-rivets supplied by AVDEL Company

1. material : Al Mg 2.5
2. Shank material: steel, zinc-chromate coating surface

Rivet No.:	Diameter		Length		Hole diameter		Max.riveting thickness		Head
	[in]	[mm]	*[in]*	[mm]	*[in]*	[mm]	*[in]*	[mm]	
1691-0410 1691-0414 1691-0512 1691-0516	*0.126* *0.126* *0.157* *0.157*	3.2 3.2 4 4	*0.3110* *0.441* *0.374* *0.050*	7.9 11.2 9.5 12.7	*0.130-0.138* *0.130-0.138* *0.161-0.170* *0.161-0.170*	3.3-3.5 3.3-3.5 4.1-4.3 4.1-4.3	*0.031-0.189* *0.157-0.311* *0.047-0.248* *0.157-0.378*	0.8-4.8 4-7.9 1.2-6.3 4-9.6	button
1604-0412	*0.126*	3.2	*0.374*	9.5	*0.130-0.134*	3.3-3.4	*0.094-0.248*	2.4-6.3	counter-sunk

After drilling and deburring the holes apply Emfimastic PU50 sealing compound or equiv. on either part. Use a spatula to make a homogenous layer (*0.004 in*, 0.1mm).
Riveting should be done before the sealing compound hardenes (approximately half an hour).
For rivets spacing see surrounding skin; usually *0.8-1.6 in* (20-40 mm) and keep a minimum distance of *0.3 in* (8 mm) from the edge.

5.5 Fiberglass parts repairs

Cracks, permanent deformations and breaks are the most prevelant type of damage.
Prior to undertaking the repair, clean the area to be repaired thoroughly, with soap and water. Remove any paint by wet or dry sanding. Superficial scars, scratches, surface abrasions, or erosion can generally be repaired by applying one or more coats of epoxy resin. The number of coats required (2 generally) depends upon the type of resin and severity of damage. The layers should have an overlap of *1.5-2 in* (30-50 mm). After the resin has cured, sand off any excess and prepare the area for primming and painting.

If the area to be repaired is large, use a temporary block or mold coated with PVC or PE foil to support the repaired area.

Listed below are producers and vendors of components to fabricate the fiberglass parts suppied in the kit of EV-97 EUROSTAR airplane.

- Fiber glass clothes
  Producer: VERTEX Litomyšl, Czech Republic
  Vendor: POLYVIA NOVA s.r.o., Nad Vývozem 4844, Zlín, Czech Republic
- Epoxy resin
  VIAPAL UP 600S/70
  Producer: VIANOVA, Austria
  Vendor: POLYVIA NOVA s.r.o., Nad Vývozem 4844, Zlín, Czech Republic
- Gelcoat
  GP 2000 S
  Producer: JOTUN, Norway
  Vendor: REINCHOLD SPOLCHEMIE a.s., Ústí nad Labem, Czech Republic
- Starting Catalyst
  PEROXIMON K 1
  Producer: ELF ATOCHEM ITALIA S.rl., Italy
  Vendor: REINCHOLD SPOLCHEMIE a.s., Ústí nad Labem, Czech Republic

5.6 Paint repairs

Small paint damage may be repainted. Use a small brush to repair small scratches. Spray cans of appropriate color may be used to repaint small to medium areas.

5.7 Airplane assembly and leveling after a repair

When major damage is repaired the following should be performed:

- Technical inspection by qualified personell with attention to wiring, tightening and securing of appropriate joints, clearances in the control systems and control surfaces movement.
- Leveling
- Weight; Balance changes
- Control surfaces deflections measurement
- Engine test run
- Test flight
- Remedy of the defect

5.8 First flight after a repair

CAUTION
Only qualified pilots may perform the test flight.

The following information is intended for the aircraft owner.
A test flight is mandatory in the following cases:
After repair or replacement of fixed surfaces - wing, fin, stabilizer, or control surfaces i.e. elevator, aileron or flap. The flight characteristics and stability can be affected when any of these parts are replaced or adjusted. Therefore a test flight should be performed to check the airplane flight characteristics, control surfaces setting, maneuverability, stability and stall characteristics. These characteristics are to be check during taxiing, takeoff, climbing, cruise, descent, approach and landing..in other words all flight modes!
Use the Flight Test Record (see Appendices of this Manual) to record data obtained during a test flight.

Test flight flying time
The Test flight flying time will take approximately 40 minutes if the Test flight is performed according to the Flight Test record.

Aircraft inspection before a Test flight
Prior to the test flight the aircraft should be thoroughly inspected, weighed, faults corrected by qualified personell. Then and only then the airplane be prepared for the flight.

Test crew
Only qualified persons (test pilots) are permitted to perform a test flight.

Any in-flight faults should be corrected immediatelly after the test flight.

5.9 Spare parts order

Use the "Spare Parts Order" form to order a replacement for a damaged or worn-out part. Do not hesitate to contact the airplane manufacturer (phone or fax) to request a required part.
On the Spare Part Order form additional information is necessary for systematic monitoring and analysis of our planes reliability.

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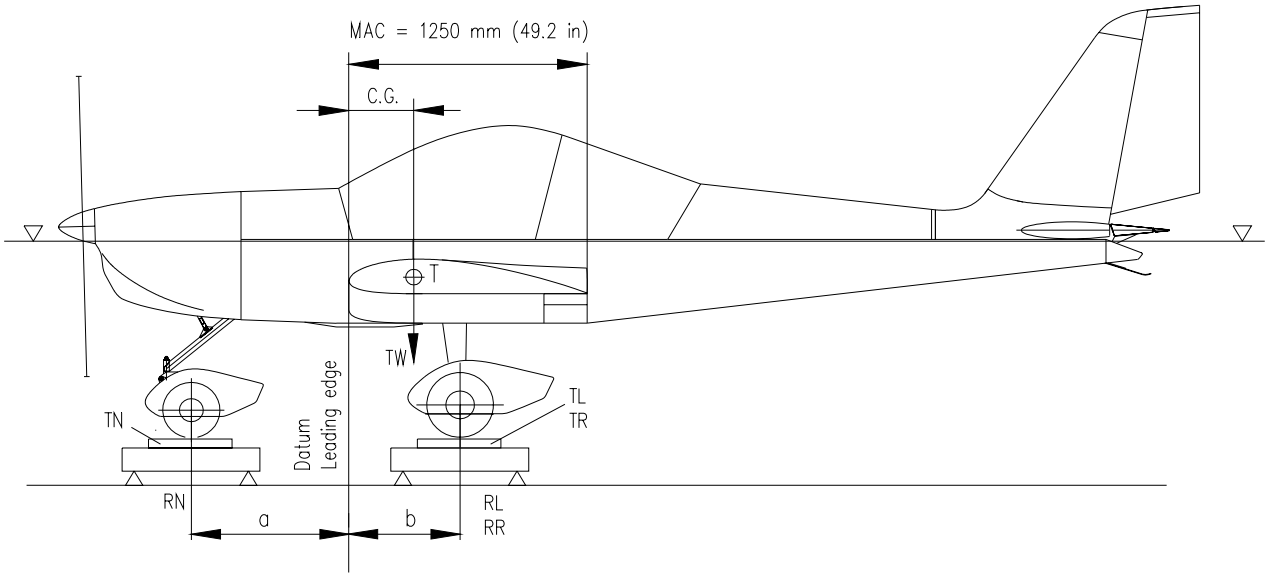
6. APPENDICES

6.1 List of Appendices

- Leveling Record
- Control Surfaces Deflection Record
- Weight and Balance Record
- Flight Test Record
- Spare Parts Order
- Airplane Failure Card
- Airplane Placards
- Airplane Wiring diagrams
- Service Information 18 UL 97 D/E, Issue Jan. 1998

Weight and Balance Record

Model :	EV-97	**Registration :**		**S/N :**	
Configuration :					



Weighing Point	**Scale Reading** R_i **[lbs] or[kg]**	**Tare** T_i **[lbs] or [kg]**	**Net Weight** $NW_i = R_i - T_i$ **[lbs] or [kg]**
Nosewheel	R_N =	T_N =	NW_N =
Left wheel	R_L =	T_L =	NW_L =
Right wheel	R_R =	T_R =	NW_R =
Total Weight [lbs] or [kg] $TW = NW_N + NW_L + NW_R = \ldots\ldots\ldots + \ldots\ldots\ldots + \ldots\ldots\ldots$			**TW =** permitted empty weight (standard equipment): *606 lbs ± 3%* 275 kg ± 3%
C.G. position from Datum (Leading edge) [in] or [mm] $C.G. = \frac{(NW_L + NW_R) \times b - NW_N \times a}{TW} = \frac{(\ldots\ldots\ldots + \ldots\ldots\ldots) \times \ldots\ldots\ldots - \ldots\ldots\ldots \times \ldots\ldots\ldots}{\ldots\ldots\ldots\ldots\ldots\ldots}$			**C.G.=**
C.G. position [% MAC] $\overline{C.G.} = \frac{C.G.\ \text{[in] or [mm]}}{MAC\ \text{[in] or [mm]}} \times 100 = \frac{\ldots\ldots\ldots}{\ldots\ldots\ldots} \times 100$			$\overline{C.G.}$ [% MAC] = permitted C.G. range of empty airplane (standard equipment):: 18 ± 2 % MAC

Weighing carried out by: Date:

Record elaborated	Ing.Javorský	Record approved	Ing.Surý	Change					
Date:	3.5.1999	Date:	3.5.1999						

Control Surfaces Deflection Record

Model:	EV-97	**Registration:**		**S/N:**	

		Specified deflection	Real deflection
Longitudinal control	Distance between neutralized control stick and instrument panel		
	Distance between control stick fully pulled and instrument panel		
	HTU upward deflection	25°±1°	
	HTU downward deflection	20°±1°	
Lateral control	Distance between neutralized control stick and fuselage side frame		
	Distance between control stick moved fully to the left/right and fuselage side frame		
	Left aileron upward deflection	20°±1°	
	Left aileron downward deflection	15°±1°	
	Right aileron upward deflection	20°±1°	
	Right aileron downward deflection	15°±1°	
Rudder control	Rudder left deflection	30°±2°	
	Rudder right deflection	30°±2°	
Wing flaps	Left flap deflection in "TAKEOFF" position	15°±2°	
	Left flap deflection in "LANDING" 1st position	30°±3°	
	Left flap max. deflection in "LANDING" 2nd position	50°±3°	
	Right flap deflection in "TAKEOFF" position	15°±2°	
	Right flap deflection in "LANDING" 1st position	30°±3°	
	Right flap max. deflection in "LANDING" 2nd position	50°±3°	
	Max. difference between left and right flap deflections	2°	
HTU trim tab	Max. upward trim tab deflection (HTU neutralized)	5°±2°	
	Max. downward trim tab deflection (HTU neutralized)	25°±5°	
	Neutral potrim tab position (HTU neutralized, trim tab control lever in neutral position)	0°	

Control Surfaces Deflection compliance: YES - NO

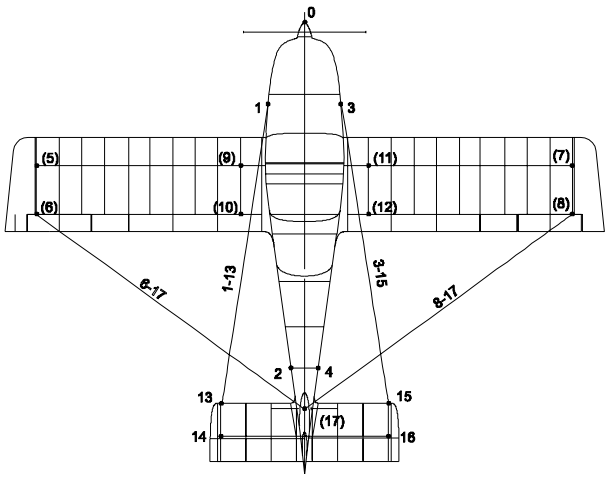
Message for the pilot :

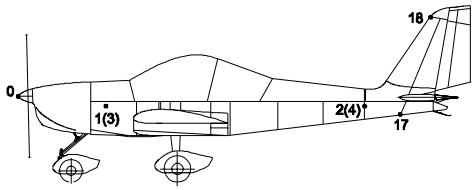
Measurements carried out by: Date:

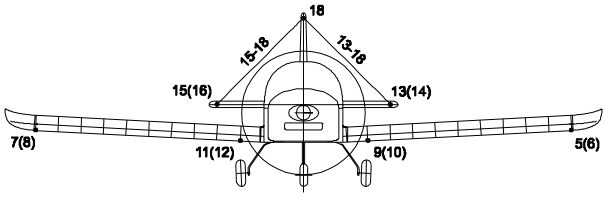
Record elaborated	Ing.Javorský	Record approved	Ing.Surý	Change					
Date:	3.5.1999	Date:	3.5.1999						

Leveling Record

Model:	EV-97	**Registration:**		**S/N:**	







Vertical measurement

Point	Measure
1	
2	
3	
4	
5	
6	
7	
8	
9	
10	
11	
12	
13	
14	
15	
16	
17	

Straight measurement

Points	Measure
1-13	
3-15	
6-17	
8-17	
13-18	
15-18	

Differences left / right side

Specified values			Real Difference
Points	Difference	Tolerance	
1-3	0	*±0.08 in*	
2-4		±2 mm	
1-2	0	*±0.08 in*	
3-4		±2 mm	
5-6	0	*±0.2 in*	
7-8		±5 mm	
9-10	0	*±0.1 in*	
11-12		±3 mm	
5-9	136 mm *5.35 in*	*±0.4 in* ±10 mm	
7-11			
6-10			
8-12			
13-15	0	*±0.4 in*	
14-16		±10 mm	
1-13	0	*±0.8 in*	
3-15		±20 mm	
6-17	0	*±0.8 in*	
8-17		±20 mm	
13-18	0	*±0.4 in*	
15-18		±10 mm	

Levelling carried out by: Date:

Record elaborated	Ing.Javorský	Record approved	Ing.Surý	Change					
Date:	3.5.1999	Date:	3.5.1999						

Flight Test Record

TAKE-OFF WEIGHT *992 lbs* (450 kg)			Page: 1	Pages: 1
Model: EV-97	S/N:	Registration:		
Engine: R 912....	S/N:	Hours flown:		
Prop.:	S/N:			

RESULTS: (Airspeeds are IAS)

Regime			Test
Meteorological conditions		On ground pressure QNH	
		On ground temperature QMU	
		Wind direction/velocity	
Engine check	Full throttle	Engine max. rpm	
		Oil temperature *122 °F* (50 °C)	
		Oil pressure *21.8-72.5 psi* (1.5-5.0 bar)	
	Ignition check	3850 rpm R912 (80 hp) *4000 rpm R 912S (100 hp)* RPM drop max 300 rpm with MAG 1 or 2 off (max. 120 rpm difference)	
	Idle	cca 1400 rpm	
Take-off and climbing to *3300 ft* (1000 m ISA)		Engine speed max. 5500 rpm	
		Oil pressure *21.8-72.5 psi* (1.5-5.0 bar)	
IAS = *62 kts* (115 km/h)		Oil temp. max. 284 °F (140 °C) R912 *266 °F* (130 °C) *R 912S*	
		CHT 302 °F (150 °C) R912 *275 °F* (135 °C) *R 912S*	
		Time of climbing *1000-3300 ft ISA* (300-1000 m ISA)	
Stalling speed at idle		Landing configuration (flaps fully extended) v_{so} = *31 kts* (58 km/h)	
		Cruise configuration v_{S1} = *40 kts* (75 km/h)	
Steep turns		max.45 °	
Never exceed speed		v_{NE} = *146 kts IAS (139 kts CAS)* 270 km/h IAS (256 CAS)	
		Engine speed max. 5800 rpm	
Cruising speed at engine speed: Altitude *1650 ft* (500 m ISA)		Airspeed at 5000 rpm min. *100 kts IAS (98 CAS)* 185 km/h IAS (182 CAS)	
4800- 5000-		Oil temp. *194-230 °F* (90 - 110 °C)	
5200- 5500-		CHT *140-212 °F* (60 – 100 °C)	
Max. horizontal speed Altitude *1650 ft* (500 m ISA)		Airspeed at 5750 rpm min. *116 kts IAS (113 CAS)* 215 km/h IAS (209 CAS)	
		Oil temp. *194-284 °F (266 °F)* (90- 140 °C (130 °C))	
		CHT *140-302 °F (275 °F)* (60-150 °C (135 °C))	
Landing		Touch down speed *kts* or km/h	

Date: Pilot´s signature:

Report: Flight Test Results comply with Type Certificate

Record elaborated	Ing.Javorský	Record approved	Ing.Surý	Change					
Date:	3.5.1999	Date:	3.5.1999						

Spare Parts Order

Please, do not complete gray-shaded areas.

Date:

From:

EVEKTOR - AEROTECHNIK
686 04 KUNOVICE
CZECH REPUBLIC
Tel.: +420 572 537 111
Fax.: +420 572 537 900

1	1.Registration..... 2.S/N..... 3.Number of Take-offs.....		4.Type..... 5.Tot. hours flown..... 6. Date of Failure detection.....		
2	Damaged part ..... NOM		Total oper.hrs.logged.....		
3	Failure detected during		1-Flight	2-Taxying	3-Daily Mainten. 4-Periodic insp.
4	Failure consequence	1-Flight completed	2-Flight interrupted	3-Take-off possible	4-Take-off impossible
5	Failure description:				
6	Repair carried out by:		1-Own means	2-Suppliers service	3-Ordering repair
7	Failure cause:				
8	We order - part(s) stated in par. 2 *) We order - a repair of part(s) stated in par.2 We order - necessary material according to the list enclosed, Number of Pages:.....				

Please, use the following format to the list of ordered parts:

Item No.	Pcs	NOM	Description	Standard, Draw.No.	Note

Other date:

Signature:.....
Date:.....

*) cancel inappropriate

9	Received by:

Airplane Failure Card

Dear customer,
We´d like to ask you for your cooperation in obtaining information required for systematic improvement of our planes reliability. Please, send or transmit completed card to the following address:

EVEKTOR - AEROTECHNIK

Quality Control Department
686 04 Kunovice
CZECH REPUBLIC

Airplane:	EV-97	**Registration:**		**S/N:**	

1.	Failure Description:							
2.	Failure Detection Date: DD-MM-YY							
3.	Damaged Part Title:							
4.	Damaged Part Nomenclature:							
5.	Damaged Part Serial Number:							
6.	Damaged Part Working hours: Hours flown - Number of Take-offs							
7.	Part accessability:	1. Easy	2. Relative easy	3. Difficult	4.Very difficult	5.Dissasembly is neccesary		
8.	Spare Part Title + Serial number:					.....		
9.	Spare part is:	1. New	2. Repaired	3. From another airplane				
10.	Classification	1. No CfA or Flight accident	2. Conditions for accident (CfA)	3. Damage	4. Accident	5. Crash		
11.	Failure detected during:	1. Periodical insp. 2. Taxying 3. Take-off run	4. Take-off 5. Flight 6. Touch down	7. Landing run 8. Cross country flight 9. Pattern flight	10. Accident 11. Othere			
12.	General Cause of Failure:	1. Constructional 2. Service 3. Wear due to operation	4. Unskilled repair 5. Manuf. Maintenance 6. Resulting	7. Objective 8. Air Personnel 9. Not determined				
13.	Consequences of service:	1. No consequences 2. Airplane put out of operation 3. Take-off interrupted	4. Flight with failured units 5. Emergency landing 6. On ground incident	7. Other				
14.	Repair Technique:	1. Adjustment 2. Part replacement 3. Unit replacement	4. Engine exchange 5. Unit repair 6. Airplane repaired	7. Airplane destroyed 8. Other				
15.	Repair lifetime - working hours							
16.	Last inspection before the failure + date							
17.	The airplane used for:							
18.	Total hours flown and Take-offs untill failure occured:							
19.	Remarks, additional information:							
20.	Claimed: Claim No.:	yes	no					
Owner:								
Operator:								

Signature: Date: